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Experimental Investigation of the Effects of Aft Blowing with Various Nozzle Exit Geometries on a 3.0 Caliber Tangent Ogive at High Angles of Attack

Forebody Pressure Distributions

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INVESTIGATION OF THE EFFECTS OF AFT
BLOWING WITH VARIOUS NOZZLE EXIT
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Abstract

An experimental investigation of the effects of aft blowing on the asymmetric vortex flow of a slender, axisymmetric body at high angles of attack has been conducted. A 3.0 caliber tangent ogive body fitted with a cylindrical afterbody was tested in a wind tunnel under subsonic, laminar flow test conditions. Asymmetric blowing from both a single nozzle and a double nozzle configuration, positioned near the body apex, was investigated. Aft blowing was observed to alter the vortex asymmetry by moving the blowing-side vortex closer to the body surface while moving the non-blowing-side vortex further away from the body. The effect of increasing the blowing coefficient was to move the blowing-side vortex closer to the body surface at a more upstream location. The data also showed that blowing was more effective in altering the initial vortex asymmetry at the higher angles of attack than at the lower. The effects of changing the nozzle exit geometry were investigated and it was observed that blowing from a nozzle with a low, broad exit geometry was more effective in reducing the vortex asymmetry than blowing from a high, narrow exit geometry.

Nomenclature

| | |
|-----------------|--|
| A_{ref} | reference area, $40 \times$ model base area |
| C_p | pressure coefficient, $(p - p_\infty)/q_\infty$ |
| C_μ | blowing coefficient, $(\dot{m}_j u_j)/(q_\infty A_{ref})$ |
| d | local diameter of the model |
| D | base diameter of the model |
| \dot{m}_j | mass flow rate through the blowing nozzle |
| p | local static pressure |
| p_∞ | free stream static pressure |
| q_∞ | free stream dynamic pressure |
| u_j | exit velocity from the blowing nozzle |
| z | axial distance from model apex |
| \bar{y} | distance from the surface of the model to the mean geometric center of the nozzle exit orifice |
| \bar{z}_{max} | maximum width of the nozzle exit |
| α | angle of attack |
| ϕ | azimuthal location from windward meridian |
| ϕ_b | azimuthal location of the non-blowing nozzle from the windward meridian |
| ϕ_j | azimuthal location of the blowing nozzle from the windward meridian |

Introduction

The flight of high-performance aircraft at high angles of attack is compromised by the effects of the forebody vortices which form and shed asymmetrically. These asymmetric forebody vortices can produce side forces and yawing moments which may render control of the aircraft difficult or even impossible. This problem is compounded at the higher angles of attack by the fact that the conventional control surfaces (vertical and horizontal stabilisers) are washed out by the wake of the fuselage and wings. The combat agility requirements of present and future generation high-performance aircraft dictate the need for controlled flight at high angles of attack, and thus there is a strong motivation to control the forebody vortex asymmetry in this flight regime.

A substantial body of evidence has been produced in experimental¹⁻⁴ and numerical⁵⁻⁸ studies which indicates that the forebody vortex asymmetry configuration is produced by small imperfections in the tip of the forebody. Many techniques have been studied to control this vortex asymmetry; a recent review has been presented by Ericsson⁷. These techniques include nose bluntness, body reshaping, boundary layer trips, forebody strakes, and forebody suction and blowing. The forebody blowing techniques⁹⁻¹⁷ include normal, forward and aft blowing with respect to the model surface. The previous research in the area of aft blowing has brought about much knowledge in the area of forebody vortex control. For example, a control mechanism has been suggested in references 13-14 for vortex control by jet blowing and is sketched in Figure 1. Once blowing is initiated on the leeward side of the body, the jet entrainment moves the blowing-side separation leeward, thus the vortex on the blowing side of the body moves closer to the body. Due to the coupling of the leeward vortices, the non-blowing-side vortex moves further from the body surface and the separation on the non-blowing side moves windward. Based on this control model, the jet blowing functions primarily to control the flow separation by entrainment due to the jet. Previous research has also shown that (1) the optimal axial location of jet blowing is found to be as close as possible to the forebody apex, since jet blowing at this position can most influence the flow separation and the strong interaction between the vortices; (2) the azimuthal location

of the jet blowing is found to be optimal in the range 120° to 150° , measured from the windward ray; and (3) the baseline system of vortices determines the effectiveness of vortex control by jet blowing. Namely, the jet blowing is more effective for control of the forebody vortex system if the baseline flowfield has only a small degree of vortex asymmetry.^{12,13,18,19}

Although previous researches have demonstrated the potential of aft blowing to provide forebody vortex control, questions remain regarding the fluid dynamic nature of the aft blowing technique. Previous experiments have examined the overall effects of aft blowing on an aircraft configuration. Thus, in contrast to previous studies, an experimental study of the flowfield in the near-tip region of an isolated forebody model was conducted. The objective of this study is to obtain further insight into the mechanisms of aft blowing through detailed measurements of surface pressures and flow visualization in the near-tip region. The effectiveness of asymmetric aft blowing from both a single nozzle and a double nozzle configuration was investigated. The effects of angle of attack, magnitude of blowing, and axial and azimuthal blowing nozzle locations are examined. In addition, the effect of the nozzle exit geometry on the blowing effectiveness is also investigated.

Apparatus and Procedure

Wind Tunnel

All experiments were conducted in the North Carolina State University Subsonic Wind Tunnel Facility, Figure 2. This is a closed return wind tunnel with a settling chamber to test section contraction ratio of 3:1. The settling chamber is equipped with 3 screens located upstream of the contraction section for the purpose of decreasing the free stream turbulence in the test section. The wind tunnel is ventilated to room pressure through a breather located at the downstream end of the test section. The test section is 0.81m in height, 1.14m in width and 1.17m in length and equipped with plexiglass sides and top to permit flow visualization. The test section velocities were regulated by a variable pitch fan located downstream of the test section. The maximum attainable test section velocity was 17.2 m/s.

Ogive Model

The model used for testing was a 3.0 caliber tangent ogive body fitted with a removable nose tip and a cylindrical afterbody as shown in Figure 3. The model was hollow and of aluminum construction. Three circumferential rows of pressure taps were located on the ogive portion of the model, at the locations shown in Figure 3. The two rows of pressure taps located nearest the model apex, rows 1 and 2, had an azimuthal tap spacing of 15° while row 3, the row farthest from the model apex, had an azimuthal tap spacing of 10° . The locations of the pressure taps are tabulated in Table 1. The model was rigidly mounted on a circular arc sting balance. A stepper motor, attached to the sting balance and controlled by a computer, was used to provide variation of the angle of attack. A cylindrical plenum chamber, with internal dimensions of 8.1cm in length and a diameter of 2.1cm, was firmly secured to the internal wall of the model. Dried pressurized air, supplied from an external source, was routed along the sting, through the base of the model and to the plenum, while short lengths of tygon tubing supplied air from the plenum to the blowing nozzle.

Figure 4 shows a schematic of a removable nose tip with the exit of the blowing nozzle located at an axial location of $x/D = 0.125$. The blowing nozzles were designed to blow aft, along a model meridian and tangential to the surface of the body. Previous work conducted by Moskovits² showed that as compared to a discrete surface perturbation of a pointed nose tip, a perturbation of a blunt nose tip was less likely to develop vortex asymmetries due to surface roughness or machining imperfections. Thus for the purposes of this study a blunted nose tip was used to minimize the possible effects of the differences in the geometries of the different blowing nozzles that were tested, and thereby accentuate the effects of blowing.

Blowing Nozzles

Table 2 shows the blowing nozzles that were manufactured for this research. Each blowing nozzle was constructed of brass and was securely fitted to its own nose tip. The geometric mean height of the nozzle exit orifice, \bar{y}/d , was used as a measure of the effective height from the surface of the body to the geometric center of the jet as it exits the blowing

nozzle. The effective width of the jet was characterized by z_{max}/d , which represents the maximum width of the exit orifice. Blowing nozzles 1 - 5 all had the same exterior dimensions of 0.25cm high, 0.44cm wide and 0.51cm long. Each nozzle exit orifice had the same cross-sectional area, but different geometries. Nozzle 1 was a semi-ellipse with a horizontal major axis; 2 was a semi-circle with a horizontal axis; 3 was an ellipse with a horizontal major axis; 4 was a semi-ellipse with a horizontal minor axis, and 5 was a full circle. The numerical designation of the blowing nozzles, 1 - 5, indicated an ascending geometric mean height. For some test cases, a blank nozzle was positioned at a symmetric location to the blowing nozzle with respect to the windward ray. The purpose of this blank nozzle was to provide an initial vortex pattern that was less asymmetric when compared with a single nozzle being placed on the model. These blank nozzles were of the same exterior dimensions as the blowing nozzles and were glued directly onto the model surface.

Nozzle Calibration

A method of calibrating the blowing nozzles was developed to determine the level of blowing. A simple calibration apparatus, shown in Figure 5, was assembled for this purpose. It consisted of a pressure regulator used to vary the plenum stagnation pressure; a pressure transducer to measure the plenum pressure; and an in-line flow meter positioned between the plenum and the blowing nozzle to measure the volumetric flow rate of the jet. Prior to the nozzle calibration, the pressure drop across the flow meter was measured, and was observed to be negligible. Each section of tubing used in the calibration procedure was of the same length as that used during the subsequent wind tunnel testing.

From the calibration apparatus, the stagnation pressure and volumetric flow rate were measured while the stagnation temperature was taken to be the ambient laboratory temperature. The blowing coefficient, C_μ , was then calculated where A_{ref} was taken to be $40(\pi D^2/4)$. This reference area was chosen to enable comparison of the blowing coefficient with previous researches.

Test Instrumentation & Parameters

Surface pressures were measured using a pair of

8.9cm of water Validyne differential pressure transducers connected to a pair of 48-port Scanivalve modules and a Hewlett-Packard 9122 computer. The transducers' sampling time was 0.167 seconds, and thus time averaged pressures were obtained.

Wind tunnel testing was conducted at a free stream velocity of 13.7 m/s. This corresponded to a laminar flow Reynolds numbers, based on the model base diameter, of 84000. The angle of attack was varied from 40° to 60° in 10° increments, while sideslip was held constant at 0°. C_μ 's investigated ranged from 0.01 to 0.02 for group B. A test case with the blowing nozzle sealed, i.e. $C_\mu = 0$, was also investigated. The azimuthal locations of the blowing nozzles were 90°, 120°, and 150° for the single nozzle configuration, while the 90° and 120° locations were tested for the double nozzle configuration.

Results and Discussion

Tables 3 and 4 show the test cases and corresponding figures for the single and double nozzle configurations respectively. Each figure consists of the surface pressure data for all 5 blowing nozzles. Although a detailed discussion of the results is not presented here, interesting features regarding the data are noted. (For a more thorough analysis of the present data references 19 - 22 are suggested.)

The following observations were made from the data presented in this report. (i) When comparing the blowing cases with the non-blowing cases it was observed that aft blowing was effective in reducing the initial vortex asymmetry. Aft blowing moved the blowing-side vortex closer to the surface of the model while the non-blowing-side vortex moved farther away from the body. It was also observed that blowing moved the separation location of the viscous layer from the body to a more leeward location. (ii) As the angle of attack was varied aft blowing was observed to be more effective at the higher angles of attack than at the lower. It is believed that this was due to the more effective augmentation of the axial flow component over the model as the angle of attack was increased. (iii) Localized differences in the C_p distributions were observed as C_μ was varied. The effect of increasing the magnitude of the blowing coefficient was to move the blowing-side vortex closer to the model surface over a shorter axial distance. (iv) Finally low, broad nozzle cross-sectional

exit geometries were observed to be more effective in reducing forebody vortex asymmetry than high, narrow cross-sections. This is consistent with the optimal conditions for the entrainment of the forebody flow by blowing, since the jet surface area is then maximum. This supports the previously proposed control mechanism of jet entrainment effects being responsible for forebody vortex control using aft blowing.

Conclusions

An experimental study has been conducted to examine the effectiveness of aft blowing as a method of forebody vortex control. A 3.0 caliber tangent ogive model fitted with a cylindrical afterbody was tested in subsonic, laminar flow conditions. Testing was conducted using both a single nozzle and a double nozzle configuration; for the double nozzle configuration, blowing was applied through only one nozzle. Blowing was optimized when a low, broad nozzle was used, when the blowing coefficient was maximized, and the model was at the highest angle of attack. The experimental results presented here will be useful for comparison with computational methods.

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References

1. Moskovitz, C.A., Hall, R.M., and DeJarnette, F.R., "Effects of Surface Perturbations on the Asymmetric Vortex Flow Over a Slender Body," AIAA 88-0483, January 1988.
2. Moskovitz, C.A., "An Experimental Investigation of the Physical Mechanisms Controlling the Asymmetric Flow Past Slender Bodies at Large Angles of Attack," Ph.D. Dissertation, North Carolina State University, Raleigh, 1989.
3. Moskovitz, C.A., Hall, R.M., and DeJarnette, F.R., "Combined Effects of Nose Bluntness and Surface Perturbations on Asymmetric Flow Past Slender Bodies," *Journal of Aircraft*, Vol. 27, October 1990, pp. 909-910.
4. Zilliac, G.G. and Degani, D., "Asymmetric Vortices on a Slender Body of Revolution," AIAA 90-0389, January 1990.
5. Hartwich, P.M., Hall R.M., and Hemsch, M.J., "Navier-Stokes Computations of Vortex Asymmetries Controlled by Small Surface Imperfections," *Journal of Spacecraft and Rockets*, Vol. 28, March-April 1991, pp. 258-264.
6. Degani, D., "Effect of Geometrical Disturbance on Vortex Asymmetry," *AIAA Journal*, Vol. 29, April 1991, pp. 560-566.
7. Ericsson, L.E., "Control of Forebody Flow Asymmetry - A Critical Review," AIAA 90-2833CP, August 1990.
8. Peake, D.J., Owen, F.K., and Johnson, D.A., "Control of Forebody Vortex Orientation to Alleviate Side Forces," AIAA 80-0183, January 1980.
9. Skow, A.M., Moore, W.A., and Lorincz, D.J., "Control of Forebody Vortex Orientation to Enhance Departure Recovery of Fighter Aircraft," *Journal of Aircraft*, Vol. 19, October 1982, pp. 812-819.
10. Malcolm, G.N. and Skow, A.M., "Enhanced Controllability Through Vortex Manipulation of Fighter Aircraft at High Angles of Attack," AIAA 86-2277CP, August 1986.
11. Malcolm, G.N., Ng, T.T., Lewis, L.C., and Murri, D.G., "Development of Non-Conventional Control Methods for High Angles of Attack Flight Using Vortex Manipulation," AIAA 89-2192CP, August 1989.
12. Rosen, B.S. and Davis, W.H., "Numerical Study of Asymmetric Air Injection to Control High Angle-of-Attack Forebody Vortices on the X-29 Aircraft," AIAA 90-3004CP, August 1990.
13. Ng, T.T. and Malcolm, G.N., "Aerodynamic Control Using Forebody Vortex Control," Extended Abstract submitted to the High Angle of Attack Technology Conference, Hampton VA, October 1990.
14. Ng, T.T. and Malcolm, G.N., "Aerodynamic Control Using Forebody Blowing and Suction," AIAA 91-0619, January 1991.
15. Cornelius, K.C., Pandit, N., Osborn, R.F., and Guyton, R.W., "An Experimental Study of Pneumatic Vortex Flow Control on High Angle of Attack Forebody Model," AIAA 92-0018, January 1992.
16. Guyton, R.W. and Maerki, G., "X-29 Forebody Jet Blowing," AIAA 92-0017, January 1992.

17. LeMay, S.P., Sewall, W.G., and Henderson, J.F., "Forebody Vortex Flow Control on the F-16C Using Tangential Slot and Jet Nossle Blowing," AIAA 92-0019, January 1992.
18. Ng, T.T., Ong, L.Y., Suarez, C.J., and Malcolm, G.N., "Wing Rock Suppression Using Forebody Vortex Control," AIAA 91-3227CP, September 1991.
19. Gittner, N.M. and Chokani, N., "An Experimental Study of the Effects of Aft Blowing on a 3.0 Caliber Tangent Ogive Body at High Angles of Attack," AIAA 91-3252CP, September 1991.
20. Gittner, N.M. and Chokani, N., "The Effects of Nossle Exit Geometry on Forebody Vortex Control Using Blowing," AIAA 92-2603CP, June 1992.
21. Gittner, N.M., "An Experimental Study of the Effects of Aft Blowing on a 3.0 Caliber Tangent Ogive Body at High Angles of Attack," MS dissertation, Department of Mechanical & Aerospace Engineering, North Carolina State University, Raleigh, NC, March 1992.

| Port | x(cm) | ϕ (deg) | Port | x(cm) | ϕ (deg) |
|------|---------|--------------|------|---------|--------------|
| 1 | -8.255 | .0 | 43 | -14.605 | 270.0 |
| 2 | -8.255 | 15.0 | 44 | -14.605 | 285.0 |
| 3 | -8.255 | 30.0 | 45 | -14.605 | 300.0 |
| 4 | -8.255 | 45.0 | 46 | -14.605 | 315.0 |
| 5 | -8.255 | 60.0 | 47 | -14.605 | 330.0 |
| 6 | -8.255 | 75.0 | 48 | -14.605 | 345.0 |
| 7 | -8.255 | 90.0 | 49 | -20.320 | .0 |
| 8 | -8.255 | 105.0 | 50 | -20.320 | 10.0 |
| 9 | -8.255 | 120.0 | 51 | -20.320 | 20.0 |
| 10 | -8.255 | 135.0 | 52 | -20.320 | 30.0 |
| 11 | -8.255 | 150.0 | 53 | -20.320 | 40.0 |
| 12 | -8.255 | 165.0 | 54 | -20.320 | 50.0 |
| 13 | -8.255 | 180.0 | 55 | -20.320 | 60.0 |
| 14 | -8.255 | 195.0 | 56 | -20.320 | 70.0 |
| 15 | -8.255 | 210.0 | 57 | -20.320 | 80.0 |
| 16 | -8.255 | 225.0 | 58 | -20.320 | 90.0 |
| 17 | -8.255 | 240.0 | 59 | -20.320 | 100.0 |
| 18 | -8.255 | 255.0 | 60 | -20.320 | 110.0 |
| 19 | -8.255 | 270.0 | 61 | -20.320 | 120.0 |
| 20 | -8.255 | 285.0 | 62 | -20.320 | 130.0 |
| 21 | -8.255 | 300.0 | 63 | -20.320 | 140.0 |
| 22 | -8.255 | 315.0 | 64 | -20.320 | 150.0 |
| 23 | -8.255 | 330.0 | 65 | -20.320 | 160.0 |
| 24 | -8.255 | 345.0 | 66 | -20.320 | 170.0 |
| 25 | -14.605 | .0 | 67 | -20.320 | 180.0 |
| 26 | -14.605 | 15.0 | 68 | -20.320 | 190.0 |
| 27 | -14.605 | 30.0 | 69 | -20.320 | 200.0 |
| 28 | -14.605 | 45.0 | 70 | -20.320 | 210.0 |
| 29 | -14.605 | 60.0 | 71 | -20.320 | 220.0 |
| 30 | -14.605 | 75.0 | 72 | -20.320 | 230.0 |
| 31 | -14.605 | 90.0 | 73 | -20.320 | 240.0 |
| 32 | -14.605 | 105.0 | 74 | -20.320 | 250.0 |
| 33 | -14.605 | 120.0 | 75 | -20.320 | 260.0 |
| 34 | -14.605 | 135.0 | 76 | -20.320 | 270.0 |
| 35 | -14.605 | 150.0 | 77 | -20.320 | 280.0 |
| 36 | -14.605 | 165.0 | 78 | -20.320 | 290.0 |
| 37 | -14.605 | 180.0 | 79 | -20.320 | 300.0 |
| 38 | -14.605 | 195.0 | 80 | -20.320 | 310.0 |
| 39 | -14.605 | 210.0 | 81 | -20.320 | 320.0 |
| 40 | -14.605 | 225.0 | 82 | -20.320 | 330.0 |
| 41 | -14.605 | 240.0 | 83 | -20.320 | 340.0 |
| 42 | -14.605 | 255.0 | 84 | -20.320 | 350.0 |

Table 1 – Pressure Port Locations

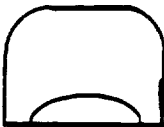
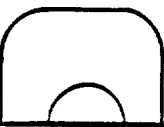
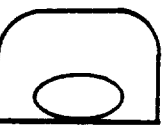
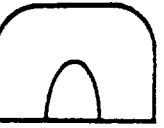
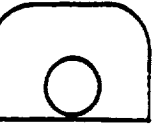
| Blowing Nozzle | Exit Geometries | z/D | \bar{y}/d | z_{max}/d |
|----------------|---|-------|-------------|-------------|
| 1 |  | 0.125 | 0.0354 | 0.334 |
| 2 |  | 0.125 | 0.0499 | 0.235 |
| 3 |  | 0.125 | 0.0588 | 0.235 |
| 4 |  | 0.125 | 0.0627 | 0.167 |
| 5 |  | 0.125 | 0.0836 | 0.167 |

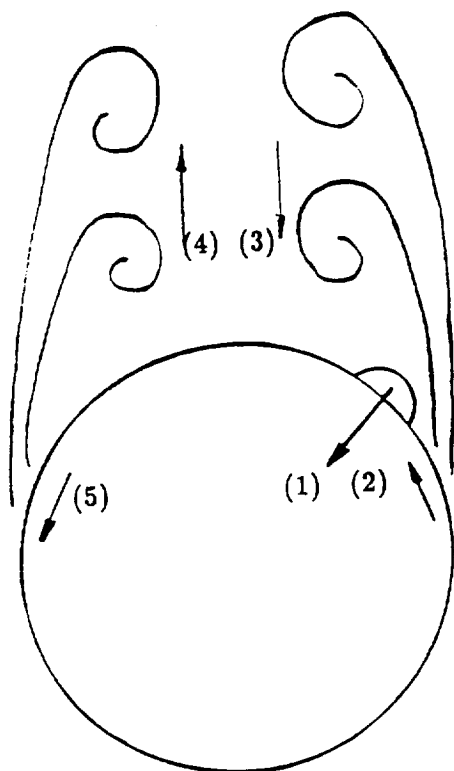
Table 1 – Blowing Nozzle Specifications

| Figure | α (deg) | ϕ_j (deg) | C_μ |
|--------|----------------|----------------|---------|
| 6 | 40 | 90 | .01 |
| 7 | 40 | 90 | .02 |
| 8 | 40 | 120 | .00 |
| 9 | 40 | 120 | .01 |
| 10 | 40 | 120 | .02 |
| 11 | 40 | 150 | .01 |
| 12 | 40 | 150 | .02 |
| 13 | 50 | 90 | .01 |
| 14 | 50 | 90 | .02 |
| 15 | 50 | 120 | .00 |
| 16 | 50 | 120 | .01 |
| 17 | 50 | 120 | .02 |
| 18 | 50 | 150 | .01 |
| 19 | 50 | 150 | .02 |
| 20 | 60 | 90 | .01 |
| 21 | 60 | 90 | .02 |
| 22 | 60 | 120 | .00 |
| 23 | 60 | 120 | .01 |
| 24 | 60 | 120 | .02 |
| 25 | 60 | 150 | .01 |
| 26 | 60 | 150 | .02 |

Table 3 - Test Matrix
Single Nozzle Configuration

| Figure | α (deg) | ϕ_j (deg) | ϕ_b (deg) | C_μ |
|--------|----------------|----------------|----------------|---------|
| 27 | 40 | 90 | 270 | .00 |
| 28 | 40 | 90 | 270 | .01 |
| 29 | 40 | 90 | 270 | .02 |
| 30 | 40 | 120 | 240 | .00 |
| 31 | 40 | 120 | 240 | .01 |
| 32 | 40 | 120 | 240 | .02 |
| 33 | 50 | 90 | 270 | .00 |
| 34 | 50 | 90 | 270 | .01 |
| 35 | 50 | 90 | 270 | .02 |
| 36 | 50 | 120 | 240 | .00 |
| 37 | 50 | 120 | 240 | .01 |
| 38 | 50 | 120 | 240 | .02 |
| 39 | 60 | 90 | 270 | .00 |
| 40 | 60 | 90 | 270 | .01 |
| 41 | 60 | 90 | 270 | .02 |
| 42 | 60 | 120 | 240 | .00 |
| 43 | 60 | 120 | 240 | .01 |
| 44 | 60 | 120 | 240 | .02 |

Table 4 – Test Matrix
Double Nozzle Configuration



- (1) Blowing is initiated.
- (2) Separation is moved leeward due to entrainment.
- (3) Blowing-side vortex moves towards body.
- (4) Non-blowing side vortex moves away from body.
- (5) Separation is moved windward.

Figure 1 - Effects of Aft Blowing on the Leeside Vortices (ref. 14)

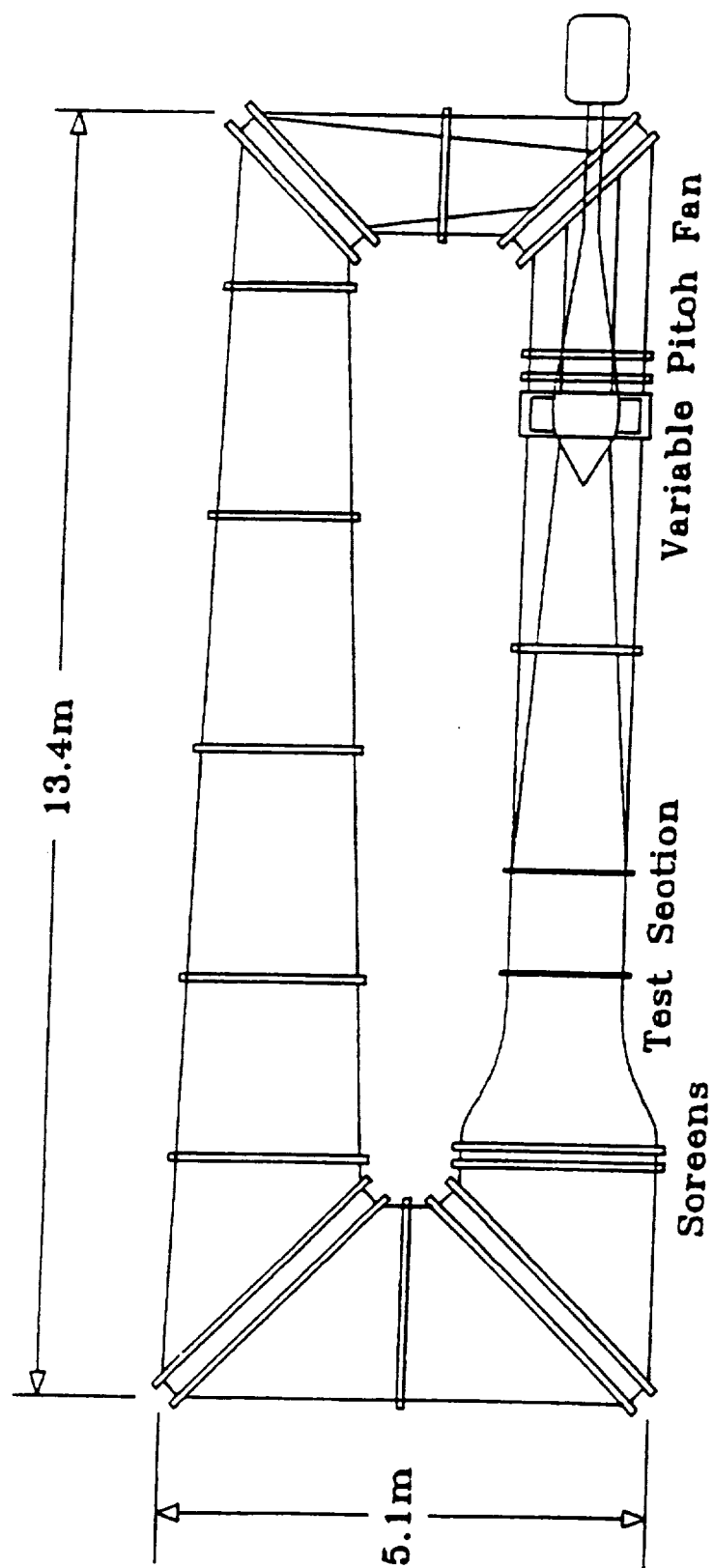


Figure 2 - North Carolina State University Subsonic Wind Tunnel Facility

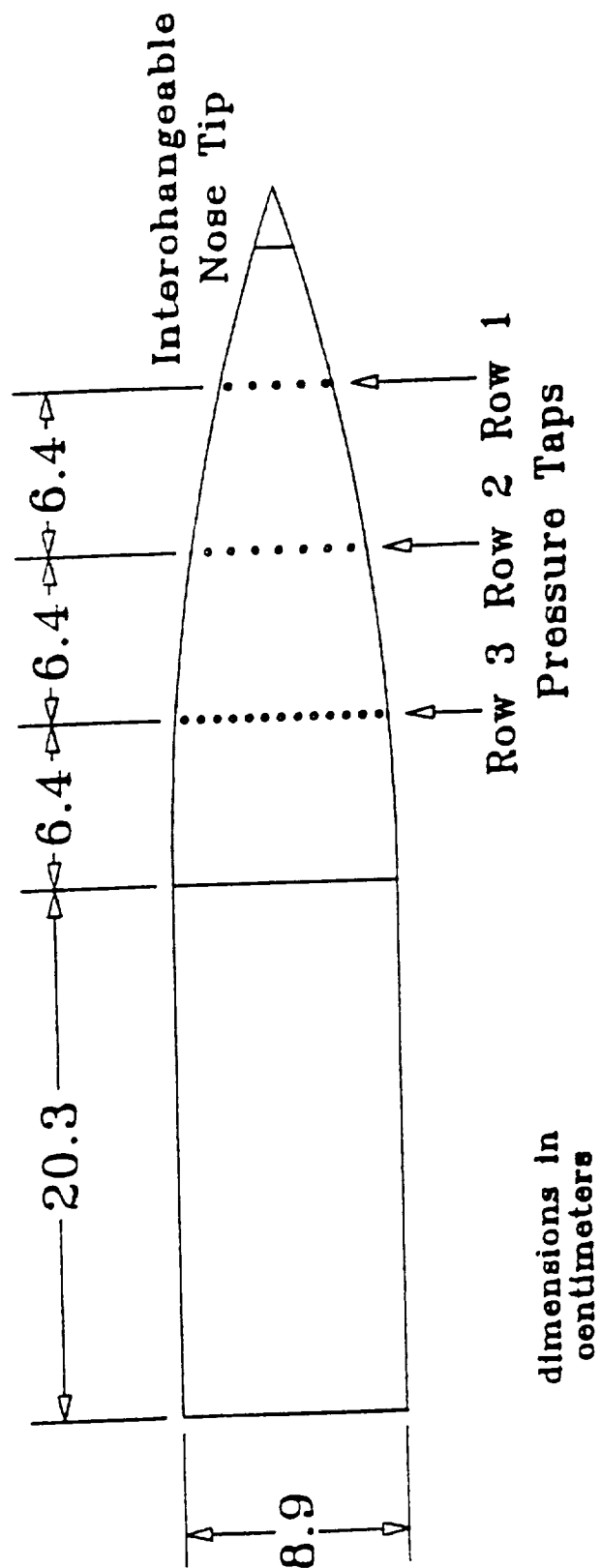


Figure 3 - 3.0 Caliber Tangent Ogive Model

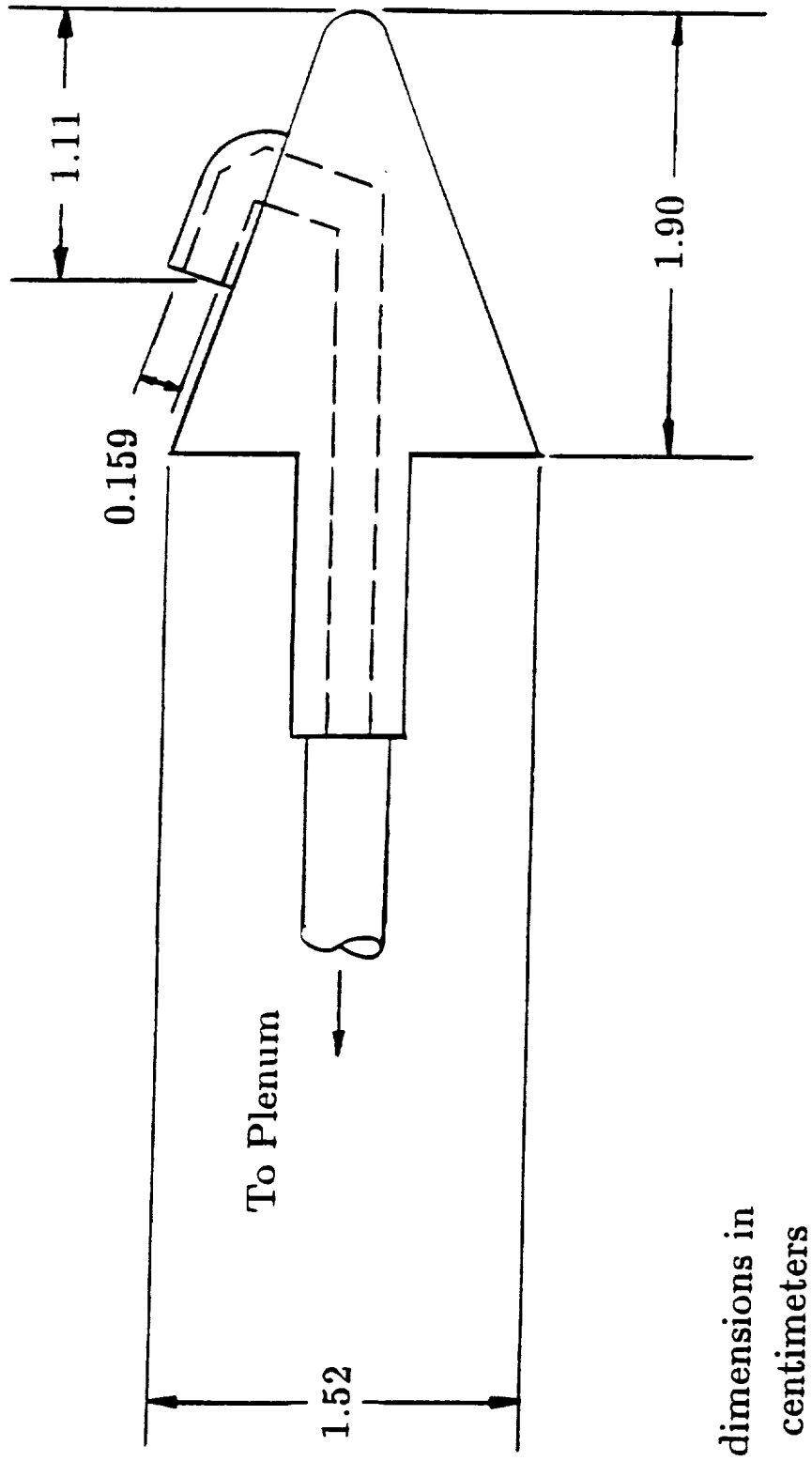


Figure 4 – Removable Nose Tip with Blowing Nozzle

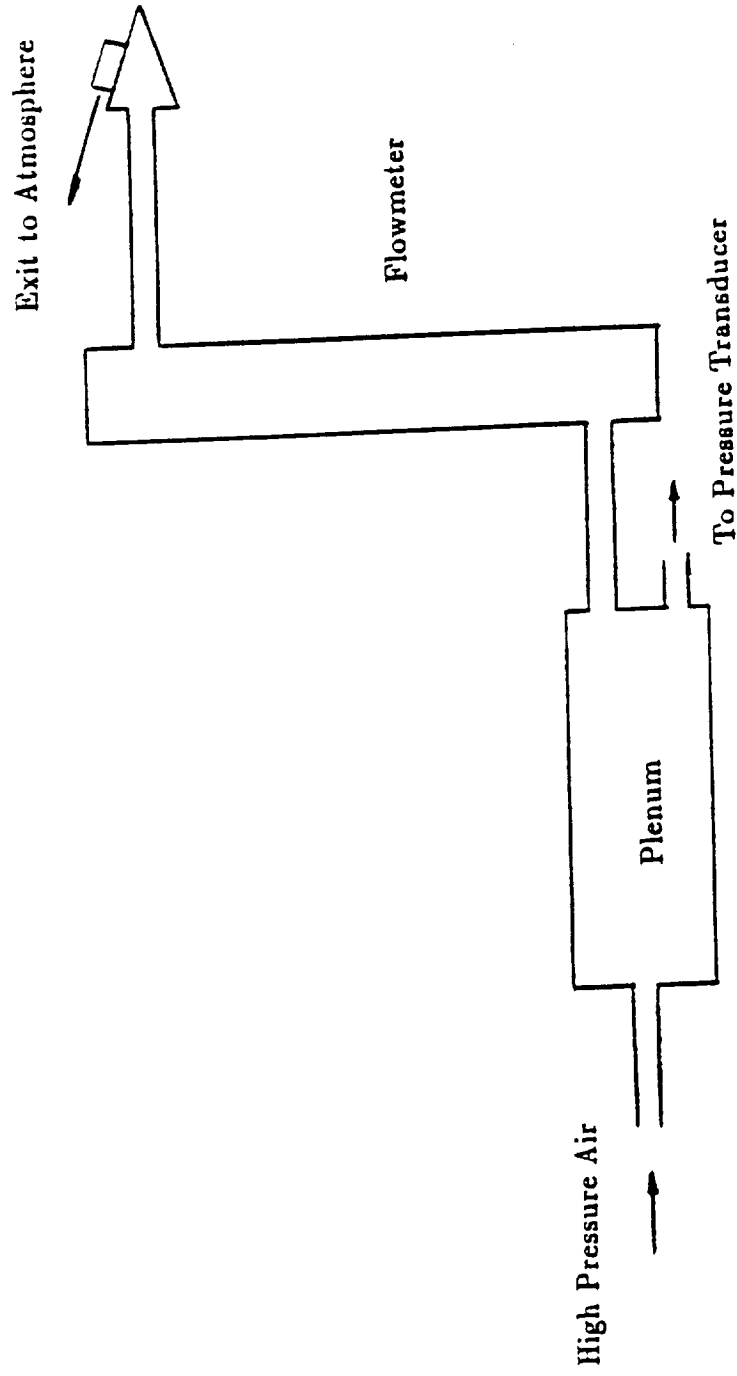


Figure 5 - Blowing Nozzle Calibration Test Schematic

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .678 | .674 | .670 | .648 | .637 |
| 2 | .552 | .563 | .570 | .594 | .552 |
| 3 | .307 | .322 | .341 | .409 | .358 |
| 4 | -.121 | -.093 | -.083 | .009 | -.028 |
| 5 | -.550 | -.507 | -.507 | -.391 | -.413 |
| 6 | -.506 | -.474 | -.494 | -.385 | -.391 |
| 7 | -.795 | -.772 | -.792 | -.715 | -.714 |
| 8 | -.907 | -.886 | -.765 | -.661 | -.780 |
| 9 | -1.244 | -1.180 | -1.421 | -1.227 | -1.221 |
| 10 | -.941 | -.915 | -.919 | -.861 | -.919 |
| 11 | -.827 | -.822 | -.909 | -.889 | -.851 |
| 12 | -.604 | -.598 | -.610 | -.613 | -.618 |
| 13 | -.534 | -.551 | -.530 | -.529 | -.539 |
| 14 | -.979 | -.980 | -.994 | -.914 | -.882 |
| 15 | -1.082 | -1.033 | -1.111 | -1.183 | -1.093 |
| 16 | -1.162 | -1.162 | -1.140 | -1.140 | -1.095 |
| 17 | -1.217 | -1.203 | -1.218 | -1.293 | -1.203 |
| 18 | -1.241 | -1.246 | -1.194 | -1.251 | -1.181 |
| 19 | -1.318 | -1.317 | -1.316 | -1.391 | -1.309 |
| 20 | -1.093 | -1.087 | -1.086 | -1.172 | -1.085 |
| 21 | -.806 | -.798 | -.846 | -.943 | -.865 |
| 22 | -.338 | -.329 | -.374 | -.471 | -.411 |
| 23 | .148 | .157 | .102 | .024 | .063 |
| 24 | .449 | .459 | .419 | .368 | .376 |
| 25 | .487 | .489 | .481 | .461 | .458 |
| 26 | .414 | .417 | .420 | .434 | .403 |
| 27 | .191 | .191 | .226 | .281 | .237 |
| 28 | -.168 | -.158 | -.117 | -.050 | -.093 |
| 29 | -.658 | -.645 | -.611 | -.498 | -.543 |
| 30 | -.936 | -.914 | -.885 | -.776 | -.813 |
| 31 | -1.152 | -1.098 | -1.117 | -1.011 | -1.034 |
| 32 | -1.033 | -.992 | -.994 | -.907 | -.923 |
| 33 | -1.189 | -1.151 | -1.156 | -1.063 | -1.079 |
| 34 | -.657 | -.642 | -.654 | -.589 | -.618 |
| 35 | -.641 | -.650 | -.677 | -.582 | -.620 |
| 36 | -.426 | -.429 | -.430 | -.408 | -.429 |
| 37 | -.478 | -.486 | -.472 | -.434 | -.470 |
| 38 | -.981 | -.998 | -.798 | -.729 | -.897 |
| 39 | -1.043 | -1.082 | -1.039 | -1.046 | -1.027 |
| 40 | -1.136 | -1.147 | -1.047 | -1.066 | -1.092 |
| 41 | -1.146 | -1.157 | -1.141 | -1.198 | -1.130 |
| 42 | -1.097 | -1.106 | -1.060 | -1.134 | -1.071 |

Figure 6 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.255 | -1.261 | -1.238 | -1.317 | -1.243 |
| 44 | -1.082 | -1.083 | -1.070 | -1.160 | -1.082 |
| 45 | -.900 | -.884 | -.921 | -1.008 | -.932 |
| 46 | -.503 | -.486 | -.533 | -.614 | -.555 |
| 47 | -.049 | -.033 | -.086 | -.173 | -.111 |
| 48 | .253 | .269 | .223 | .167 | .194 |
| 49 | .415 | .409 | .407 | .395 | .380 |
| 50 | .403 | .404 | .420 | .433 | .405 |
| 51 | .237 | .234 | .261 | .298 | .264 |
| 52 | .012 | .008 | .051 | .119 | .065 |
| 53 | -.241 | -.249 | -.202 | -.116 | -.166 |
| 54 | -.597 | -.579 | -.536 | -.439 | -.483 |
| 55 | -.827 | -.812 | -.769 | -.669 | -.698 |
| 56 | -1.131 | -1.108 | -1.086 | -.970 | -1.004 |
| 57 | -1.200 | -1.181 | -1.161 | -1.050 | -1.080 |
| 58 | -1.201 | -1.165 | -1.189 | -1.085 | -1.102 |
| 59 | -1.130 | -1.087 | -1.095 | -.996 | -1.021 |
| 60 | -1.191 | -1.151 | -1.182 | -1.086 | -1.102 |
| 61 | -1.123 | -1.085 | -1.093 | -1.009 | -1.018 |
| 62 | -1.343 | -1.180 | -1.198 | -1.098 | -1.125 |
| 63 | -1.119 | -1.084 | -1.085 | -1.056 | -1.181 |
| 64 | -.540 | -.489 | -.521 | -.449 | -.468 |
| 65 | -.507 | -.500 | -.530 | -.454 | -.510 |
| 66 | -.345 | -.357 | -.391 | -.317 | -.329 |
| 67 | -.434 | -.429 | -.464 | -.369 | -.395 |
| 68 | -.841 | -.834 | -.765 | -.697 | -.741 |
| 69 | -1.025 | -1.039 | -.891 | -.985 | -1.014 |
| 70 | -1.191 | -1.173 | -1.033 | -1.123 | -1.142 |
| 71 | -.953 | -.962 | -.919 | -1.006 | -.957 |
| 72 | -1.061 | -1.068 | -1.045 | -1.147 | -1.068 |
| 73 | -.919 | -.934 | -.913 | -.986 | -.911 |
| 74 | -.975 | -.986 | -.985 | -1.081 | -.981 |
| 75 | -.922 | -.934 | -.913 | -.996 | -.914 |
| 76 | -1.061 | -1.079 | -1.054 | -1.148 | -1.048 |
| 77 | -1.020 | -1.035 | -1.011 | -1.102 | -1.021 |
| 78 | -1.010 | -1.021 | -1.029 | -1.134 | -1.036 |
| 79 | -.807 | -.817 | -.822 | -.923 | -.849 |
| 80 | -.585 | -.588 | -.627 | -.733 | -.649 |
| 81 | -.308 | -.315 | -.341 | -.442 | -.377 |
| 82 | -.014 | -.025 | -.068 | -.149 | -.089 |
| 83 | .184 | .173 | .145 | .081 | .116 |
| 84 | .396 | .396 | .379 | .343 | .343 |

Figure 6 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .684 | .686 | .665 | .646 | .638 |
| 2 | .556 | .559 | .583 | .594 | .574 |
| 3 | .317 | .331 | .353 | .449 | .374 |
| 4 | -.076 | -.073 | -.115 | .097 | .009 |
| 5 | -.470 | -.477 | -.582 | -.256 | -.355 |
| 6 | -.362 | -.381 | -.624 | -.228 | -.298 |
| 7 | -1.187 | -.884 | -.936 | -.558 | -.925 |
| 8 | -1.678 | -1.252 | -1.441 | -.769 | -1.278 |
| 9 | -1.437 | -1.436 | -1.671 | -1.135 | -1.587 |
| 10 | -1.062 | -1.001 | -.849 | -.846 | -1.056 |
| 11 | -.892 | -.873 | -.909 | -.861 | -.910 |
| 12 | -.584 | -.554 | -.539 | -.600 | -.589 |
| 13 | -.540 | -.540 | -.486 | -.564 | -.480 |
| 14 | -.964 | -1.014 | -.981 | -.987 | -.975 |
| 15 | -1.070 | -1.035 | -1.130 | -1.266 | -1.068 |
| 16 | -1.158 | -1.165 | -1.141 | -1.227 | -1.108 |
| 17 | -1.220 | -1.223 | -1.214 | -1.393 | -1.184 |
| 18 | -1.244 | -1.241 | -1.199 | -1.339 | -1.188 |
| 19 | -1.333 | -1.343 | -1.294 | -1.498 | -1.291 |
| 20 | -1.100 | -1.090 | -1.095 | -1.242 | -1.098 |
| 21 | -.824 | -.820 | -.829 | -1.029 | -.848 |
| 22 | -.359 | -.342 | -.375 | -.526 | -.401 |
| 23 | .141 | .148 | .105 | -.025 | .076 |
| 24 | .441 | .444 | .431 | .333 | .401 |
| 25 | .490 | .492 | .468 | .459 | .443 |
| 26 | .410 | .415 | .435 | .434 | .418 |
| 27 | .197 | .201 | .222 | .309 | .235 |
| 28 | -.162 | -.148 | -.115 | -.014 | -.094 |
| 29 | -.618 | -.623 | -.557 | -.409 | -.520 |
| 30 | -.879 | -.860 | -.842 | -.671 | -.806 |
| 31 | -1.069 | -1.038 | -1.043 | -.867 | -.972 |
| 32 | -.928 | -.903 | -.940 | -.747 | -.873 |
| 33 | -1.020 | -.950 | -.968 | -.911 | -.956 |
| 34 | -.780 | -.832 | -.876 | -.546 | -.863 |
| 35 | -.618 | -.594 | -.644 | -.541 | -.627 |
| 36 | -.541 | -.546 | -.608 | -.408 | -.619 |
| 37 | -.511 | -.505 | -.508 | -.447 | -.525 |
| 38 | -.998 | -1.017 | -.852 | -.863 | -.924 |
| 39 | -1.090 | -1.138 | -.976 | -1.150 | -1.026 |
| 40 | -1.141 | -1.136 | -1.077 | -1.164 | -1.095 |
| 41 | -1.166 | -1.188 | -1.092 | -1.303 | -1.113 |
| 42 | -1.094 | -1.089 | -1.084 | -1.212 | -1.069 |

Figure 7 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.265 | -1.280 | -1.219 | -1.429 | -1.209 |
| 44 | -1.080 | -1.067 | -1.090 | -1.224 | -1.082 |
| 45 | -.911 | -.898 | -.909 | -1.094 | -.911 |
| 46 | -.504 | -.477 | -.537 | -.668 | -.551 |
| 47 | -.059 | -.048 | -.078 | -.194 | -.109 |
| 48 | .242 | .256 | .240 | .150 | .206 |
| 49 | .412 | .402 | .411 | .379 | .389 |
| 50 | .411 | .409 | .412 | .438 | .401 |
| 51 | .237 | .238 | .268 | .313 | .264 |
| 52 | .012 | .019 | .054 | .142 | .069 |
| 53 | -.244 | -.230 | -.193 | -.078 | -.160 |
| 54 | -.580 | -.567 | -.522 | -.392 | -.483 |
| 55 | -.802 | -.776 | -.770 | -.597 | -.715 |
| 56 | -1.117 | -1.092 | -1.058 | -.891 | -.999 |
| 57 | -1.177 | -1.133 | -1.166 | -.951 | -1.102 |
| 58 | -1.178 | -1.151 | -1.147 | -.974 | -1.101 |
| 59 | -1.091 | -1.048 | -1.091 | -.898 | -1.047 |
| 60 | -1.179 | -1.147 | -1.143 | -.994 | -1.103 |
| 61 | -1.103 | -1.065 | -1.094 | -.906 | -1.054 |
| 62 | -1.137 | -1.094 | -1.112 | -.989 | -1.065 |
| 63 | -1.157 | -1.063 | -1.086 | -.915 | -1.250 |
| 64 | -.511 | -.515 | -.569 | -.431 | -.530 |
| 65 | -.522 | -.513 | -.550 | -.421 | -.583 |
| 66 | -.371 | -.384 | -.417 | -.308 | -.372 |
| 67 | -.447 | -.457 | -.498 | -.360 | -.428 |
| 68 | -.858 | -.842 | -.806 | -.744 | -.747 |
| 69 | -1.104 | -.982 | -.974 | -1.044 | -1.068 |
| 70 | -1.202 | -1.148 | -1.076 | -1.253 | -1.144 |
| 71 | -.995 | -.963 | -.974 | -1.091 | -.991 |
| 72 | -1.087 | -1.104 | -1.056 | -1.228 | -1.067 |
| 73 | -.940 | -.938 | -.941 | -1.059 | -.924 |
| 74 | -1.001 | -1.030 | -.982 | -1.151 | -.978 |
| 75 | -.942 | -.950 | -.931 | -1.060 | -.925 |
| 76 | -1.079 | -1.103 | -1.051 | -1.241 | -1.027 |
| 77 | -1.035 | -1.029 | -1.037 | -1.171 | -1.029 |
| 78 | -1.033 | -1.054 | -1.028 | -1.217 | -1.019 |
| 79 | -.825 | -.823 | -.848 | -.981 | -.864 |
| 80 | -.607 | -.607 | -.628 | -.788 | -.646 |
| 81 | -.328 | -.316 | -.360 | -.481 | -.384 |
| 82 | -.022 | -.037 | -.067 | -.182 | -.095 |
| 83 | .178 | .163 | .152 | .058 | .117 |
| 84 | .400 | .396 | .369 | .329 | .343 |

Figure 7 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .547 | .656 | .585 | .598 | .648 |
| 2 | .346 | .438 | .338 | .343 | .582 |
| 3 | .244 | .537 | .252 | .275 | .394 |
| 4 | -.202 | .050 | -.226 | -.212 | -.014 |
| 5 | -.696 | -.468 | -.745 | -.709 | -.447 |
| 6 | -.876 | -.821 | -.936 | -.902 | -.705 |
| 7 | -1.057 | -1.174 | -1.127 | -1.095 | -.962 |
| 8 | -1.118 | -1.072 | -1.114 | -1.092 | -.700 |
| 9 | -1.075 | -1.206 | -1.145 | -1.127 | -.727 |
| 10 | -1.128 | -1.096 | -1.151 | -1.165 | -.714 |
| 11 | -1.144 | -1.250 | -1.090 | -1.306 | -.973 |
| 12 | -.852 | -.870 | -.823 | -.824 | -.696 |
| 13 | -.675 | -.759 | -.666 | -.737 | -.628 |
| 14 | -1.365 | -.993 | -1.372 | -1.321 | -.930 |
| 15 | -1.180 | -1.348 | -1.213 | -1.217 | -1.068 |
| 16 | -1.506 | -1.290 | -1.493 | -1.469 | -1.136 |
| 17 | -1.445 | -1.540 | -1.504 | -1.491 | -1.243 |
| 18 | -1.245 | -1.343 | -1.209 | -1.219 | -1.264 |
| 19 | -1.516 | -1.688 | -1.575 | -1.563 | -1.381 |
| 20 | -1.546 | -1.460 | -1.503 | -1.506 | -1.140 |
| 21 | -.785 | -1.123 | -.820 | -.811 | -.876 |
| 22 | -.995 | -1.175 | -1.192 | -1.244 | -.429 |
| 23 | .260 | .033 | .298 | .277 | .062 |
| 24 | .271 | .156 | .252 | .181 | .383 |
| 25 | .661 | .663 | .688 | .706 | .468 |
| 26 | .578 | .604 | .571 | .562 | .450 |
| 27 | .260 | .471 | .254 | .267 | .305 |
| 28 | -.154 | .049 | -.157 | -.149 | -.025 |
| 29 | -.593 | -.379 | -.635 | -.624 | -.412 |
| 30 | -.869 | -.726 | -.861 | -.842 | -.658 |
| 31 | -.860 | -.950 | -.931 | -.915 | -.807 |
| 32 | -.950 | -.918 | -.942 | -.935 | -.792 |
| 33 | -.903 | -1.004 | -.969 | -.962 | -.890 |
| 34 | -1.041 | -.914 | -.997 | -1.041 | -.815 |
| 35 | -1.083 | -1.111 | -1.149 | -1.099 | -.942 |
| 36 | -.923 | -1.007 | -.870 | -.891 | -.826 |
| 37 | -.638 | -.756 | -.660 | -.670 | -.617 |
| 38 | -1.459 | -.827 | -1.429 | -1.388 | -.814 |
| 39 | -1.221 | -1.438 | -1.280 | -1.249 | -1.171 |
| 40 | -1.524 | -1.190 | -1.490 | -1.487 | -1.058 |
| 41 | -1.420 | -1.554 | -1.482 | -1.463 | -1.267 |
| 42 | -1.558 | -1.369 | -1.511 | -1.514 | -1.178 |

Figure 8 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.554 | -1.672 | -1.625 | -1.605 | -1.370 |
| 44 | -1.396 | -1.419 | -1.350 | -1.361 | -1.203 |
| 45 | -1.000 | -1.309 | -1.056 | -1.031 | -1.028 |
| 46 | -.527 | -.779 | -.515 | -.515 | -.624 |
| 47 | .028 | -.246 | .020 | .014 | -.172 |
| 48 | .406 | .192 | .408 | .385 | .174 |
| 49 | .513 | .505 | .530 | .547 | .423 |
| 50 | .521 | .520 | .521 | .514 | .406 |
| 51 | .338 | .522 | .348 | .360 | .370 |
| 52 | .118 | .274 | .114 | .122 | .196 |
| 53 | -.167 | .069 | -.175 | -.169 | -.024 |
| 54 | -.421 | -.222 | -.429 | -.416 | -.244 |
| 55 | -.669 | -.523 | -.693 | -.710 | -.510 |
| 56 | -.768 | -.680 | -.774 | -.772 | -.629 |
| 57 | -.651 | -.745 | -.677 | -.690 | -.717 |
| 58 | -.725 | -.736 | -.737 | -.729 | -.643 |
| 59 | -.559 | -.598 | -.579 | -.588 | -.664 |
| 60 | -.709 | -.714 | -.722 | -.714 | -.621 |
| 61 | -.687 | -.729 | -.723 | -.740 | -.721 |
| 62 | -.599 | -.587 | -.615 | -.591 | -.677 |
| 63 | -.881 | -.872 | -.952 | -.899 | -.806 |
| 64 | -.770 | -.785 | -.818 | -.725 | -.783 |
| 65 | -.950 | -1.071 | -1.018 | -.955 | -.852 |
| 66 | -.706 | -.866 | -.714 | -.670 | -1.086 |
| 67 | -.615 | -.762 | -.643 | -.645 | -.575 |
| 68 | -1.156 | -.617 | -1.093 | -1.144 | -.607 |
| 69 | -1.403 | -1.441 | -1.436 | -1.463 | -1.286 |
| 70 | -1.516 | -1.315 | -1.473 | -1.482 | -1.150 |
| 71 | -1.560 | -1.569 | -1.610 | -1.618 | -1.284 |
| 72 | -1.514 | -1.485 | -1.485 | -1.474 | -1.263 |
| 73 | -1.463 | -1.541 | -1.532 | -1.535 | -1.253 |
| 74 | -1.485 | -1.404 | -1.439 | -1.443 | -1.200 |
| 75 | -1.582 | -1.549 | -1.627 | -1.627 | -1.275 |
| 76 | -1.645 | -1.507 | -1.601 | -1.599 | -1.297 |
| 77 | -1.570 | -1.711 | -1.620 | -1.612 | -1.387 |
| 78 | -1.385 | -1.452 | -1.349 | -1.358 | -1.232 |
| 79 | -1.072 | -1.379 | -1.115 | -1.092 | -1.111 |
| 80 | -.738 | -.976 | -.723 | -.705 | -.817 |
| 81 | -.263 | -.466 | -.272 | -.262 | -.517 |
| 82 | .003 | -.255 | .008 | -.013 | -.215 |
| 83 | .239 | .096 | .245 | .265 | .094 |
| 84 | .471 | .321 | .472 | .474 | .261 |

Figure 8 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .525 | .655 | .659 | .660 | .642 |
| 2 | .575 | .580 | .610 | .578 | .549 |
| 3 | .278 | .359 | .428 | .336 | .340 |
| 4 | -.076 | -.058 | .018 | -.092 | -.069 |
| 5 | -.561 | -.525 | -.446 | -.551 | -.506 |
| 6 | -.828 | -.797 | -.756 | -.793 | -.752 |
| 7 | -1.095 | -1.069 | -1.066 | -1.035 | -.999 |
| 8 | -.686 | -.672 | -.642 | -.781 | -.764 |
| 9 | -.891 | -.826 | -.805 | -.736 | -.807 |
| 10 | -.781 | -.740 | -.935 | -.853 | -.774 |
| 11 | -.825 | -.818 | -.886 | -.828 | -.786 |
| 12 | -.637 | -.611 | -.704 | -.691 | -.621 |
| 13 | -.563 | -.621 | -.597 | -.566 | -.546 |
| 14 | -.881 | -.965 | -.726 | -1.079 | -.984 |
| 15 | -1.034 | -1.069 | -1.066 | -1.036 | -1.009 |
| 16 | -1.112 | -1.148 | -1.057 | -1.228 | -1.145 |
| 17 | -1.190 | -1.191 | -1.199 | -1.237 | -1.175 |
| 18 | -1.213 | -1.216 | -1.191 | -1.312 | -1.214 |
| 19 | -1.338 | -1.331 | -1.346 | -1.377 | -1.310 |
| 20 | -1.105 | -1.105 | -1.140 | -1.149 | -1.083 |
| 21 | -.829 | -.830 | -.900 | -.824 | -.800 |
| 22 | -.384 | -.392 | -.468 | -.369 | -.372 |
| 23 | .110 | .098 | .028 | .122 | .112 |
| 24 | .426 | .419 | .379 | .447 | .408 |
| 25 | .488 | .482 | .462 | .485 | .465 |
| 26 | .429 | .434 | .458 | .430 | .412 |
| 27 | .224 | .237 | .301 | .228 | .210 |
| 28 | -.160 | -.133 | -.049 | -.148 | -.156 |
| 29 | -.616 | -.585 | -.507 | -.606 | -.583 |
| 30 | -.912 | -.878 | -.831 | -.904 | -.850 |
| 31 | -1.172 | -1.132 | -1.134 | -1.079 | -1.074 |
| 32 | -1.104 | -1.068 | -1.086 | -1.055 | -1.018 |
| 33 | -1.154 | -1.123 | -1.137 | -1.092 | -1.061 |
| 34 | -.829 | -.817 | -1.000 | -.748 | -.681 |
| 35 | -.696 | -.682 | -.673 | -.702 | -.675 |
| 36 | -.519 | -.514 | -.627 | -.493 | -.466 |
| 37 | -.438 | -.430 | -.439 | -.450 | -.448 |
| 38 | -.879 | -.849 | -.752 | -.924 | -.882 |
| 39 | -1.068 | -1.114 | -1.078 | -1.079 | -1.048 |
| 40 | -1.075 | -1.051 | -1.026 | -1.145 | -1.072 |
| 41 | -1.153 | -1.164 | -1.146 | -1.192 | -1.134 |
| 42 | -1.071 | -1.063 | -1.073 | -1.152 | -1.070 |

Figure 9 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.232 | -1.232 | -1.217 | -1.293 | -1.225 |
| 44 | -1.096 | -1.097 | -1.133 | -1.155 | -1.077 |
| 45 | -.904 | -.913 | -.961 | -.927 | -.889 |
| 46 | -.521 | -.538 | -.602 | -.542 | -.507 |
| 47 | -.091 | -.094 | -.161 | -.087 | -.092 |
| 48 | .226 | .229 | .183 | .234 | .216 |
| 49 | .450 | .435 | .430 | .443 | .426 |
| 50 | .389 | .388 | .412 | .393 | .364 |
| 51 | .285 | .289 | .336 | .279 | .265 |
| 52 | .070 | .081 | .146 | .072 | .056 |
| 53 | -.232 | -.216 | -.138 | -.219 | -.222 |
| 54 | -.495 | -.476 | -.407 | -.477 | -.469 |
| 55 | -.847 | -.816 | -.754 | -.802 | -.800 |
| 56 | -1.010 | -.977 | -.947 | -.966 | -.948 |
| 57 | -1.236 | -1.196 | -1.173 | -1.148 | -1.147 |
| 58 | -1.173 | -1.142 | -1.164 | -1.082 | -1.081 |
| 59 | -1.182 | -1.142 | -1.168 | -1.078 | -1.101 |
| 60 | -1.103 | -1.076 | -1.104 | -1.032 | -1.030 |
| 61 | -1.186 | -1.140 | -1.151 | -1.090 | -1.087 |
| 62 | -1.204 | -1.156 | -1.143 | -1.164 | -1.130 |
| 63 | -1.151 | -1.087 | -1.261 | -.873 | -.874 |
| 64 | -.559 | -.557 | -.573 | -.583 | -.555 |
| 65 | -.530 | -.931 | -1.092 | -.471 | -.723 |
| 66 | -.437 | -.460 | -.506 | -.408 | -.594 |
| 67 | -.403 | -.526 | -.522 | -.412 | -.485 |
| 68 | -.617 | -.611 | -.529 | -.688 | -.712 |
| 69 | -1.077 | -1.088 | -1.087 | -1.036 | -1.002 |
| 70 | -.984 | -.972 | -1.022 | -1.066 | -.983 |
| 71 | -1.036 | -1.047 | -1.143 | -1.055 | -.982 |
| 72 | -1.002 | -1.011 | -.993 | -1.088 | -1.006 |
| 73 | -.987 | -.995 | -1.024 | -1.055 | -.980 |
| 74 | -.942 | -.945 | -.947 | -1.034 | -.950 |
| 75 | -.967 | -.982 | -.985 | -1.052 | -.981 |
| 76 | -.983 | -.995 | -.982 | -1.112 | -1.015 |
| 77 | -1.095 | -1.107 | -1.100 | -1.171 | -1.107 |
| 78 | -.981 | -.995 | -1.027 | -1.076 | -.995 |
| 79 | -.879 | -.887 | -.944 | -.910 | -.878 |
| 80 | -.627 | -.632 | -.717 | -.653 | -.625 |
| 81 | -.357 | -.367 | -.433 | -.353 | -.356 |
| 82 | -.092 | -.098 | -.162 | -.081 | -.095 |
| 83 | .186 | .165 | .116 | .194 | .183 |
| 84 | .320 | .306 | .279 | .338 | .309 |

Figure 9 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .459 | .650 | .649 | .661 | .637 |
| 2 | .571 | .583 | .600 | .600 | .545 |
| 3 | .264 | .360 | .418 | .361 | .334 |
| 4 | -.070 | -.054 | .018 | -.042 | -.070 |
| 5 | -.557 | -.533 | -.440 | -.492 | -.517 |
| 6 | -.800 | -.775 | -.714 | -.699 | -.728 |
| 7 | -1.042 | -1.018 | -.987 | -.905 | -.938 |
| 8 | -.760 | -.728 | -.693 | -.684 | -.801 |
| 9 | -1.791 | -1.219 | -1.375 | -.790 | -1.293 |
| 10 | -.984 | -.834 | -1.039 | -.829 | -.883 |
| 11 | -1.039 | -.925 | -1.041 | -.812 | -.943 |
| 12 | -.643 | -.647 | -.739 | -.696 | -.652 |
| 13 | -.601 | -.617 | -.561 | -.553 | -.552 |
| 14 | -.955 | -.965 | -.712 | -1.220 | -.959 |
| 15 | -1.070 | -1.067 | -1.064 | -1.126 | -1.003 |
| 16 | -1.142 | -1.161 | -1.034 | -1.366 | -1.121 |
| 17 | -1.212 | -1.200 | -1.192 | -1.347 | -1.170 |
| 18 | -1.221 | -1.238 | -1.174 | -1.443 | -1.193 |
| 19 | -1.347 | -1.344 | -1.329 | -1.486 | -1.304 |
| 20 | -1.112 | -1.129 | -1.124 | -1.253 | -1.062 |
| 21 | -.833 | -.842 | -.889 | -.901 | -.794 |
| 22 | -.386 | -.402 | -.466 | -.432 | -.363 |
| 23 | .102 | .096 | .036 | .088 | .109 |
| 24 | .416 | .424 | .375 | .427 | .400 |
| 25 | .485 | .473 | .454 | .499 | .469 |
| 26 | .432 | .429 | .445 | .453 | .399 |
| 27 | .242 | .239 | .298 | .245 | .220 |
| 28 | -.135 | -.133 | -.044 | -.125 | -.133 |
| 29 | -.586 | -.564 | -.507 | -.544 | -.567 |
| 30 | -.877 | -.860 | -.816 | -.817 | -.823 |
| 31 | -1.134 | -1.095 | -1.104 | -.980 | -1.059 |
| 32 | -1.072 | -1.047 | -1.047 | -.952 | -.986 |
| 33 | -1.126 | -1.086 | -1.096 | -.995 | -1.038 |
| 34 | -.716 | -.729 | -.754 | -.597 | -.812 |
| 35 | -.780 | -.792 | -.856 | -.667 | -.737 |
| 36 | -.493 | -.521 | -.555 | -.421 | -.559 |
| 37 | -.502 | -.513 | -.542 | -.444 | -.497 |
| 38 | -.829 | -.872 | -.755 | -1.015 | -.870 |
| 39 | -1.111 | -1.092 | -1.055 | -1.115 | -1.046 |
| 40 | -1.042 | -1.093 | -1.012 | -1.243 | -1.054 |
| 41 | -1.171 | -1.155 | -1.134 | -1.269 | -1.124 |
| 42 | -1.068 | -1.099 | -1.054 | -1.254 | -1.045 |

Figure 10 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.238 | -1.237 | -1.201 | -1.436 | -1.211 |
| 44 | -1.102 | -1.122 | -1.119 | -1.230 | -1.059 |
| 45 | -.911 | -.914 | -.953 | -1.000 | -.886 |
| 46 | -.521 | -.550 | -.596 | -.580 | -.510 |
| 47 | -.092 | -.112 | -.173 | .084 | -.082 |
| 48 | .231 | .210 | .164 | .402 | .217 |
| 49 | .451 | .426 | .425 | .438 | .424 |
| 50 | .396 | .393 | .406 | .410 | .363 |
| 51 | .289 | .286 | .334 | .294 | .266 |
| 52 | .074 | .084 | .144 | .099 | .061 |
| 53 | -.217 | -.197 | -.138 | -.187 | -.213 |
| 54 | -.473 | -.459 | -.394 | -.437 | -.457 |
| 55 | -.838 | -.791 | -.737 | -.770 | -.793 |
| 56 | -.993 | -.962 | -.919 | -.924 | -.934 |
| 57 | -1.228 | -1.157 | -1.168 | -1.081 | -1.143 |
| 58 | -1.172 | -1.117 | -1.158 | -1.020 | -1.071 |
| 59 | -1.179 | -1.122 | -1.169 | -1.002 | -1.094 |
| 60 | -1.100 | -1.064 | -1.091 | -.955 | -1.012 |
| 61 | -1.167 | -1.112 | -1.130 | -1.031 | -1.091 |
| 62 | -1.199 | -1.159 | -1.132 | -1.067 | -1.045 |
| 63 | -.698 | -.633 | -.735 | -.635 | -.562 |
| 64 | -.580 | -.590 | -.588 | -.535 | -.620 |
| 65 | -.444 | -.542 | -.631 | -.398 | -.473 |
| 66 | -.402 | -.439 | -.447 | -.377 | -.667 |
| 67 | -.425 | -.438 | -.439 | -.352 | -.437 |
| 68 | -.617 | -.613 | -.518 | -.692 | -.706 |
| 69 | -1.134 | -1.237 | -1.129 | -1.194 | -1.113 |
| 70 | -.999 | -1.092 | -1.044 | -1.200 | -.996 |
| 71 | -1.081 | -1.112 | -1.146 | -1.172 | -1.033 |
| 72 | -1.017 | -1.046 | -1.026 | -1.176 | -.997 |
| 73 | -1.012 | -1.020 | -1.025 | -1.127 | -.982 |
| 74 | -.955 | -.979 | -.939 | -1.109 | -.931 |
| 75 | -.988 | -.985 | -.976 | -1.133 | -.980 |
| 76 | -1.005 | -1.019 | -.963 | -1.194 | -.993 |
| 77 | -1.120 | -1.124 | -1.096 | -1.248 | -1.094 |
| 78 | -1.007 | -1.023 | -1.011 | -1.129 | -.968 |
| 79 | -.891 | -.906 | -.938 | -.964 | -.873 |
| 80 | -.626 | -.665 | -.694 | -.695 | -.613 |
| 81 | -.353 | -.379 | -.440 | -.401 | -.349 |
| 82 | -.080 | -.119 | -.160 | -.114 | -.086 |
| 83 | .191 | .164 | .121 | .166 | .170 |
| 84 | .327 | .303 | .284 | .326 | .294 |

Figure 10 (continued) – Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .682 | .676 | .661 | .681 | .647 |
| 2 | .559 | .587 | .592 | .565 | .559 |
| 3 | .644 | .334 | .328 | .290 | .309 |
| 4 | -.124 | -.056 | -.067 | -.124 | -.071 |
| 5 | -.495 | -.557 | -.559 | -.628 | -.565 |
| 6 | -.996 | -.916 | -.949 | -.998 | -.919 |
| 7 | -1.266 | -1.195 | -1.222 | -1.250 | -1.196 |
| 8 | -1.280 | -1.208 | -1.234 | -1.258 | -1.205 |
| 9 | -1.295 | -1.220 | -1.246 | -1.266 | -1.215 |
| 10 | -1.190 | -1.074 | -1.085 | -1.070 | -1.148 |
| 11 | -.761 | -.709 | -.858 | -.827 | -.902 |
| 12 | -.284 | -.280 | -.339 | -.422 | -.330 |
| 13 | -.627 | -.625 | -.631 | -.595 | -.644 |
| 14 | -.749 | -.672 | -.686 | -.824 | -.789 |
| 15 | -1.091 | -1.141 | -1.089 | -1.095 | -1.074 |
| 16 | -1.051 | -1.056 | -1.032 | -1.095 | -1.054 |
| 17 | -1.205 | -1.247 | -1.185 | -1.238 | -1.165 |
| 18 | -1.187 | -1.260 | -1.203 | -1.261 | -1.171 |
| 19 | -1.314 | -1.355 | -1.295 | -1.339 | -1.275 |
| 20 | -1.068 | -1.138 | -1.113 | -1.106 | -1.071 |
| 21 | -.813 | -.855 | -.832 | -.812 | -.806 |
| 22 | -.334 | -.375 | -.380 | -.328 | -.344 |
| 23 | .163 | .114 | .100 | .159 | .113 |
| 24 | .464 | .436 | .430 | .466 | .424 |
| 25 | .499 | .486 | .479 | .496 | .457 |
| 26 | .427 | .441 | .443 | .437 | .423 |
| 27 | .212 | .243 | .238 | .203 | .215 |
| 28 | -.163 | -.117 | -.127 | -.177 | -.130 |
| 29 | -.646 | -.577 | -.580 | -.650 | -.607 |
| 30 | -.954 | -.901 | -.932 | -.974 | -.923 |
| 31 | -1.220 | -1.129 | -1.178 | -1.189 | -1.161 |
| 32 | -1.137 | -1.071 | -1.137 | -1.141 | -1.098 |
| 33 | -1.237 | -1.141 | -1.194 | -1.195 | -1.172 |
| 34 | -1.009 | -1.007 | -1.085 | -.999 | -.946 |
| 35 | -.869 | -.808 | -.819 | -.801 | -.869 |
| 36 | -.687 | -.616 | -.659 | -.674 | -.695 |
| 37 | -.477 | -.447 | -.438 | -.434 | -.481 |
| 38 | -.778 | -.713 | -.685 | -.727 | -.771 |
| 39 | -1.079 | -1.070 | -1.092 | -1.112 | -1.059 |
| 40 | -.981 | -1.023 | -.961 | -1.004 | -.984 |
| 41 | -1.148 | -1.183 | -1.141 | -1.175 | -1.119 |
| 42 | -1.024 | -1.109 | -1.042 | -1.083 | -1.025 |

Figure 11 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 150^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.221 | -1.282 | -1.206 | -1.257 | -1.185 |
| 44 | -1.053 | -1.138 | -1.100 | -1.096 | -1.067 |
| 45 | -.897 | -.943 | -.911 | -.900 | -.897 |
| 46 | -.509 | -.550 | -.565 | -.509 | -.543 |
| 47 | -.052 | -.101 | -.106 | -.046 | -.093 |
| 48 | .251 | .217 | .208 | .265 | .214 |
| 49 | .406 | .400 | .405 | .415 | .384 |
| 50 | .428 | .435 | .429 | .423 | .408 |
| 51 | .264 | .297 | .302 | .274 | .272 |
| 52 | .081 | .117 | .119 | .076 | .091 |
| 53 | -.185 | -.137 | -.133 | -.181 | -.162 |
| 54 | -.553 | -.477 | -.485 | -.549 | -.509 |
| 55 | -.771 | -.705 | -.726 | -.774 | -.734 |
| 56 | -1.109 | -1.002 | -1.029 | -1.076 | -1.035 |
| 57 | -1.154 | -1.070 | -1.130 | -1.139 | -1.113 |
| 58 | -1.229 | -1.109 | -1.194 | -1.173 | -1.180 |
| 59 | -1.108 | -1.027 | -1.103 | -1.087 | -1.078 |
| 60 | -1.160 | -1.054 | -1.120 | -1.107 | -1.105 |
| 61 | -1.095 | -1.012 | -1.084 | -1.087 | -1.048 |
| 62 | -1.242 | -1.147 | -1.177 | -1.186 | -1.183 |
| 63 | -1.065 | -1.054 | -1.093 | -1.010 | -.989 |
| 64 | -.757 | -.693 | -.799 | -.774 | -.736 |
| 65 | -.571 | -.566 | -.625 | -.581 | -.593 |
| 66 | -.476 | -.403 | -.474 | -.478 | -.482 |
| 67 | -.411 | -.376 | -.398 | -.415 | -.421 |
| 68 | -.610 | -.521 | -.500 | -.558 | -.478 |
| 69 | -.382 | -.357 | -.382 | -.388 | -.682 |
| 70 | -1.118 | -1.295 | -1.266 | -1.158 | -1.132 |
| 71 | -.812 | -.910 | -.951 | -.883 | -.953 |
| 72 | -1.070 | -1.168 | -1.080 | -1.118 | -1.054 |
| 73 | -.941 | -1.030 | -.992 | -1.000 | -.957 |
| 74 | -.980 | -1.055 | -.964 | -1.014 | -.962 |
| 75 | -.924 | -1.013 | -.946 | -.993 | -.927 |
| 76 | -1.055 | -1.148 | -1.020 | -1.110 | -1.028 |
| 77 | -1.028 | -1.119 | -1.053 | -1.092 | -1.032 |
| 78 | -1.029 | -1.110 | -1.030 | -1.055 | -1.020 |
| 79 | -.819 | -.902 | -.872 | -.860 | -.843 |
| 80 | -.613 | -.674 | -.645 | -.605 | -.628 |
| 81 | -.325 | -.375 | -.374 | -.323 | -.353 |
| 82 | -.039 | -.097 | -.093 | -.055 | -.088 |
| 83 | .165 | .131 | .125 | .164 | .124 |
| 84 | .379 | .342 | .338 | .378 | .329 |

Figure 11 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 150^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .677 | .669 | .662 | .687 | .645 |
| 2 | .559 | .589 | .573 | .548 | .555 |
| 3 | .583 | .322 | .327 | .308 | .306 |
| 4 | -.113 | -.076 | -.068 | -.109 | -.080 |
| 5 | -.521 | -.574 | -.584 | -.595 | -.563 |
| 6 | -1.003 | -.961 | -.958 | -.916 | -.921 |
| 7 | -1.254 | -1.202 | -1.242 | -1.188 | -1.191 |
| 8 | -1.274 | -1.219 | -1.263 | -1.206 | -1.200 |
| 9 | -1.293 | -1.237 | -1.283 | -1.225 | -1.210 |
| 10 | -1.463 | -1.272 | -1.349 | -1.018 | -1.412 |
| 11 | -.922 | -.963 | -1.152 | -.969 | -1.076 |
| 12 | -.232 | -.215 | -.220 | -.470 | -.255 |
| 13 | -.714 | -.729 | -.791 | -.716 | -.721 |
| 14 | -.581 | -.630 | -.641 | -.809 | -.719 |
| 15 | -1.147 | -1.194 | -1.151 | -1.180 | -1.126 |
| 16 | -1.018 | -1.078 | -1.021 | -1.125 | -1.052 |
| 17 | -1.258 | -1.273 | -1.205 | -1.366 | -1.197 |
| 18 | -1.257 | -1.299 | -1.182 | -1.345 | -1.192 |
| 19 | -1.364 | -1.368 | -1.303 | -1.472 | -1.297 |
| 20 | -1.125 | -1.162 | -1.087 | -1.162 | -1.088 |
| 21 | -.840 | -.857 | -.845 | -.903 | -.830 |
| 22 | -.350 | -.374 | -.376 | -.366 | -.368 |
| 23 | .129 | .101 | .102 | .121 | .097 |
| 24 | .437 | .440 | .422 | .431 | .410 |
| 25 | .490 | .471 | .477 | .499 | .459 |
| 26 | .433 | .443 | .427 | .425 | .412 |
| 27 | .210 | .239 | .234 | .216 | .220 |
| 28 | -.155 | -.118 | -.132 | -.147 | -.134 |
| 29 | -.638 | -.594 | -.598 | -.614 | -.591 |
| 30 | -.942 | -.926 | -.931 | -.869 | -.906 |
| 31 | -1.186 | -1.136 | -1.196 | -1.100 | -1.136 |
| 32 | -1.110 | -1.098 | -1.128 | -1.009 | -1.082 |
| 33 | -1.209 | -1.151 | -1.212 | -1.128 | -1.170 |
| 34 | -.848 | -.867 | -.903 | -.881 | -.834 |
| 35 | -.985 | -.961 | -1.008 | -.802 | -.959 |
| 36 | -.763 | -.693 | -.770 | -.693 | -.784 |
| 37 | -.491 | -.485 | -.501 | -.474 | -.509 |
| 38 | -.724 | -.714 | -.704 | -.676 | -.734 |
| 39 | -1.116 | -1.106 | -1.150 | -1.211 | -1.111 |
| 40 | -1.012 | -1.038 | -.958 | -.999 | -.983 |
| 41 | -1.201 | -1.197 | -1.173 | -1.307 | -1.156 |
| 42 | -1.094 | -1.138 | -1.030 | -1.150 | -1.047 |

Figure 12 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 150^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.284 | -1.279 | -1.205 | -1.391 | -1.213 |
| 44 | -1.113 | -1.151 | -1.076 | -1.149 | -1.080 |
| 45 | -.940 | -.957 | -.911 | -.993 | -.904 |
| 46 | -.545 | -.582 | -.549 | -.540 | -.539 |
| 47 | -.069 | -.111 | -.106 | -.083 | -.091 |
| 48 | .244 | .215 | .201 | .234 | .209 |
| 49 | .408 | .403 | .401 | .400 | .376 |
| 50 | .428 | .426 | .427 | .438 | .408 |
| 51 | .283 | .306 | .294 | .282 | .267 |
| 52 | .091 | .108 | .107 | .112 | .098 |
| 53 | -.166 | -.142 | -.150 | -.143 | -.154 |
| 54 | -.526 | -.489 | -.500 | -.511 | -.491 |
| 55 | -.747 | -.720 | -.733 | -.691 | -.712 |
| 56 | -1.064 | -1.007 | -1.055 | -1.008 | -1.019 |
| 57 | -1.117 | -1.090 | -1.135 | -1.009 | -1.089 |
| 58 | -1.183 | -1.132 | -1.234 | -1.078 | -1.162 |
| 59 | -1.071 | -1.051 | -1.110 | -.965 | -1.059 |
| 60 | -1.111 | -1.063 | -1.147 | -1.026 | -1.092 |
| 61 | -1.060 | -1.040 | -1.085 | -.962 | -1.040 |
| 62 | -1.204 | -1.186 | -1.235 | -1.152 | -1.177 |
| 63 | -.827 | -.823 | -.889 | -.870 | -.774 |
| 64 | -.734 | -.707 | -.749 | -.708 | -.722 |
| 65 | -.536 | -.566 | -.633 | -.565 | -.582 |
| 66 | -.426 | -.444 | -.500 | -.465 | -.469 |
| 67 | -.384 | -.400 | -.438 | -.394 | -.433 |
| 68 | -.528 | -.564 | -.541 | -.486 | -.483 |
| 69 | -.395 | -.395 | -.405 | -.376 | -.718 |
| 70 | -1.339 | -1.256 | -1.282 | -1.220 | -1.229 |
| 71 | -.917 | -.925 | -.868 | -.885 | -.974 |
| 72 | -1.167 | -1.164 | -1.094 | -1.249 | -1.107 |
| 73 | -1.008 | -1.056 | -.958 | -1.054 | -.973 |
| 74 | -1.035 | -1.064 | -.965 | -1.146 | -.994 |
| 75 | -.989 | -1.051 | -.922 | -1.061 | -.949 |
| 76 | -1.131 | -1.144 | -1.033 | -1.268 | -1.062 |
| 77 | -1.092 | -1.144 | -1.036 | -1.149 | -1.050 |
| 78 | -1.088 | -1.110 | -1.036 | -1.178 | -1.046 |
| 79 | -.866 | -.923 | -.853 | -.896 | -.853 |
| 80 | -.648 | -.665 | -.645 | -.690 | -.646 |
| 81 | -.346 | -.383 | -.369 | -.356 | -.371 |
| 82 | -.077 | -.095 | .019 | -.083 | -.095 |
| 83 | .144 | .130 | .232 | .142 | .112 |
| 84 | .359 | .336 | .351 | .362 | .330 |

Figure 12 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 150^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .796 | .786 | .782 | .751 | .738 |
| 2 | .641 | .635 | .666 | .698 | .656 |
| 3 | .284 | .286 | .339 | .417 | .365 |
| 4 | -.301 | -.294 | -.237 | -.124 | -.159 |
| 5 | -.886 | -.873 | -.814 | -.664 | -.684 |
| 6 | -.810 | -.812 | -.766 | -.662 | -.676 |
| 7 | -1.106 | -1.061 | -1.131 | -.997 | -.963 |
| 8 | -.930 | -.892 | -.952 | -.845 | -.832 |
| 9 | -1.408 | -1.486 | -1.214 | -1.043 | -1.174 |
| 10 | -.999 | -.929 | -1.030 | -.882 | -.904 |
| 11 | -1.014 | -1.006 | -.923 | -.853 | -.932 |
| 12 | -.735 | -.732 | -.771 | -.734 | -.727 |
| 13 | -.812 | -.864 | -.801 | -.756 | -.802 |
| 14 | -1.376 | -1.408 | -1.331 | -1.400 | -1.350 |
| 15 | -1.398 | -1.447 | -1.402 | -1.479 | -1.426 |
| 16 | -1.605 | -1.606 | -1.592 | -1.696 | -1.594 |
| 17 | -1.635 | -1.609 | -1.618 | -1.728 | -1.612 |
| 18 | -1.629 | -1.602 | -1.598 | -1.712 | -1.589 |
| 19 | -1.835 | -1.782 | -1.814 | -1.932 | -1.779 |
| 20 | -1.513 | -1.479 | -1.517 | -1.654 | -1.522 |
| 21 | -1.209 | -1.158 | -1.236 | -1.394 | -1.093 |
| 22 | -.566 | -.545 | -.604 | -.762 | -.389 |
| 23 | .075 | .107 | .048 | -.093 | .003 |
| 24 | .495 | .505 | .486 | .395 | .452 |
| 25 | .656 | .652 | .644 | .618 | .600 |
| 26 | .541 | .526 | .558 | .577 | .549 |
| 27 | .213 | .207 | .256 | .332 | .279 |
| 28 | -.284 | -.288 | -.223 | -.127 | -.171 |
| 29 | -.963 | -.940 | -.909 | -.757 | -.774 |
| 30 | -1.319 | -1.295 | -1.279 | -1.148 | -1.153 |
| 31 | -1.596 | -1.536 | -1.566 | -1.400 | -1.395 |
| 32 | -1.450 | -1.401 | -1.424 | -1.285 | -1.279 |
| 33 | -1.648 | -1.584 | -1.622 | -1.469 | -1.452 |
| 34 | -1.176 | -.985 | -1.434 | -1.336 | -1.191 |
| 35 | -.759 | -.775 | -.750 | -.679 | -.690 |
| 36 | -.633 | -.598 | -.769 | -.696 | -.649 |
| 37 | -.664 | -.750 | -.752 | -.623 | -.695 |
| 38 | -1.373 | -1.414 | -1.440 | -1.492 | -1.383 |
| 39 | -1.502 | -1.435 | -1.516 | -1.607 | -1.484 |
| 40 | -1.443 | -1.461 | -1.477 | -1.573 | -1.488 |
| 41 | -1.479 | -1.424 | -1.466 | -1.597 | -1.454 |
| 42 | -1.304 | -1.294 | -1.317 | -1.424 | -1.329 |

Figure 13 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.531 | -1.500 | -1.503 | -1.622 | -1.506 |
| 44 | -1.338 | -1.318 | -1.359 | -1.484 | -1.378 |
| 45 | -1.141 | -1.091 | -1.150 | -1.293 | -1.160 |
| 46 | -.626 | -.595 | -.657 | -.804 | -.691 |
| 47 | -.027 | -.027 | -.066 | -.202 | -.119 |
| 48 | .371 | .356 | .343 | .245 | .301 |
| 49 | .617 | .601 | .610 | .601 | .582 |
| 50 | .598 | .572 | .606 | .624 | .585 |
| 51 | .349 | .336 | .384 | .435 | .393 |
| 52 | .014 | .000 | .053 | .155 | .109 |
| 53 | -.347 | -.342 | -.302 | -.184 | -.220 |
| 54 | -.845 | -.855 | -.795 | -.647 | -.679 |
| 55 | -1.156 | -1.158 | -1.122 | -.976 | -.998 |
| 56 | -1.619 | -1.577 | -1.573 | -1.382 | -1.396 |
| 57 | -1.686 | -1.640 | -1.667 | -1.493 | -1.506 |
| 58 | -1.679 | -1.612 | -1.675 | -1.492 | -1.461 |
| 59 | -1.572 | -1.537 | -1.561 | -1.424 | -1.407 |
| 60 | -1.705 | -1.646 | -1.681 | -1.521 | -1.510 |
| 61 | -1.626 | -1.592 | -1.608 | -1.479 | -1.472 |
| 62 | -1.946 | -1.709 | -1.708 | -1.563 | -1.553 |
| 63 | -1.633 | -1.626 | -1.619 | -1.482 | -1.725 |
| 64 | -.909 | -.772 | -.927 | -.934 | -.788 |
| 65 | -.804 | -.811 | -.770 | -.732 | -.784 |
| 66 | -.728 | -.708 | -.742 | -.700 | -.668 |
| 67 | -.841 | -.893 | -.851 | -.797 | -.808 |
| 68 | -1.066 | -1.162 | -1.163 | -1.099 | -1.111 |
| 69 | -.958 | -1.020 | -1.040 | -1.036 | -1.029 |
| 70 | -1.030 | -1.052 | -1.092 | -1.156 | -1.076 |
| 71 | -.935 | -.939 | -.911 | -1.022 | -.931 |
| 72 | -1.055 | -1.063 | -1.077 | -1.201 | -1.104 |
| 73 | -.982 | -.980 | -.946 | -1.085 | -1.002 |
| 74 | -1.066 | -1.049 | -1.059 | -1.187 | -1.082 |
| 75 | -.994 | -.986 | -.976 | -1.097 | -1.025 |
| 76 | -1.104 | -1.095 | -1.076 | -1.215 | -1.117 |
| 77 | -1.121 | -1.115 | -1.100 | -1.238 | -1.157 |
| 78 | -1.138 | -1.119 | -1.130 | -1.280 | -1.163 |
| 79 | -.918 | -.898 | -.920 | -1.071 | -.971 |
| 80 | -.648 | -.626 | -.674 | -.840 | -.728 |
| 81 | -.303 | -.290 | -.324 | -.473 | -.392 |
| 82 | .076 | .072 | .024 | -.094 | -.036 |
| 83 | .330 | .318 | .293 | .206 | .245 |
| 84 | .614 | .600 | .587 | .537 | .529 |

Figure 13 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .791 | .795 | .785 | .753 | .753 |
| 2 | .636 | .654 | .679 | .698 | .668 |
| 3 | .297 | .284 | .353 | .483 | .374 |
| 4 | -.239 | -.263 | -.181 | .009 | -.134 |
| 5 | -.776 | -.809 | -.715 | -.466 | -.642 |
| 6 | -.641 | -.691 | -.615 | -.411 | -.577 |
| 7 | -.963 | -1.151 | -.940 | -.754 | -.904 |
| 8 | -1.416 | -1.667 | -1.282 | -.640 | -1.283 |
| 9 | -1.641 | -1.726 | -1.970 | -1.048 | -1.707 |
| 10 | -1.328 | -1.352 | -1.244 | -.775 | -1.377 |
| 11 | -1.087 | -1.100 | -1.189 | -.859 | -1.199 |
| 12 | -.814 | -.844 | -.834 | -.686 | -.869 |
| 13 | -.833 | -.898 | -.848 | -.795 | -.864 |
| 14 | -1.471 | -1.462 | -1.363 | -1.494 | -1.344 |
| 15 | -1.498 | -1.479 | -1.431 | -1.565 | -1.437 |
| 16 | -1.617 | -1.630 | -1.613 | -1.800 | -1.569 |
| 17 | -1.652 | -1.603 | -1.623 | -1.881 | -1.593 |
| 18 | -1.592 | -1.602 | -1.616 | -1.831 | -1.547 |
| 19 | -1.814 | -1.767 | -1.803 | -2.112 | -1.754 |
| 20 | -1.491 | -1.491 | -1.540 | -1.743 | -1.497 |
| 21 | -1.205 | -1.153 | -1.239 | -1.507 | -1.235 |
| 22 | -.566 | -.543 | -.627 | -.795 | -.646 |
| 23 | .064 | .097 | .041 | -.152 | -.012 |
| 24 | .483 | .517 | .479 | .352 | .424 |
| 25 | .650 | .645 | .645 | .620 | .617 |
| 26 | .534 | .545 | .560 | .563 | .546 |
| 27 | .222 | .218 | .261 | .357 | .269 |
| 28 | -.268 | -.268 | -.221 | -.101 | -.203 |
| 29 | -.913 | -.919 | -.850 | -.692 | -.801 |
| 30 | -1.247 | -1.288 | -1.224 | -1.032 | -1.170 |
| 31 | -1.479 | -1.480 | -1.452 | -1.258 | -1.390 |
| 32 | -1.298 | -1.351 | -1.300 | -1.091 | -1.236 |
| 33 | -1.564 | -1.545 | -1.531 | -1.348 | -1.472 |
| 34 | -.764 | -.816 | -.817 | -.789 | -.774 |
| 35 | -.799 | -.849 | -.822 | -.668 | -.808 |
| 36 | -.499 | -.555 | -.521 | -.467 | -.509 |
| 37 | -.759 | -.809 | -.735 | -.661 | -.736 |
| 38 | -1.405 | -1.471 | -1.457 | -1.465 | -1.367 |
| 39 | -1.429 | -1.412 | -1.519 | -1.657 | -1.466 |
| 40 | -1.500 | -1.506 | -1.523 | -1.604 | -1.477 |
| 41 | -1.453 | -1.408 | -1.480 | -1.687 | -1.441 |
| 42 | -1.338 | -1.318 | -1.342 | -1.476 | -1.304 |

Figure 14 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.542 | -1.485 | -1.513 | -1.754 | -1.477 |
| 44 | -1.355 | -1.348 | -1.388 | -1.547 | -1.362 |
| 45 | -1.158 | -1.089 | -1.169 | -1.379 | -1.149 |
| 46 | -.647 | -.609 | -.694 | -.843 | -.691 |
| 47 | -.064 | -.018 | -.112 | -.238 | -.133 |
| 48 | .329 | .379 | .306 | .214 | .277 |
| 49 | .604 | .621 | .619 | .581 | .589 |
| 50 | .597 | .579 | .599 | .633 | .594 |
| 51 | .361 | .351 | .395 | .435 | .397 |
| 52 | .026 | .017 | .061 | .163 | .100 |
| 53 | -.325 | -.333 | -.285 | -.164 | -.230 |
| 54 | -.816 | -.824 | -.750 | -.614 | -.699 |
| 55 | -1.118 | -1.148 | -1.081 | -.907 | -1.014 |
| 56 | -1.547 | -1.553 | -1.508 | -1.387 | -1.468 |
| 57 | -1.617 | -1.660 | -1.622 | -1.454 | -1.578 |
| 58 | -1.607 | -1.611 | -1.616 | -1.486 | -1.559 |
| 59 | -1.496 | -1.543 | -1.525 | -1.352 | -1.463 |
| 60 | -1.630 | -1.609 | -1.622 | -1.513 | -1.582 |
| 61 | -1.546 | -1.576 | -1.556 | -1.414 | -1.505 |
| 62 | -1.937 | -1.659 | -1.649 | -1.560 | -1.631 |
| 63 | -1.551 | -1.604 | -1.581 | -1.421 | -1.770 |
| 64 | -.762 | -.717 | -.689 | -.765 | -.687 |
| 65 | -.747 | -.803 | -.767 | -.775 | -.800 |
| 66 | -.669 | -.685 | -.637 | -.734 | -.626 |
| 67 | -.866 | -.890 | -.817 | -.870 | -.802 |
| 68 | -1.224 | -1.233 | -1.144 | -1.159 | -1.136 |
| 69 | -1.068 | -1.125 | -1.070 | -1.074 | -1.054 |
| 70 | -1.167 | -1.139 | -1.136 | -1.181 | -1.161 |
| 71 | -.984 | -.984 | -.975 | -1.079 | -.976 |
| 72 | -1.100 | -1.110 | -1.123 | -1.248 | -1.091 |
| 73 | -1.005 | -1.031 | -1.014 | -1.141 | -1.001 |
| 74 | -1.119 | -1.071 | -1.106 | -1.252 | -1.077 |
| 75 | -1.030 | -1.034 | -1.037 | -1.153 | -1.015 |
| 76 | -1.156 | -1.097 | -1.143 | -1.288 | -1.111 |
| 77 | -1.153 | -1.139 | -1.173 | -1.282 | -1.139 |
| 78 | -1.171 | -1.148 | -1.178 | -1.345 | -1.170 |
| 79 | -.936 | -.953 | -.981 | -1.107 | -.976 |
| 80 | -.678 | -.644 | -.706 | -.880 | -.721 |
| 81 | -.321 | -.312 | -.352 | -.497 | -.382 |
| 82 | .039 | .071 | .008 | -.116 | -.028 |
| 83 | .297 | .334 | .292 | .189 | .247 |
| 84 | .600 | .601 | .578 | .525 | .549 |

Figure 14 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .979 | .771 | .671 | .753 | .721 |
| 2 | .541 | .542 | .423 | .405 | .704 |
| 3 | .366 | .610 | .300 | .352 | .453 |
| 4 | -.251 | .030 | -.274 | -.251 | -.038 |
| 5 | -.887 | -.620 | -.878 | -.904 | -.611 |
| 6 | -1.064 | -.974 | -1.046 | -1.079 | -.859 |
| 7 | -1.242 | -1.328 | -1.214 | -1.254 | -1.108 |
| 8 | -1.181 | -1.256 | -1.311 | -1.218 | -.808 |
| 9 | -1.274 | -1.393 | -1.267 | -1.307 | -.871 |
| 10 | -1.293 | -1.278 | -1.340 | -1.345 | -.860 |
| 11 | -1.508 | -1.528 | -1.433 | -1.523 | -1.237 |
| 12 | -1.108 | -1.164 | -1.146 | -1.161 | -1.069 |
| 13 | -.979 | -.939 | -.996 | -.979 | -.806 |
| 14 | -1.913 | -1.715 | -2.059 | -1.983 | -1.740 |
| 15 | -1.839 | -1.987 | -1.772 | -1.857 | -1.658 |
| 16 | -2.055 | -1.981 | -2.232 | -2.153 | -1.888 |
| 17 | -2.238 | -2.230 | -2.105 | -2.263 | -1.897 |
| 18 | -1.770 | -1.999 | -1.922 | -1.858 | -1.924 |
| 19 | -2.407 | -2.438 | -2.251 | -2.437 | -2.089 |
| 20 | -2.068 | -2.038 | -2.241 | -2.164 | -1.817 |
| 21 | -1.411 | -1.740 | -1.310 | -1.452 | -1.435 |
| 22 | -1.554 | -1.168 | -1.766 | -1.667 | -.828 |
| 23 | .162 | -.260 | .134 | .152 | -.154 |
| 24 | .160 | .172 | .161 | .173 | .346 |
| 25 | .857 | .733 | .799 | .861 | .552 |
| 26 | .646 | .758 | .726 | .702 | .604 |
| 27 | .421 | .610 | .388 | .419 | .431 |
| 28 | -.101 | .157 | -.095 | -.082 | .029 |
| 29 | -.607 | -.366 | -.585 | -.603 | -.425 |
| 30 | -.733 | -.700 | -.804 | -.744 | -.690 |
| 31 | -.862 | -.865 | -.832 | -.869 | -.772 |
| 32 | -.823 | -.881 | -.894 | -.843 | -.818 |
| 33 | -.901 | -.950 | -.861 | -.914 | -.936 |
| 34 | -.905 | -.861 | -1.013 | -.925 | -.868 |
| 35 | -1.037 | -1.113 | -1.035 | -1.050 | -.976 |
| 36 | -.960 | -1.051 | -1.078 | -.991 | -.982 |
| 37 | -.967 | -.880 | -.915 | -.991 | -.761 |
| 38 | -2.006 | -1.688 | -2.243 | -2.109 | -1.703 |
| 39 | -1.950 | -2.033 | -1.829 | -1.991 | -1.731 |
| 40 | -2.152 | -2.016 | -2.346 | -2.263 | -1.931 |
| 41 | -2.282 | -2.284 | -2.140 | -2.320 | -1.951 |
| 42 | -2.138 | -2.042 | -2.324 | -2.253 | -1.914 |

Figure 15 – Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -2.498 | -2.452 | -2.349 | -2.536 | -2.124 |
| 44 | -1.926 | -2.092 | -2.091 | -2.037 | -1.928 |
| 45 | -1.684 | -1.975 | -1.594 | -1.694 | -1.624 |
| 46 | -.854 | -1.239 | -.903 | -.878 | -1.058 |
| 47 | -.132 | -.488 | -.145 | -.146 | -.386 |
| 48 | .358 | .138 | .421 | .373 | .139 |
| 49 | .738 | .626 | .672 | .742 | .573 |
| 50 | .645 | .690 | .734 | .684 | .597 |
| 51 | .603 | .740 | .537 | .605 | .562 |
| 52 | .259 | .466 | .295 | .276 | .372 |
| 53 | -.019 | .239 | -.032 | -.009 | .110 |
| 54 | -.312 | -.086 | -.310 | -.293 | -.149 |
| 55 | -.588 | -.433 | -.562 | -.591 | -.447 |
| 56 | -.568 | -.569 | -.595 | -.576 | -.561 |
| 57 | -.485 | -.580 | -.503 | -.520 | -.559 |
| 58 | -.517 | -.562 | -.564 | -.536 | -.540 |
| 59 | -.454 | -.486 | -.445 | -.462 | -.537 |
| 60 | -.529 | -.548 | -.561 | -.536 | -.521 |
| 61 | -.533 | -.589 | -.512 | -.544 | -.612 |
| 62 | -.453 | -.443 | -.482 | -.458 | -.580 |
| 63 | -.634 | -.654 | -.632 | -.652 | -.598 |
| 64 | -.564 | -.579 | -.608 | -.568 | -.650 |
| 65 | -.754 | -.803 | -.759 | -.784 | -.649 |
| 66 | -.633 | -.701 | -.692 | -.640 | -.919 |
| 67 | -1.590 | -.856 | -1.491 | -1.584 | -.758 |
| 68 | -2.024 | -1.638 | -2.231 | -2.145 | -1.729 |
| 69 | -2.490 | -2.356 | -2.323 | -2.493 | -2.178 |
| 70 | -2.286 | -2.189 | -2.502 | -2.416 | -2.078 |
| 71 | -2.504 | -2.530 | -2.361 | -2.555 | -2.172 |
| 72 | -2.128 | -2.171 | -2.319 | -2.239 | -2.005 |
| 73 | -2.400 | -2.396 | -2.243 | -2.453 | -2.036 |
| 74 | -2.091 | -2.093 | -2.260 | -2.199 | -1.955 |
| 75 | -2.580 | -2.384 | -2.444 | -2.647 | -2.089 |
| 76 | -2.250 | -2.273 | -2.473 | -2.391 | -2.130 |
| 77 | -2.479 | -2.550 | -2.336 | -2.531 | -2.193 |
| 78 | -1.878 | -2.126 | -2.044 | -1.991 | -1.980 |
| 79 | -1.691 | -2.047 | -1.568 | -1.717 | -1.615 |
| 80 | -1.048 | -1.456 | -1.094 | -1.099 | -1.118 |
| 81 | -.397 | -.705 | -.390 | -.406 | -.817 |
| 82 | -.081 | -.461 | -.075 | -.082 | -.357 |
| 83 | .329 | .081 | .310 | .342 | .065 |
| 84 | .546 | .377 | .626 | .584 | .350 |

Figure 15 (continued) – Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .385 | .781 | .766 | .773 | .763 |
| 2 | .637 | .661 | .694 | .663 | .661 |
| 3 | .163 | .346 | .438 | .318 | .332 |
| 4 | -.251 | -.216 | -.102 | -.236 | -.222 |
| 5 | -.944 | -.880 | -.784 | -.864 | -.820 |
| 6 | -1.264 | -1.222 | -1.173 | -1.139 | -1.101 |
| 7 | -1.584 | -1.565 | -1.561 | -1.414 | -1.382 |
| 8 | -1.154 | -1.155 | -1.094 | -1.099 | -1.085 |
| 9 | -1.011 | -.978 | -1.124 | -.936 | -.894 |
| 10 | -.987 | -1.017 | -.903 | -.845 | -.839 |
| 11 | -1.022 | -.990 | -1.149 | -.988 | -.893 |
| 12 | -.819 | -.837 | -.804 | -.740 | -.748 |
| 13 | -.805 | -.830 | -.806 | -.677 | -.737 |
| 14 | -1.395 | -1.447 | -1.300 | -1.859 | -1.481 |
| 15 | -1.483 | -1.515 | -1.477 | -1.514 | -1.472 |
| 16 | -1.552 | -1.594 | -1.562 | -1.764 | -1.656 |
| 17 | -1.587 | -1.614 | -1.638 | -1.718 | -1.635 |
| 18 | -1.496 | -1.532 | -1.544 | -1.722 | -1.625 |
| 19 | -1.797 | -1.782 | -1.822 | -1.922 | -1.808 |
| 20 | -1.429 | -1.476 | -1.537 | -1.598 | -1.518 |
| 21 | -1.155 | -1.174 | -1.297 | -1.236 | -1.174 |
| 22 | -.570 | -.588 | -.719 | -.624 | -.597 |
| 23 | .053 | .043 | -.074 | .037 | .047 |
| 24 | .465 | .463 | .392 | .475 | .468 |
| 25 | .635 | .647 | .621 | .634 | .634 |
| 26 | .541 | .561 | .580 | .565 | .556 |
| 27 | .260 | .267 | .358 | .256 | .259 |
| 28 | -.253 | -.249 | -.131 | -.242 | -.247 |
| 29 | -.931 | -.892 | -.796 | -.855 | -.835 |
| 30 | -1.286 | -1.258 | -1.183 | -1.187 | -1.201 |
| 31 | -1.592 | -1.576 | -1.578 | -1.392 | -1.405 |
| 32 | -1.496 | -1.485 | -1.456 | -1.359 | -1.392 |
| 33 | -1.629 | -1.606 | -1.612 | -1.479 | -1.459 |
| 34 | -1.509 | -1.513 | -1.604 | -1.252 | -1.342 |
| 35 | -.856 | -.825 | -.835 | -.822 | -.773 |
| 36 | -.805 | -.857 | -.898 | -.730 | -.754 |
| 37 | -.617 | -.631 | -.563 | -.669 | -.651 |
| 38 | -1.253 | -1.304 | -1.256 | -1.340 | -1.307 |
| 39 | -1.325 | -1.401 | -1.412 | -1.442 | -1.402 |
| 40 | -1.367 | -1.414 | -1.372 | -1.571 | -1.486 |
| 41 | -1.390 | -1.430 | -1.468 | -1.554 | -1.453 |
| 42 | -1.250 | -1.272 | -1.293 | -1.468 | -1.368 |

Figure 16 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.441 | -1.469 | -1.456 | -1.667 | -1.546 |
| 44 | -1.312 | -1.348 | -1.394 | -1.493 | -1.415 |
| 45 | -1.109 | -1.141 | -1.209 | -1.232 | -1.150 |
| 46 | -.628 | -.662 | -.749 | -.711 | -.672 |
| 47 | -.083 | -.091 | -.205 | -.115 | -.099 |
| 48 | .326 | .328 | .240 | .321 | .315 |
| 49 | .651 | .650 | .650 | .657 | .636 |
| 50 | .550 | .560 | .594 | .564 | .566 |
| 51 | .400 | .401 | .473 | .409 | .388 |
| 52 | .098 | .090 | .181 | .113 | .102 |
| 53 | -.331 | -.322 | -.206 | -.308 | -.309 |
| 54 | -.682 | -.681 | -.571 | -.649 | -.659 |
| 55 | -1.204 | -1.185 | -1.083 | -1.120 | -1.114 |
| 56 | -1.392 | -1.390 | -1.304 | -1.292 | -1.328 |
| 57 | -1.740 | -1.704 | -1.654 | -1.494 | -1.571 |
| 58 | -1.575 | -1.567 | -1.541 | -1.343 | -1.448 |
| 59 | -1.634 | -1.606 | -1.623 | -1.421 | -1.483 |
| 60 | -1.511 | -1.491 | -1.494 | -1.330 | -1.405 |
| 61 | -1.689 | -1.655 | -1.649 | -1.493 | -1.514 |
| 62 | -1.688 | -1.660 | -1.570 | -1.545 | -1.623 |
| 63 | -2.071 | -2.060 | -2.016 | -1.735 | -1.813 |
| 64 | -1.240 | -1.186 | -1.436 | -1.103 | -1.044 |
| 65 | -.943 | -1.781 | -1.808 | -.874 | -1.599 |
| 66 | -1.035 | -.999 | -1.261 | -1.160 | -1.098 |
| 67 | -.800 | -.968 | -1.034 | -.834 | -1.042 |
| 68 | -1.063 | -1.119 | -1.103 | -1.281 | -1.196 |
| 69 | -1.051 | -1.166 | -1.175 | -1.262 | -1.133 |
| 70 | -1.007 | -1.077 | -1.034 | -1.289 | -1.133 |
| 71 | -1.001 | -1.064 | -1.107 | -1.201 | -1.064 |
| 72 | -1.001 | -1.030 | -1.028 | -1.226 | -1.110 |
| 73 | -1.036 | -1.072 | -1.093 | -1.237 | -1.100 |
| 74 | -.976 | -1.005 | -1.017 | -1.202 | -1.081 |
| 75 | -1.046 | -1.061 | -1.091 | -1.245 | -1.115 |
| 76 | -1.000 | -1.020 | -1.030 | -1.271 | -1.136 |
| 77 | -1.194 | -1.217 | -1.217 | -1.416 | -1.277 |
| 78 | -1.068 | -1.094 | -1.136 | -1.265 | -1.170 |
| 79 | -.989 | -1.015 | -1.081 | -1.116 | -1.026 |
| 80 | -.677 | -.703 | -.790 | -.778 | -.717 |
| 81 | -.348 | -.371 | -.459 | -.390 | -.351 |
| 82 | -.009 | -.026 | -.110 | -.035 | -.012 |
| 83 | .332 | .320 | .387 | .331 | .327 |
| 84 | .490 | .485 | .559 | .501 | .503 |

Figure 16 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .413 | .769 | .902 | .776 | .752 |
| 2 | .657 | .654 | .695 | .650 | .633 |
| 3 | .181 | .326 | .436 | .365 | .315 |
| 4 | -.245 | -.240 | -.129 | -.165 | -.228 |
| 5 | -.907 | -.871 | -.795 | -.765 | -.842 |
| 6 | -1.209 | -1.178 | -1.124 | -.999 | -1.088 |
| 7 | -1.511 | -1.486 | -1.454 | -1.233 | -1.335 |
| 8 | -1.122 | -1.123 | -1.065 | -.952 | -1.051 |
| 9 | -1.364 | -1.332 | -1.287 | -.798 | -1.352 |
| 10 | -1.079 | -1.153 | -1.370 | -.804 | -1.180 |
| 11 | -1.176 | -1.180 | -1.326 | -.848 | -1.185 |
| 12 | -.857 | -.875 | -.987 | -.706 | -.851 |
| 13 | -.849 | -.871 | -.857 | -.728 | -.840 |
| 14 | -1.502 | -1.454 | -1.288 | -1.625 | -1.378 |
| 15 | -1.517 | -1.469 | -1.474 | -1.561 | -1.434 |
| 16 | -1.631 | -1.584 | -1.536 | -1.833 | -1.560 |
| 17 | -1.617 | -1.556 | -1.613 | -1.864 | -1.564 |
| 18 | -1.540 | -1.492 | -1.521 | -1.855 | -1.530 |
| 19 | -1.776 | -1.704 | -1.793 | -2.082 | -1.736 |
| 20 | -1.482 | -1.429 | -1.520 | -1.689 | -1.442 |
| 21 | -1.177 | -1.160 | -1.276 | -1.342 | -1.127 |
| 22 | -.591 | -.591 | -.702 | -.685 | -.565 |
| 23 | .057 | .050 | -.053 | -.007 | .053 |
| 24 | .481 | .467 | .402 | .430 | .458 |
| 25 | .648 | .644 | .631 | .644 | .616 |
| 26 | .559 | .555 | .584 | .546 | .532 |
| 27 | .265 | .268 | .354 | .293 | .258 |
| 28 | -.252 | -.241 | -.133 | -.207 | -.229 |
| 29 | -.919 | -.889 | -.780 | -.805 | -.844 |
| 30 | -1.279 | -1.247 | -1.170 | -1.105 | -1.183 |
| 31 | -1.575 | -1.562 | -1.547 | -1.301 | -1.414 |
| 32 | -1.490 | -1.482 | -1.433 | -1.259 | -1.364 |
| 33 | -1.595 | -1.589 | -1.539 | -1.360 | -1.443 |
| 34 | -1.079 | -1.044 | -1.085 | -.947 | -.879 |
| 35 | -.919 | -.963 | -.989 | -.758 | -.958 |
| 36 | -.647 | -.663 | -.700 | -.585 | -.583 |
| 37 | -.700 | -.708 | -.661 | -.749 | -.708 |
| 38 | -1.349 | -1.334 | -1.235 | -1.413 | -1.344 |
| 39 | -1.388 | -1.347 | -1.413 | -1.530 | -1.394 |
| 40 | -1.424 | -1.383 | -1.357 | -1.593 | -1.457 |
| 41 | -1.435 | -1.390 | -1.457 | -1.639 | -1.424 |
| 42 | -1.290 | -1.247 | -1.277 | -1.497 | -1.302 |

Figure 17 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.482 | -1.433 | -1.448 | -1.777 | -1.481 |
| 44 | -1.362 | -1.344 | -1.385 | -1.525 | -1.346 |
| 45 | -1.147 | -1.123 | -1.215 | -1.309 | -1.119 |
| 46 | -.665 | -.658 | -.761 | -.738 | -.647 |
| 47 | -.112 | -.080 | -.206 | -.156 | -.083 |
| 48 | .315 | .329 | .245 | .279 | .320 |
| 49 | .660 | .639 | .640 | .654 | .623 |
| 50 | .568 | .562 | .582 | .548 | .541 |
| 51 | .410 | .408 | .479 | .418 | .381 |
| 52 | .104 | .118 | .204 | .110 | .094 |
| 53 | -.327 | -.304 | -.206 | -.289 | -.311 |
| 54 | -.686 | -.665 | -.564 | -.615 | -.649 |
| 55 | -1.206 | -1.182 | -1.078 | -1.097 | -1.240 |
| 56 | -1.415 | -1.390 | -1.302 | -1.247 | -1.418 |
| 57 | -1.721 | -1.696 | -1.654 | -1.490 | -1.581 |
| 58 | -1.588 | -1.572 | -1.570 | -1.307 | -1.442 |
| 59 | -1.652 | -1.590 | -1.612 | -1.427 | -1.433 |
| 60 | -1.536 | -1.500 | -1.492 | -1.297 | -1.360 |
| 61 | -1.684 | -1.630 | -1.626 | -1.459 | -1.514 |
| 62 | -1.742 | -1.681 | -1.621 | -1.541 | -1.598 |
| 63 | -1.796 | -1.628 | -1.739 | -1.509 | -1.258 |
| 64 | -.904 | -.892 | -.914 | -.881 | -.799 |
| 65 | -.877 | -1.454 | -1.624 | -.817 | -1.061 |
| 66 | -.632 | -.678 | -.694 | -.720 | -.800 |
| 67 | -.849 | -.841 | -.947 | -.951 | -.856 |
| 68 | -1.079 | -1.105 | -.962 | -1.155 | -1.107 |
| 69 | -1.222 | -1.202 | -1.266 | -1.228 | -1.199 |
| 70 | -1.139 | -1.126 | -1.114 | -1.170 | -1.166 |
| 71 | -1.087 | -1.127 | -1.174 | -1.252 | -1.122 |
| 72 | -1.035 | -1.036 | -1.052 | -1.212 | -1.085 |
| 73 | -1.077 | -1.075 | -1.111 | -1.304 | -1.097 |
| 74 | -1.024 | -1.021 | -1.025 | -1.205 | -1.049 |
| 75 | -1.079 | -1.070 | -1.103 | -1.285 | -1.101 |
| 76 | -1.047 | -1.045 | -1.042 | -1.264 | -1.099 |
| 77 | -1.237 | -1.182 | -1.239 | -1.466 | -1.236 |
| 78 | -1.114 | -1.081 | -1.153 | -1.261 | -1.123 |
| 79 | -1.019 | -1.016 | -1.092 | -1.152 | -1.007 |
| 80 | -.707 | -.709 | -.804 | -.763 | -.696 |
| 81 | -.364 | -.386 | -.481 | -.424 | -.365 |
| 82 | -.023 | -.044 | -.131 | -.056 | -.027 |
| 83 | .340 | .319 | .235 | .317 | .308 |
| 84 | .507 | .490 | .431 | .477 | .475 |

Figure 17 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .801 | .782 | .790 | .799 | .773 |
| 2 | .624 | .657 | .656 | .630 | .631 |
| 3 | .566 | .284 | .287 | .219 | .274 |
| 4 | -.324 | -.243 | -.253 | -.336 | -.253 |
| 5 | -.919 | -.932 | -.962 | -1.057 | -.946 |
| 6 | -1.505 | -1.394 | -1.432 | -1.485 | -1.385 |
| 7 | -1.821 | -1.690 | -1.829 | -1.782 | -1.717 |
| 8 | -1.838 | -1.719 | -1.848 | -1.787 | -1.732 |
| 9 | -1.855 | -1.748 | -1.867 | -1.792 | -1.746 |
| 10 | -1.574 | -1.472 | -1.569 | -1.447 | -1.410 |
| 11 | -.945 | -.891 | -1.016 | -.893 | -1.046 |
| 12 | -.363 | -.339 | -.441 | -.442 | -.414 |
| 13 | -.793 | -.704 | -.763 | -.702 | -.782 |
| 14 | -1.266 | -1.074 | -1.139 | -1.378 | -1.276 |
| 15 | -1.380 | -1.606 | -1.493 | -1.464 | -1.371 |
| 16 | -1.504 | -1.460 | -1.420 | -1.570 | -1.460 |
| 17 | -1.547 | -1.664 | -1.570 | -1.621 | -1.501 |
| 18 | -1.488 | -1.550 | -1.446 | -1.569 | -1.421 |
| 19 | -1.719 | -1.767 | -1.689 | -1.773 | -1.657 |
| 20 | -1.402 | -1.486 | -1.412 | -1.448 | -1.370 |
| 21 | -1.125 | -1.181 | -1.153 | -1.136 | -1.113 |
| 22 | -.502 | -.559 | -.534 | -.483 | -.502 |
| 23 | .123 | .066 | .075 | .123 | .080 |
| 24 | .515 | .486 | .499 | .531 | .484 |
| 25 | .669 | .634 | .655 | .663 | .635 |
| 26 | .548 | .564 | .559 | .537 | .540 |
| 27 | .192 | .253 | .241 | .205 | .237 |
| 28 | -.329 | -.240 | -.275 | -.319 | -.255 |
| 29 | -1.017 | -.860 | -.951 | -.986 | -.883 |
| 30 | -1.404 | -1.263 | -1.380 | -1.380 | -1.276 |
| 31 | -1.717 | -1.528 | -1.708 | -1.627 | -1.611 |
| 32 | -1.609 | -1.475 | -1.611 | -1.567 | -1.505 |
| 33 | -1.762 | -1.578 | -1.760 | -1.684 | -1.648 |
| 34 | -1.476 | -1.354 | -1.543 | -1.454 | -1.389 |
| 35 | -1.184 | -1.104 | -1.233 | -1.188 | -1.127 |
| 36 | -.879 | -.799 | -.940 | -.907 | -.862 |
| 37 | -.653 | -.593 | -.665 | -.657 | -.631 |
| 38 | -1.304 | -1.164 | -1.180 | -1.267 | -1.173 |
| 39 | -1.403 | -1.480 | -1.367 | -1.569 | -1.351 |
| 40 | -1.319 | -1.332 | -1.247 | -1.338 | -1.240 |
| 41 | -1.386 | -1.486 | -1.379 | -1.447 | -1.346 |
| 42 | -1.192 | -1.296 | -1.194 | -1.277 | -1.161 |

Figure 18 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 150^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.404 | -1.495 | -1.365 | -1.492 | -1.353 |
| 44 | -1.263 | -1.380 | -1.282 | -1.324 | -1.247 |
| 45 | -1.058 | -1.146 | -1.075 | -1.115 | -1.059 |
| 46 | -.575 | -.679 | -.631 | -.634 | -.612 |
| 47 | -.003 | -.072 | -.046 | -.035 | -.069 |
| 48 | .378 | .330 | .345 | .341 | .313 |
| 49 | .605 | .592 | .602 | .599 | .573 |
| 50 | .607 | .613 | .612 | .607 | .600 |
| 51 | .372 | .409 | .402 | .372 | .381 |
| 52 | .077 | .148 | .118 | .082 | .122 |
| 53 | -.298 | -.208 | -.243 | -.280 | -.233 |
| 54 | -.863 | -.703 | -.796 | -.819 | -.738 |
| 55 | -1.162 | -1.015 | -1.114 | -1.103 | -1.034 |
| 56 | -1.635 | -1.413 | -1.580 | -1.555 | -1.483 |
| 57 | -1.670 | -1.474 | -1.665 | -1.574 | -1.534 |
| 58 | -1.750 | -1.504 | -1.783 | -1.618 | -1.651 |
| 59 | -1.600 | -1.414 | -1.623 | -1.500 | -1.499 |
| 60 | -1.680 | -1.447 | -1.695 | -1.567 | -1.564 |
| 61 | -1.605 | -1.432 | -1.616 | -1.522 | -1.488 |
| 62 | -1.769 | -1.549 | -1.772 | -1.695 | -1.691 |
| 63 | -1.641 | -1.452 | -1.681 | -1.564 | -1.540 |
| 64 | -1.481 | -1.321 | -1.583 | -1.486 | -1.353 |
| 65 | -1.045 | -.950 | -1.106 | -1.007 | -.970 |
| 66 | -.957 | -.804 | -.914 | -.936 | -.896 |
| 67 | -.766 | -.664 | -.781 | -.787 | -.732 |
| 68 | -.950 | -.946 | -.949 | -1.087 | -.798 |
| 69 | -.795 | -.690 | -.843 | -.791 | -.968 |
| 70 | -1.101 | -1.237 | -1.217 | -1.314 | -1.135 |
| 71 | -.942 | -1.037 | -1.009 | -1.043 | -1.034 |
| 72 | -1.066 | -1.243 | -1.068 | -1.205 | -1.089 |
| 73 | -.977 | -1.147 | -.983 | -1.065 | -.993 |
| 74 | -1.013 | -1.158 | -.989 | -1.108 | -1.011 |
| 75 | -.948 | -1.099 | -.934 | -1.031 | -.946 |
| 76 | -1.039 | -1.182 | -.989 | -1.148 | -1.030 |
| 77 | -1.076 | -1.229 | -1.042 | -1.171 | -1.073 |
| 78 | -1.103 | -1.236 | -1.096 | -1.174 | -1.106 |
| 79 | -.871 | -1.006 | -.890 | -.939 | -.892 |
| 80 | -.627 | -.745 | -.661 | -.659 | -.659 |
| 81 | -.271 | -.383 | -.317 | -.295 | -.311 |
| 82 | .068 | -.022 | .030 | .064 | .009 |
| 83 | .322 | .255 | .297 | .318 | .259 |
| 84 | .584 | .537 | .556 | .584 | .544 |

Figure 18 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 150^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .790 | .795 | .797 | .790 | .779 |
| 2 | .636 | .635 | .664 | .642 | .649 |
| 3 | .607 | .288 | .274 | .273 | .276 |
| 4 | -.307 | -.252 | -.277 | -.270 | -.243 |
| 5 | -.871 | -.987 | -.998 | -.934 | -.939 |
| 6 | -1.480 | -1.398 | -1.488 | -1.363 | -1.382 |
| 7 | -1.778 | -1.770 | -1.874 | -1.708 | -1.727 |
| 8 | -1.803 | -1.792 | -1.897 | -1.701 | -1.739 |
| 9 | -1.829 | -1.814 | -1.921 | -1.694 | -1.751 |
| 10 | -1.673 | -1.570 | -1.802 | -1.343 | -1.790 |
| 11 | -1.084 | -1.154 | -1.429 | -1.075 | -1.315 |
| 12 | -.264 | -.236 | -.311 | -.527 | -.352 |
| 13 | -.755 | -.823 | -.968 | -.748 | -.896 |
| 14 | -1.155 | -1.018 | -1.168 | -1.374 | -1.251 |
| 15 | -1.524 | -1.613 | -1.488 | -1.617 | -1.461 |
| 16 | -1.531 | -1.443 | -1.483 | -1.673 | -1.485 |
| 17 | -1.664 | -1.713 | -1.568 | -1.793 | -1.554 |
| 18 | -1.605 | -1.537 | -1.491 | -1.769 | -1.470 |
| 19 | -1.810 | -1.827 | -1.707 | -1.954 | -1.697 |
| 20 | -1.500 | -1.462 | -1.452 | -1.626 | -1.412 |
| 21 | -1.188 | -1.223 | -1.159 | -1.267 | -1.140 |
| 22 | -.549 | -.554 | -.549 | -.584 | -.515 |
| 23 | .079 | .053 | .081 | .052 | .069 |
| 24 | .496 | .473 | .506 | .482 | .486 |
| 25 | .649 | .653 | .659 | .635 | .634 |
| 26 | .555 | .546 | .569 | .566 | .557 |
| 27 | .220 | .255 | .241 | .254 | .250 |
| 28 | -.293 | -.241 | -.281 | -.233 | -.234 |
| 29 | -.953 | -.931 | -.946 | -.840 | -.881 |
| 30 | -1.352 | -1.293 | -1.401 | -1.213 | -1.281 |
| 31 | -1.622 | -1.603 | -1.751 | -1.394 | -1.614 |
| 32 | -1.547 | -1.470 | -1.667 | -1.387 | -1.513 |
| 33 | -1.683 | -1.656 | -1.794 | -1.469 | -1.662 |
| 34 | -1.207 | -1.208 | -1.395 | -1.304 | -1.227 |
| 35 | -1.211 | -1.239 | -1.312 | -1.044 | -1.229 |
| 36 | -.925 | -.851 | -1.057 | -.822 | -1.009 |
| 37 | -.626 | -.620 | -.665 | -.584 | -.643 |
| 38 | -1.167 | -1.156 | -1.225 | -1.433 | -1.207 |
| 39 | -1.543 | -1.578 | -1.521 | -1.508 | -1.492 |
| 40 | -1.297 | -1.315 | -1.298 | -1.581 | -1.305 |
| 41 | -1.513 | -1.560 | -1.447 | -1.610 | -1.429 |
| 42 | -1.280 | -1.273 | -1.214 | -1.501 | -1.212 |

Figure 19 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 150^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.501 | -1.523 | -1.362 | -1.724 | -1.390 |
| 44 | -1.356 | -1.344 | -1.296 | -1.531 | -1.288 |
| 45 | -1.140 | -1.183 | -1.073 | -1.279 | -1.091 |
| 46 | -.663 | -.681 | -.632 | -.747 | -.649 |
| 47 | -.046 | -.089 | -.045 | -.134 | -.068 |
| 48 | .350 | .312 | .351 | .297 | .324 |
| 49 | .605 | .577 | .606 | .596 | .587 |
| 50 | .609 | .625 | .614 | .618 | .609 |
| 51 | .396 | .395 | .405 | .418 | .409 |
| 52 | .108 | .131 | .122 | .147 | .126 |
| 53 | -.252 | -.225 | -.245 | -.195 | -.229 |
| 54 | -.780 | -.737 | -.801 | -.686 | -.737 |
| 55 | -1.079 | -1.015 | -1.141 | -.967 | -1.042 |
| 56 | -1.518 | -1.472 | -1.609 | -1.337 | -1.482 |
| 57 | -1.560 | -1.483 | -1.722 | -1.333 | -1.549 |
| 58 | -1.606 | -1.588 | -1.839 | -1.344 | -1.640 |
| 59 | -1.492 | -1.423 | -1.677 | -1.276 | -1.501 |
| 60 | -1.554 | -1.519 | -1.740 | -1.326 | -1.551 |
| 61 | -1.530 | -1.440 | -1.682 | -1.322 | -1.502 |
| 62 | -1.684 | -1.650 | -1.850 | -1.504 | -1.684 |
| 63 | -1.374 | -1.320 | -1.610 | -1.340 | -1.377 |
| 64 | -1.101 | -1.156 | -1.271 | -1.135 | -1.127 |
| 65 | -.799 | -.807 | -.992 | -.830 | -.861 |
| 66 | -.743 | -.730 | -.913 | -.750 | -.784 |
| 67 | -.657 | -.615 | -.770 | -.708 | -.669 |
| 68 | -1.010 | -.961 | -.938 | -1.138 | -.797 |
| 69 | -.655 | -.604 | -.731 | -.699 | -.946 |
| 70 | -1.286 | -1.514 | -1.337 | -1.548 | -1.331 |
| 71 | -1.014 | -1.059 | -.998 | -1.204 | -1.101 |
| 72 | -1.250 | -1.269 | -1.090 | -1.445 | -1.157 |
| 73 | -1.120 | -1.122 | -1.018 | -1.310 | -1.055 |
| 74 | -1.150 | -1.150 | -.992 | -1.323 | -1.036 |
| 75 | -1.082 | -1.051 | -.947 | -1.268 | -.981 |
| 76 | -1.184 | -1.177 | -.993 | -1.404 | -1.056 |
| 77 | -1.217 | -1.178 | -1.057 | -1.413 | -1.100 |
| 78 | -1.215 | -1.244 | -1.092 | -1.382 | -1.138 |
| 79 | -.980 | -.972 | -.901 | -1.125 | -.926 |
| 80 | -.703 | -.746 | -.667 | -.807 | -.687 |
| 81 | -.334 | -.358 | -.334 | -.421 | -.343 |
| 82 | .018 | -.006 | .029 | -.043 | .003 |
| 83 | .290 | .267 | .297 | .244 | .264 |
| 84 | .562 | .550 | .556 | .542 | .544 |

Figure 19 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 150^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .828 | .854 | .815 | .808 | .797 |
| 2 | .689 | .661 | .720 | .753 | .733 |
| 3 | .272 | .190 | .313 | .406 | .366 |
| 4 | -.408 | -.534 | -.395 | -.268 | -.273 |
| 5 | -1.088 | -1.258 | -1.102 | -.943 | -.913 |
| 6 | -.971 | -1.129 | -1.058 | -.928 | -.910 |
| 7 | -1.211 | -1.476 | -1.339 | -1.282 | -1.174 |
| 8 | -1.091 | -1.274 | -1.174 | -1.108 | -1.039 |
| 9 | -1.102 | -1.323 | -1.283 | -1.365 | -1.071 |
| 10 | -.886 | -1.175 | -.993 | -.875 | -.891 |
| 11 | -.828 | -.962 | -.824 | -.844 | -.755 |
| 12 | -.708 | -.935 | -.725 | -.687 | -.709 |
| 13 | -1.083 | -1.353 | -1.065 | -1.122 | -1.004 |
| 14 | -1.844 | -1.916 | -1.856 | -1.916 | -1.818 |
| 15 | -1.975 | -1.990 | -2.010 | -2.155 | -1.990 |
| 16 | -2.125 | -1.969 | -2.093 | -2.127 | -2.133 |
| 17 | -2.175 | -1.943 | -2.121 | -2.228 | -2.154 |
| 18 | -2.111 | -1.786 | -2.007 | -2.042 | -2.060 |
| 19 | -2.385 | -2.074 | -2.307 | -2.387 | -2.379 |
| 20 | -2.034 | -1.745 | -2.003 | -2.078 | -2.075 |
| 21 | -1.672 | -1.440 | -1.659 | -1.792 | -1.763 |
| 22 | -.889 | -.710 | -.904 | -1.006 | -1.012 |
| 23 | -.094 | .030 | -.123 | -.244 | -.212 |
| 24 | .460 | .528 | .440 | .377 | .387 |
| 25 | .750 | .778 | .758 | .737 | .725 |
| 26 | .648 | .616 | .664 | .692 | .674 |
| 27 | .246 | .199 | .292 | .371 | .332 |
| 28 | -.340 | -.413 | -.302 | -.206 | -.235 |
| 29 | -1.163 | -1.297 | -1.146 | -1.008 | -.985 |
| 30 | -1.586 | -1.733 | -1.613 | -1.475 | -1.447 |
| 31 | -1.840 | -2.079 | -1.923 | -1.777 | -1.724 |
| 32 | -1.710 | -1.900 | -1.787 | -1.663 | -1.616 |
| 33 | -1.917 | -2.159 | -2.011 | -1.911 | -1.848 |
| 34 | -1.949 | -2.289 | -2.185 | -2.052 | -1.974 |
| 35 | -.923 | -1.033 | -1.016 | -1.081 | -.950 |
| 36 | -1.165 | -1.349 | -1.304 | -1.264 | -1.228 |
| 37 | -1.280 | -1.242 | -1.182 | -1.246 | -1.215 |
| 38 | -1.402 | -1.575 | -1.379 | -1.329 | -1.373 |
| 39 | -1.543 | -1.513 | -1.470 | -1.476 | -1.540 |
| 40 | -1.607 | -1.457 | -1.489 | -1.488 | -1.598 |
| 41 | -1.629 | -1.436 | -1.531 | -1.594 | -1.622 |
| 42 | -1.582 | -1.299 | -1.447 | -1.518 | -1.574 |

Figure 20 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 90^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.803 | -1.466 | -1.604 | -1.652 | -1.738 |
| 44 | -1.663 | -1.397 | -1.580 | -1.633 | -1.688 |
| 45 | -1.410 | -1.167 | -1.337 | -1.427 | -1.441 |
| 46 | -.820 | -.657 | -.822 | -.910 | -.921 |
| 47 | -.099 | -.011 | -.130 | -.204 | -.210 |
| 48 | .401 | .434 | .367 | .334 | .311 |
| 49 | .782 | .775 | .776 | .764 | .764 |
| 50 | .757 | .742 | .778 | .809 | .784 |
| 51 | .491 | .444 | .537 | .578 | .559 |
| 52 | .111 | .044 | .153 | .265 | .224 |
| 53 | -.303 | -.384 | -.266 | -.144 | -.173 |
| 54 | -.918 | -1.011 | -.838 | -.708 | -.760 |
| 55 | -1.296 | -1.382 | -1.232 | -1.104 | -1.148 |
| 56 | -1.780 | -1.920 | -1.741 | -1.641 | -1.617 |
| 57 | -1.864 | -2.005 | -1.875 | -1.775 | -1.767 |
| 58 | -1.806 | -2.043 | -1.861 | -1.765 | -1.708 |
| 59 | -1.783 | -1.942 | -1.821 | -1.704 | -1.681 |
| 60 | -1.787 | -2.061 | -1.892 | -1.815 | -1.726 |
| 61 | -1.666 | -1.878 | -1.768 | -1.661 | -1.588 |
| 62 | -1.895 | -1.978 | -1.860 | -1.782 | -1.738 |
| 63 | -1.668 | -1.834 | -1.718 | -1.655 | -1.887 |
| 64 | -2.015 | -2.316 | -2.254 | -2.313 | -2.235 |
| 65 | -1.083 | -1.214 | -1.473 | -1.665 | -1.427 |
| 66 | -1.305 | -1.324 | -1.240 | -1.221 | -1.237 |
| 67 | -1.241 | -1.191 | -1.224 | -1.222 | -1.242 |
| 68 | -1.331 | -1.272 | -1.191 | -1.178 | -1.265 |
| 69 | -1.207 | -1.125 | -1.122 | -1.078 | -1.180 |
| 70 | -1.200 | -1.053 | -1.071 | -1.096 | -1.138 |
| 71 | -1.053 | -.914 | -.977 | -.978 | -1.014 |
| 72 | -1.072 | -.905 | -.995 | -.990 | -1.071 |
| 73 | -1.003 | -.852 | -.952 | -.942 | -1.012 |
| 74 | -1.027 | -.877 | -.951 | -.988 | -1.014 |
| 75 | -.962 | -.807 | -.893 | -.915 | -.962 |
| 76 | -.992 | -.848 | -.921 | -.972 | -.977 |
| 77 | -1.021 | -.869 | -.929 | -.960 | -.995 |
| 78 | -1.107 | -.973 | -1.053 | -1.123 | -1.126 |
| 79 | -.933 | -.806 | -.909 | -.969 | -.976 |
| 80 | -.688 | -.589 | -.691 | -.789 | -.760 |
| 81 | -.316 | -.244 | -.342 | -.430 | -.407 |
| 82 | .099 | .176 | .072 | -.025 | .018 |
| 83 | .417 | .460 | .389 | .309 | .347 |
| 84 | .757 | .782 | .729 | .689 | .692 |

Figure 20 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 90^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .819 | .839 | .823 | .803 | .803 |
| 2 | .671 | .651 | .707 | .762 | .734 |
| 3 | .231 | .223 | .331 | .480 | .394 |
| 4 | -.418 | -.490 | -.312 | -.130 | -.212 |
| 5 | -1.067 | -1.202 | -.955 | -.739 | -.817 |
| 6 | -.905 | -1.049 | -.791 | -.695 | -.728 |
| 7 | -1.121 | -1.183 | -1.077 | -.995 | -.996 |
| 8 | -1.034 | -1.052 | -.911 | -.869 | -.933 |
| 9 | -1.732 | -1.984 | -1.693 | -.937 | -1.704 |
| 10 | -1.197 | -1.245 | -1.054 | -.831 | -1.116 |
| 11 | -1.280 | -1.412 | -1.216 | -.740 | -1.283 |
| 12 | -.985 | -1.063 | -.871 | -.700 | -.926 |
| 13 | -1.274 | -1.412 | -1.200 | -1.168 | -1.115 |
| 14 | -2.019 | -1.932 | -1.999 | -1.804 | -1.914 |
| 15 | -2.039 | -1.947 | -2.154 | -2.056 | -1.987 |
| 16 | -2.137 | -1.980 | -2.132 | -2.188 | -2.159 |
| 17 | -2.062 | -1.990 | -2.161 | -2.328 | -2.134 |
| 18 | -1.948 | -1.843 | -1.966 | -2.201 | -2.045 |
| 19 | -2.236 | -2.070 | -2.307 | -2.590 | -2.307 |
| 20 | -1.931 | -1.765 | -1.949 | -2.221 | -2.040 |
| 21 | -1.581 | -1.460 | -1.691 | -1.957 | -1.725 |
| 22 | -.851 | -.745 | -.908 | -1.125 | -1.000 |
| 23 | -.053 | .014 | -.147 | -.329 | -.213 |
| 24 | .473 | .510 | .415 | .314 | .383 |
| 25 | .752 | .757 | .745 | .717 | .725 |
| 26 | .628 | .607 | .648 | .694 | .670 |
| 27 | .236 | .184 | .308 | .392 | .340 |
| 28 | -.349 | -.434 | -.258 | -.157 | -.212 |
| 29 | -1.132 | -1.278 | -1.080 | -.947 | -.973 |
| 30 | -1.558 | -1.716 | -1.497 | -1.401 | -1.438 |
| 31 | -1.857 | -1.930 | -1.809 | -1.727 | -1.668 |
| 32 | -1.691 | -1.741 | -1.584 | -1.528 | -1.519 |
| 33 | -1.933 | -1.967 | -1.903 | -1.865 | -1.789 |
| 34 | -1.195 | -1.224 | -1.360 | -1.856 | -1.143 |
| 35 | -.981 | -.998 | -.951 | -1.014 | -.860 |
| 36 | -.828 | -.804 | -.845 | -1.264 | -.738 |
| 37 | -1.203 | -1.191 | -1.128 | -1.412 | -1.048 |
| 38 | -1.530 | -1.591 | -1.410 | -1.513 | -1.382 |
| 39 | -1.591 | -1.560 | -1.552 | -1.695 | -1.525 |
| 40 | -1.626 | -1.528 | -1.517 | -1.631 | -1.593 |
| 41 | -1.572 | -1.508 | -.533 | -1.706 | -1.649 |
| 42 | -1.497 | -1.398 | -1.484 | -1.575 | -1.582 |

Figure 21 - Pressure Coefficients

$$\alpha = 60^\circ, \phi_j = 90^\circ, C_\mu = .02$$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.673 | -1.588 | -1.718 | -1.770 | -1.758 |
| 44 | -1.593 | -1.491 | -1.612 | -1.712 | -1.720 |
| 45 | -1.350 | -1.225 | -1.419 | -1.513 | -1.461 |
| 46 | -.805 | -.716 | -.850 | -.967 | -.950 |
| 47 | -.101 | -.035 | -.161 | -.266 | -.217 |
| 48 | .385 | .413 | .343 | .283 | .308 |
| 49 | .774 | .763 | .752 | .756 | .769 |
| 50 | .739 | .733 | .779 | .821 | .774 |
| 51 | .476 | .458 | .511 | .592 | .553 |
| 52 | .084 | .037 | .178 | .280 | .202 |
| 53 | -.337 | -.387 | -.237 | -.125 | -.206 |
| 54 | -.930 | -.990 | -.843 | -.692 | -.755 |
| 55 | -1.317 | -1.369 | -1.211 | -1.082 | -1.158 |
| 56 | -1.807 | -1.904 | -1.738 | -1.497 | -1.651 |
| 57 | -1.905 | -2.010 | -1.822 | -1.593 | -1.819 |
| 58 | -1.854 | -1.982 | -1.900 | -1.606 | -1.821 |
| 59 | -1.805 | -1.902 | -1.766 | -1.557 | -1.742 |
| 60 | -1.819 | -2.020 | -1.946 | -1.775 | -1.766 |
| 61 | -1.712 | -1.846 | -1.852 | -1.644 | -1.730 |
| 62 | -1.872 | -2.048 | -1.942 | -1.772 | -1.861 |
| 63 | -1.751 | -1.887 | -1.796 | -1.690 | -2.074 |
| 64 | -1.659 | -1.674 | -1.807 | -1.719 | -1.574 |
| 65 | -.968 | -1.039 | -1.028 | -1.532 | -1.063 |
| 66 | -1.228 | -1.179 | -1.237 | -1.416 | -1.150 |
| 67 | -1.207 | -1.149 | -1.138 | -1.426 | -1.151 |
| 68 | -1.386 | -1.374 | -1.324 | -1.412 | -1.275 |
| 69 | -1.209 | -1.216 | -1.152 | -1.264 | -1.157 |
| 70 | -1.180 | -1.124 | -1.165 | -1.216 | -1.200 |
| 71 | -1.044 | -.976 | -1.006 | -1.068 | -1.068 |
| 72 | -1.034 | -.948 | -1.080 | -1.071 | -1.122 |
| 73 | -.990 | -.905 | -1.005 | -1.012 | -1.076 |
| 74 | -1.006 | -1.011 | -1.064 | -1.040 | -1.104 |
| 75 | -.951 | -.944 | -.968 | -.942 | -1.054 |
| 76 | -1.023 | -.982 | -1.039 | -.983 | -1.088 |
| 77 | -1.085 | -1.039 | -1.031 | -.937 | -1.123 |
| 78 | -1.126 | -1.037 | -1.167 | -1.135 | -1.218 |
| 79 | -.954 | -.861 | -.966 | -.975 | -1.059 |
| 80 | -.687 | -.620 | -.764 | -.822 | -.817 |
| 81 | -.317 | -.267 | -.375 | -.449 | -.449 |
| 82 | .102 | .156 | .028 | -.042 | -.015 |
| 83 | .421 | .441 | .351 | .305 | .334 |
| 84 | .737 | .760 | .725 | .687 | .689 |

Figure 21 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 90^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .786 | .801 | .830 | .817 | .749 |
| 2 | .459 | .673 | .423 | .439 | .747 |
| 3 | .301 | .615 | .342 | .331 | .477 |
| 4 | -.367 | .020 | -.347 | -.332 | -.074 |
| 5 | -.991 | -.718 | -1.030 | -.996 | -.710 |
| 6 | -1.134 | -1.002 | -1.186 | -1.142 | -.917 |
| 7 | -1.276 | -1.287 | -1.342 | -1.288 | -1.124 |
| 8 | -1.303 | -1.421 | -1.287 | -1.224 | -.859 |
| 9 | -1.258 | -1.368 | -1.355 | -1.278 | -.916 |
| 10 | -1.419 | -1.398 | -1.420 | -1.314 | -.878 |
| 11 | -1.502 | -1.502 | -1.611 | -1.508 | -1.275 |
| 12 | -1.367 | -1.470 | -1.334 | -1.292 | -1.232 |
| 13 | -1.924 | -1.214 | -2.089 | -2.025 | -1.207 |
| 14 | -2.888 | -2.961 | -2.668 | -2.752 | -2.491 |
| 15 | -2.925 | -2.593 | -2.976 | -3.011 | -2.306 |
| 16 | -3.214 | -3.247 | -2.993 | -3.061 | -2.716 |
| 17 | -3.194 | -3.050 | -3.277 | -3.282 | -2.691 |
| 18 | -2.763 | -3.127 | -2.634 | -2.621 | -2.747 |
| 19 | -3.360 | -3.420 | -3.443 | -3.470 | -2.970 |
| 20 | -3.257 | -3.200 | -3.072 | -3.082 | -2.551 |
| 21 | -2.089 | -2.506 | -2.039 | -2.146 | -2.098 |
| 22 | -2.640 | -2.244 | -2.462 | -2.533 | -1.276 |
| 23 | -.102 | -.613 | .354 | -.106 | -.392 |
| 24 | -.082 | -.149 | .652 | -.121 | .264 |
| 25 | .911 | .757 | 1.002 | .939 | .632 |
| 26 | .816 | .930 | .796 | .751 | .693 |
| 27 | .442 | .676 | .498 | .453 | .505 |
| 28 | -.095 | .218 | -.105 | -.115 | .063 |
| 29 | -.640 | -.373 | -.596 | -.621 | -.431 |
| 30 | -.684 | -.677 | -.633 | -.652 | -.604 |
| 31 | -.824 | -.746 | -.814 | -.823 | -.700 |
| 32 | -.787 | -.859 | -.747 | -.755 | -.717 |
| 33 | -.797 | -.819 | -.794 | -.805 | -.853 |
| 34 | -.799 | -.840 | -.779 | -.775 | -.812 |
| 35 | -.912 | -.883 | -.921 | -.906 | -.815 |
| 36 | -1.031 | -.944 | -.894 | -1.007 | -.840 |
| 37 | -2.395 | -1.420 | -2.426 | -2.375 | -1.440 |
| 38 | -2.132 | -2.555 | -2.146 | -2.011 | -2.314 |
| 39 | -2.921 | -2.280 | -2.913 | -2.712 | -2.173 |
| 40 | -2.818 | -2.938 | -2.629 | -2.503 | -2.559 |
| 41 | -2.998 | -2.843 | -3.081 | -2.964 | -2.553 |
| 42 | -3.132 | -2.892 | -2.940 | -2.831 | -2.562 |

Figure 22 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -3.168 | -3.133 | -3.370 | -3.345 | -2.832 |
| 44 | -2.718 | -2.950 | -2.688 | -2.657 | -2.515 |
| 45 | -2.155 | -2.571 | -2.257 | -2.160 | -2.140 |
| 46 | -1.196 | -1.767 | -1.194 | -1.119 | -1.375 |
| 47 | -.234 | -.691 | -.198 | -.217 | -.563 |
| 48 | .470 | .150 | .435 | .439 | .099 |
| 49 | .862 | .737 | .918 | .877 | .735 |
| 50 | .872 | .935 | .794 | .794 | .739 |
| 51 | .642 | .836 | .691 | .670 | .659 |
| 52 | .295 | .602 | .258 | .274 | .404 |
| 53 | -.162 | .227 | -.127 | -.124 | .125 |
| 54 | -.505 | -.121 | -.475 | -.450 | -.178 |
| 55 | -.765 | -.573 | -.758 | -.764 | -.568 |
| 56 | -.713 | -.713 | -.691 | -.671 | -.619 |
| 57 | -.647 | -.623 | -.619 | -.691 | -.601 |
| 58 | -.729 | -.703 | -.628 | -.717 | -.588 |
| 59 | -.579 | -.563 | -.592 | -.645 | -.574 |
| 60 | -.754 | -.731 | -.694 | -.789 | -.577 |
| 61 | -.716 | -.744 | -.693 | -.819 | -.614 |
| 62 | -.610 | -.636 | -.573 | -.644 | -.624 |
| 63 | -.966 | -.840 | -.914 | -.947 | -.763 |
| 64 | -1.010 | -.828 | -.883 | -.915 | -.859 |
| 65 | -1.577 | -1.229 | -1.450 | -1.605 | -.863 |
| 66 | -1.662 | -1.495 | -1.747 | -1.577 | -1.158 |
| 67 | -2.043 | -1.972 | -2.158 | -1.952 | -1.606 |
| 68 | -2.066 | -2.320 | -1.985 | -1.796 | -1.573 |
| 69 | -2.085 | -2.028 | -2.399 | -2.198 | -2.042 |
| 70 | -2.107 | -2.147 | -2.236 | -2.021 | -2.007 |
| 71 | -2.433 | -2.199 | -2.318 | -2.078 | -2.026 |
| 72 | -2.477 | -2.291 | -2.176 | -1.927 | -2.071 |
| 73 | -2.131 | -2.235 | -2.630 | -2.232 | -2.163 |
| 74 | -2.009 | -2.210 | -2.347 | -2.027 | -2.023 |
| 75 | -2.655 | -2.220 | -2.551 | -2.016 | -1.969 |
| 76 | -2.704 | -2.300 | -2.436 | -1.946 | -1.982 |
| 77 | -2.504 | -2.659 | -2.698 | -2.448 | -2.431 |
| 78 | -2.180 | -2.495 | -2.155 | -2.006 | -2.157 |
| 79 | -1.714 | -2.128 | -1.798 | -1.670 | -1.902 |
| 80 | -1.174 | -1.693 | -1.117 | -1.035 | -1.411 |
| 81 | -.325 | -.711 | -.363 | -.331 | -.873 |
| 82 | .093 | -.431 | .029 | .061 | -.347 |
| 83 | .484 | .174 | .501 | .511 | .152 |
| 84 | .833 | .603 | .775 | .773 | .463 |

Figure 22 (continued) – Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .387 | .858 | .808 | .823 | .830 |
| 2 | .677 | .691 | .739 | .706 | .708 |
| 3 | .072 | .285 | .410 | .273 | .307 |
| 4 | -.451 | -.428 | -.266 | -.400 | -.348 |
| 5 | -1.305 | -1.305 | -1.081 | -1.206 | -1.063 |
| 6 | -1.665 | -1.687 | -1.485 | -1.474 | -1.345 |
| 7 | -2.026 | -2.068 | -1.889 | -1.742 | -1.627 |
| 8 | -1.525 | -1.514 | -1.346 | -1.371 | -1.248 |
| 9 | -1.405 | -1.436 | -1.415 | -1.257 | -1.114 |
| 10 | -.958 | -1.011 | -.694 | -.723 | -.636 |
| 11 | -1.274 | -1.325 | -1.214 | -1.021 | -.918 |
| 12 | -.936 | -.944 | -.715 | -.695 | -.675 |
| 13 | -1.116 | -1.164 | -.909 | -1.067 | -.997 |
| 14 | -1.844 | -1.857 | -1.860 | -1.941 | -1.868 |
| 15 | -1.850 | -1.909 | -2.089 | -2.040 | -1.973 |
| 16 | -1.939 | -1.946 | -2.077 | -2.185 | -2.114 |
| 17 | -1.882 | -1.939 | -2.123 | -2.159 | -2.074 |
| 18 | -1.754 | -1.763 | -1.912 | -2.094 | -1.988 |
| 19 | -2.013 | -2.074 | -2.223 | -2.365 | -2.260 |
| 20 | -1.769 | -1.756 | -1.954 | -2.025 | -1.939 |
| 21 | -1.424 | -1.474 | -1.672 | -1.603 | -1.556 |
| 22 | -.788 | -.783 | -.982 | -.885 | -.852 |
| 23 | -.049 | -.038 | -.231 | -.065 | .220 |
| 24 | .461 | .469 | .345 | .481 | .731 |
| 25 | .747 | .783 | .741 | .757 | .729 |
| 26 | .652 | .664 | .692 | .663 | .659 |
| 27 | .266 | .305 | .417 | .277 | .294 |
| 28 | -.373 | -.337 | -.194 | -.357 | -.301 |
| 29 | -1.209 | -1.186 | -.988 | -1.118 | -1.026 |
| 30 | -1.658 | -1.606 | -1.453 | -1.520 | -1.433 |
| 31 | -1.966 | -2.024 | -1.863 | -1.679 | -1.623 |
| 32 | -1.919 | -1.902 | -1.768 | -1.714 | -1.631 |
| 33 | -2.108 | -2.126 | -1.973 | -1.813 | -1.751 |
| 34 | -2.212 | -2.174 | -2.027 | -1.897 | -1.864 |
| 35 | -1.321 | -1.351 | -1.444 | -1.157 | -1.044 |
| 36 | -1.264 | -1.233 | -1.181 | -1.148 | -1.111 |
| 37 | -.978 | -.959 | -1.074 | -1.213 | -1.136 |
| 38 | -1.366 | -1.419 | -1.332 | -1.372 | -1.306 |
| 39 | -1.303 | -1.415 | -1.412 | -1.530 | -1.445 |
| 40 | -1.340 | -1.375 | -1.449 | -1.596 | -1.511 |
| 41 | -1.376 | -1.460 | -1.548 | -1.647 | -1.592 |
| 42 | -1.273 | -1.280 | -1.433 | -1.601 | -1.540 |

Figure 23 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.397 | -1.460 | -1.570 | -1.797 | -1.720 |
| 44 | -1.418 | -1.446 | -1.591 | -1.717 | -1.666 |
| 45 | -1.184 | -1.237 | -1.416 | -1.409 | -1.356 |
| 46 | -.698 | -.720 | -.915 | -.838 | -.825 |
| 47 | -.098 | -.074 | -.270 | -.144 | -.130 |
| 48 | .383 | .401 | .265 | .376 | .366 |
| 49 | .822 | .851 | .815 | .834 | .803 |
| 50 | .725 | .729 | .755 | .739 | .726 |
| 51 | .526 | .550 | .640 | .520 | .521 |
| 52 | .163 | .178 | .300 | .169 | .181 |
| 53 | -.336 | -.318 | -.179 | -.356 | -.309 |
| 54 | -.773 | -.748 | -.612 | -.785 | -.735 |
| 55 | -1.373 | -1.409 | -1.216 | -1.340 | -1.247 |
| 56 | -1.616 | -1.627 | -1.476 | -1.587 | -1.500 |
| 57 | -1.970 | -1.953 | -1.885 | -1.812 | -1.780 |
| 58 | -1.815 | -1.760 | -1.742 | -1.680 | -1.646 |
| 59 | -1.878 | -1.930 | -1.834 | -1.736 | -1.697 |
| 60 | -1.752 | -1.767 | -1.687 | -1.661 | -1.652 |
| 61 | -1.909 | -1.949 | -1.802 | -1.861 | -1.830 |
| 62 | -1.783 | -1.751 | -1.664 | -1.926 | -1.871 |
| 63 | -2.000 | -2.350 | -2.103 | -2.465 | -2.351 |
| 64 | -2.068 | -2.322 | -2.310 | -2.370 | -2.249 |
| 65 | -1.660 | -1.813 | -2.161 | -1.512 | -2.210 |
| 66 | -2.106 | -2.421 | -2.384 | -2.434 | -2.427 |
| 67 | -1.157 | -1.161 | -1.240 | -1.180 | -1.387 |
| 68 | -1.370 | -1.431 | -1.444 | -1.564 | -1.554 |
| 69 | -1.117 | -1.137 | -1.168 | -1.211 | -1.159 |
| 70 | -1.013 | -1.007 | -1.051 | -1.193 | -1.118 |
| 71 | -.996 | -1.002 | -1.077 | -1.123 | -1.058 |
| 72 | -.881 | -.891 | -.957 | -1.095 | -1.046 |
| 73 | -.921 | -.974 | -1.088 | -1.115 | -1.074 |
| 74 | -.839 | -.870 | -.997 | -1.062 | -1.045 |
| 75 | -.865 | -.938 | -1.031 | -1.097 | -1.034 |
| 76 | -.818 | -.856 | -.948 | -1.051 | -.994 |
| 77 | -.910 | -.950 | -1.036 | -1.204 | -1.139 |
| 78 | -.955 | -.974 | -1.070 | -1.177 | -1.124 |
| 79 | -.881 | -.934 | -1.057 | -1.052 | -1.013 |
| 80 | -.643 | -.654 | -.789 | -.759 | -.729 |
| 81 | -.289 | -.326 | -.471 | -.333 | -.345 |
| 82 | .076 | .050 | -.075 | .057 | .046 |
| 83 | .460 | .456 | .339 | .464 | .438 |
| 84 | .657 | .641 | .576 | .670 | .649 |

Figure 23 (continued) – Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .368 | .834 | .795 | .819 | .826 |
| 2 | .660 | .688 | .748 | .720 | .675 |
| 3 | .067 | .266 | .395 | .320 | .280 |
| 4 | -.463 | -.449 | -.266 | -.324 | -.391 |
| 5 | -1.294 | -1.315 | -1.050 | -1.030 | -1.167 |
| 6 | -1.609 | -1.681 | -1.413 | -1.267 | -1.439 |
| 7 | -1.925 | -2.047 | -1.776 | -1.504 | -1.710 |
| 8 | -1.504 | -1.580 | -1.345 | -1.159 | -1.334 |
| 9 | -1.276 | -1.376 | -1.206 | -.949 | -1.189 |
| 10 | -1.359 | -1.650 | -1.451 | -.799 | -1.187 |
| 11 | -1.369 | -1.531 | -1.361 | -.911 | -1.276 |
| 12 | -1.130 | -1.249 | -1.121 | -.723 | -.984 |
| 13 | -1.181 | -1.347 | -1.001 | -1.210 | -1.117 |
| 14 | -1.923 | -1.872 | -1.900 | -1.804 | -1.916 |
| 15 | -1.871 | -1.908 | -2.011 | -1.972 | -2.111 |
| 16 | -1.960 | -1.928 | -2.075 | -2.227 | -2.049 |
| 17 | -1.886 | -1.876 | -2.053 | -2.280 | -2.101 |
| 18 | -1.749 | -1.716 | -1.915 | -2.285 | -1.889 |
| 19 | -2.020 | -2.068 | -2.171 | -2.587 | -2.219 |
| 20 | -1.766 | -1.784 | -1.970 | -2.192 | -1.849 |
| 21 | -1.422 | -1.438 | -1.641 | -1.749 | -1.530 |
| 22 | -.764 | -.768 | -1.004 | -.976 | -.808 |
| 23 | -.051 | -.033 | -.214 | -.142 | -.074 |
| 24 | .449 | .480 | .371 | .443 | .444 |
| 25 | .741 | .771 | .719 | .774 | .763 |
| 26 | .642 | .653 | .704 | .668 | .639 |
| 27 | .265 | .258 | .394 | .312 | .294 |
| 28 | -.360 | -.403 | -.203 | -.288 | -.309 |
| 29 | -1.154 | -1.236 | -.999 | -1.054 | -1.103 |
| 30 | -1.600 | -1.705 | -1.495 | -1.441 | -1.459 |
| 31 | -1.961 | -2.084 | -1.857 | -1.651 | -1.752 |
| 32 | -1.879 | -1.975 | -1.792 | -1.661 | -1.681 |
| 33 | -2.030 | -2.146 | -1.882 | -1.721 | -1.859 |
| 34 | -1.919 | -1.907 | -1.914 | -1.710 | -1.417 |
| 35 | -1.123 | -1.185 | -1.023 | -.993 | -1.067 |
| 36 | -1.094 | -1.098 | -1.087 | -1.188 | -.864 |
| 37 | -.917 | -.898 | -.862 | -1.393 | -.991 |
| 38 | -1.544 | -1.552 | -1.365 | -1.555 | -1.327 |
| 39 | -1.457 | -1.478 | -1.369 | -1.637 | -1.449 |
| 40 | -1.472 | -1.446 | -1.493 | -1.730 | -1.488 |
| 41 | -1.467 | -1.499 | -1.546 | -1.717 | -1.586 |
| 42 | -1.312 | -1.336 | -1.475 | -1.712 | -1.467 |

Figure 24 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.452 | -1.388 | -1.566 | -1.889 | -1.699 |
| 44 | -1.459 | -1.419 | -1.638 | -1.778 | -1.605 |
| 45 | -1.224 | -1.183 | -1.416 | -1.472 | -1.363 |
| 46 | -.752 | -.711 | -.949 | -.899 | -.793 |
| 47 | -.101 | -.070 | -.284 | -.148 | -.120 |
| 48 | .379 | .399 | .270 | .371 | .374 |
| 49 | .813 | .827 | .795 | .833 | .819 |
| 50 | .715 | .730 | .768 | .748 | .699 |
| 51 | .514 | .518 | .616 | .533 | .526 |
| 52 | .151 | .149 | .290 | .182 | .160 |
| 53 | -.361 | -.340 | -.154 | -.363 | -.331 |
| 54 | -.790 | -.791 | -.593 | -.797 | -.753 |
| 55 | -1.370 | -1.464 | -1.216 | -1.338 | -1.356 |
| 56 | -1.622 | -1.731 | -1.518 | -1.582 | -1.568 |
| 57 | -1.976 | -2.088 | -1.935 | -1.906 | -1.929 |
| 58 | -1.801 | -1.915 | -1.863 | -1.738 | -1.730 |
| 59 | -1.929 | -1.975 | -1.820 | -1.800 | -1.782 |
| 60 | -1.786 | -1.838 | -1.743 | -1.738 | -1.640 |
| 61 | -1.889 | -2.033 | -1.838 | -1.832 | -1.913 |
| 62 | -1.746 | -1.960 | -1.864 | -1.913 | -1.947 |
| 63 | -2.197 | -2.466 | -2.304 | -2.433 | -2.414 |
| 64 | -2.068 | -2.230 | -2.131 | -2.151 | -1.557 |
| 65 | -1.289 | -1.454 | -2.180 | -2.197 | -2.033 |
| 66 | -2.064 | -2.269 | -2.295 | -2.223 | -1.552 |
| 67 | -1.091 | -1.125 | -1.348 | -1.383 | -1.172 |
| 68 | -1.368 | -1.431 | -1.528 | -1.617 | -1.334 |
| 69 | -1.156 | -1.208 | -1.228 | -1.295 | -1.241 |
| 70 | -1.018 | -1.060 | -1.141 | -1.220 | -1.132 |
| 71 | -.991 | -1.031 | -1.105 | -1.138 | -1.133 |
| 72 | -.899 | -.882 | -1.027 | -1.075 | -1.058 |
| 73 | -.981 | -.949 | -1.087 | -1.082 | -1.138 |
| 74 | -.914 | -.873 | -1.054 | -1.035 | -1.051 |
| 75 | -.957 | -.985 | -1.090 | -1.058 | -1.121 |
| 76 | -.888 | -.911 | -1.037 | -1.002 | -1.044 |
| 77 | -1.029 | -1.040 | -1.127 | -1.153 | -1.231 |
| 78 | -1.027 | -1.052 | -1.185 | -1.126 | -1.152 |
| 79 | -.941 | -.930 | -1.102 | -1.031 | -1.084 |
| 80 | -.668 | -.659 | -.849 | -.734 | -.751 |
| 81 | -.332 | -.306 | -.494 | -.326 | -.372 |
| 82 | .039 | .073 | -.092 | .078 | .025 |
| 83 | .433 | .461 | .335 | .473 | .448 |
| 84 | .631 | .658 | .591 | .682 | .632 |

Figure 24 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .852 | .859 | .851 | .857 | .841 |
| 2 | .612 | .656 | .655 | .638 | .650 |
| 3 | .610 | .123 | .143 | .078 | .142 |
| 4 | -.648 | -.554 | -.545 | -.598 | -.517 |
| 5 | -1.343 | -1.485 | -1.488 | -1.515 | -1.434 |
| 6 | -2.076 | -1.981 | -2.033 | -1.994 | -1.934 |
| 7 | -2.465 | -2.344 | -2.446 | -2.279 | -2.267 |
| 8 | -2.506 | -2.381 | -2.473 | -2.319 | -2.313 |
| 9 | -2.548 | -2.418 | -2.500 | -2.360 | -2.359 |
| 10 | -1.867 | -1.884 | -1.939 | -1.941 | -1.784 |
| 11 | -1.240 | -1.109 | -1.188 | -.946 | -1.162 |
| 12 | -.542 | -.443 | -.599 | -.496 | -.545 |
| 13 | -1.196 | -.962 | -.990 | -.866 | -.948 |
| 14 | -1.807 | -1.826 | -1.746 | -2.009 | -1.888 |
| 15 | -1.752 | -1.930 | -1.821 | -1.845 | -1.751 |
| 16 | -1.840 | -1.933 | -1.852 | -2.061 | -1.887 |
| 17 | -1.784 | -1.951 | -1.860 | -1.985 | -1.851 |
| 18 | -1.635 | -1.756 | -1.678 | -1.862 | -1.708 |
| 19 | -1.926 | -2.061 | -1.972 | -2.140 | -1.993 |
| 20 | -1.601 | -1.724 | -1.678 | -1.778 | -1.693 |
| 21 | -1.319 | -1.430 | -1.395 | -1.433 | -1.407 |
| 22 | -.605 | -.673 | -.669 | -.661 | -.686 |
| 23 | .102 | .044 | .037 | .076 | .026 |
| 24 | .549 | .535 | .526 | .559 | .513 |
| 25 | .795 | .790 | .781 | .793 | .769 |
| 26 | .592 | .622 | .629 | .618 | .620 |
| 27 | .101 | .175 | .184 | .125 | .181 |
| 28 | -.575 | -.480 | -.483 | -.541 | -.461 |
| 29 | -1.518 | -1.361 | -1.355 | -1.402 | -1.294 |
| 30 | -2.008 | -1.838 | -1.883 | -1.855 | -1.777 |
| 31 | -2.486 | -2.214 | -2.373 | -2.098 | -2.177 |
| 32 | -2.305 | -2.105 | -2.221 | -2.073 | -2.068 |
| 33 | -2.538 | -2.304 | -2.451 | -2.190 | -2.271 |
| 34 | -2.214 | -2.011 | -2.162 | -1.976 | -1.968 |
| 35 | -2.007 | -1.801 | -1.968 | -1.821 | -1.750 |
| 36 | -1.451 | -1.293 | -1.430 | -1.253 | -1.267 |
| 37 | -1.425 | -1.081 | -1.271 | -1.137 | -1.052 |
| 38 | -1.488 | -1.573 | -1.492 | -1.685 | -1.492 |
| 39 | -1.372 | -1.530 | -1.437 | -1.494 | -1.385 |
| 40 | -1.219 | -1.383 | -1.299 | -1.539 | -1.375 |
| 41 | -1.236 | -1.462 | -1.326 | -1.506 | -1.375 |
| 42 | -1.062 | -1.236 | -1.162 | -1.360 | -1.227 |

Figure 25 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 150^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.185 | -1.386 | -1.262 | -1.533 | -1.351 |
| 44 | -1.183 | -1.364 | -1.288 | -1.458 | -1.353 |
| 45 | -.974 | -1.132 | -1.075 | -1.190 | -1.118 |
| 46 | -.526 | -.656 | -.638 | -.660 | -.660 |
| 47 | .109 | -.005 | .031 | .028 | .001 |
| 48 | .497 | .423 | .452 | .467 | .424 |
| 49 | .762 | .765 | .764 | .781 | .751 |
| 50 | .726 | .749 | .757 | .740 | .740 |
| 51 | .412 | .439 | .459 | .433 | .457 |
| 52 | .017 | .040 | .088 | .025 | .081 |
| 53 | -.455 | -.439 | -.384 | -.449 | -.376 |
| 54 | -1.205 | -1.146 | -1.119 | -1.150 | -1.100 |
| 55 | -1.591 | -1.543 | -1.535 | -1.538 | -1.491 |
| 56 | -2.296 | -2.162 | -2.207 | -2.074 | -2.137 |
| 57 | -2.374 | -2.246 | -2.330 | -2.108 | -2.220 |
| 58 | -2.653 | -2.431 | -2.604 | -2.161 | -2.321 |
| 59 | -2.341 | -2.204 | -2.335 | -2.032 | -2.110 |
| 60 | -2.519 | -2.298 | -2.475 | -2.142 | -2.235 |
| 61 | -2.311 | -2.189 | -2.326 | -2.105 | -2.128 |
| 62 | -2.610 | -2.398 | -2.623 | -2.273 | -2.377 |
| 63 | -2.601 | -2.291 | -2.440 | -2.212 | -2.254 |
| 64 | -2.770 | -2.519 | -2.665 | -2.537 | -2.394 |
| 65 | -2.300 | -2.037 | -2.267 | -1.915 | -1.853 |
| 66 | -2.051 | -1.754 | -2.010 | -1.493 | -1.665 |
| 67 | -1.532 | -1.388 | -1.540 | -1.272 | -1.345 |
| 68 | -1.336 | -1.283 | -1.376 | -1.325 | -1.407 |
| 69 | -1.277 | -1.260 | -1.344 | -1.097 | -1.395 |
| 70 | -.935 | -1.000 | -1.011 | -1.232 | -1.144 |
| 71 | -.829 | -.926 | -.919 | -1.037 | -1.015 |
| 72 | -.695 | -.844 | -.794 | -1.024 | -.913 |
| 73 | -.656 | -.817 | -.751 | -.947 | -.857 |
| 74 | -.648 | -.823 | -.743 | -.981 | -.855 |
| 75 | -.602 | -.782 | -.695 | -.908 | -.799 |
| 76 | -.628 | -.808 | -.711 | -.984 | -.823 |
| 77 | -.636 | -.845 | -.718 | -1.035 | -.855 |
| 78 | -.747 | -.949 | -.841 | -1.078 | -.969 |
| 79 | -.591 | -.759 | -.688 | -.848 | -.785 |
| 80 | -.412 | -.544 | -.499 | -.564 | -.558 |
| 81 | -.096 | -.191 | -.169 | -.174 | -.202 |
| 82 | .259 | .203 | .195 | .199 | .164 |
| 83 | .497 | .467 | .456 | .475 | .434 |
| 84 | .779 | .769 | .761 | .779 | .736 |

Figure 25 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 150^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .841 | .846 | .860 | .829 | .836 |
| 2 | .643 | .662 | .645 | .679 | .658 |
| 3 | .598 | .105 | .124 | .157 | .128 |
| 4 | -.618 | -.592 | -.560 | -.505 | -.544 |
| 5 | -1.294 | -1.511 | -1.554 | -1.311 | -1.438 |
| 6 | -2.058 | -2.103 | -2.066 | -1.833 | -1.974 |
| 7 | -2.331 | -2.357 | -2.535 | -2.002 | -2.308 |
| 8 | -2.378 | -2.394 | -2.573 | -2.076 | -2.331 |
| 9 | -2.424 | -2.432 | -2.611 | -2.150 | -2.353 |
| 10 | -1.873 | -2.046 | -2.032 | -1.670 | -2.102 |
| 11 | -1.296 | -1.401 | -1.655 | -1.099 | -1.483 |
| 12 | -.338 | -.396 | -.453 | -.577 | -.442 |
| 13 | -1.006 | -1.069 | -1.220 | -.880 | -1.084 |
| 14 | -1.880 | -1.884 | -1.827 | -2.143 | -1.903 |
| 15 | -2.011 | -1.941 | -1.904 | -1.912 | -1.790 |
| 16 | -2.023 | -2.043 | -1.891 | -2.316 | -1.947 |
| 17 | -2.018 | -1.953 | -1.898 | -2.156 | -1.878 |
| 18 | -1.831 | -1.829 | -1.674 | -2.170 | -1.753 |
| 19 | -2.100 | -2.059 | -2.009 | -2.351 | -2.034 |
| 20 | -1.769 | -1.784 | -1.673 | -2.055 | -1.742 |
| 21 | -1.428 | -1.424 | -1.418 | -1.599 | -1.428 |
| 22 | -.692 | -.704 | -.666 | -.824 | -.702 |
| 23 | .060 | .048 | .049 | -.031 | .026 |
| 24 | .551 | .552 | .532 | .518 | .527 |
| 25 | .779 | .775 | .786 | .759 | .770 |
| 26 | .620 | .635 | .615 | .668 | .619 |
| 27 | .142 | .142 | .160 | .227 | .169 |
| 28 | -.512 | -.532 | -.513 | -.383 | -.487 |
| 29 | -1.437 | -1.411 | -1.448 | -1.175 | -1.316 |
| 30 | -1.945 | -1.972 | -1.958 | -1.612 | -1.832 |
| 31 | -2.240 | -2.258 | -2.480 | -1.700 | -2.182 |
| 32 | -2.195 | -2.227 | -2.272 | -1.789 | -2.116 |
| 33 | -2.338 | -2.346 | -2.548 | -1.818 | -2.295 |
| 34 | -1.894 | -1.991 | -2.105 | -1.800 | -1.877 |
| 35 | -1.681 | -1.742 | -1.859 | -1.451 | -1.669 |
| 36 | -1.325 | -1.422 | -1.521 | -1.053 | -1.415 |
| 37 | -.905 | -1.051 | -1.195 | -.947 | -.953 |
| 38 | -1.849 | -1.756 | -1.622 | -1.868 | -1.694 |
| 39 | -1.428 | -1.598 | -1.565 | -1.629 | -1.579 |
| 40 | -1.504 | -1.550 | -1.399 | -1.906 | -1.540 |
| 41 | -1.425 | -1.480 | -1.411 | -1.768 | -1.486 |
| 42 | -1.296 | -1.296 | -1.186 | -1.705 | -1.299 |

Figure 26 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 150^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.412 | -1.384 | -1.288 | -1.881 | -1.389 |
| 44 | -1.392 | -1.401 | -1.283 | -1.788 | -1.400 |
| 45 | -1.135 | -1.110 | -1.085 | -1.448 | -1.146 |
| 46 | -.654 | -.641 | -.627 | -.900 | -.681 |
| 47 | .028 | .024 | .035 | -.116 | -.022 |
| 48 | .467 | .466 | .446 | .398 | .417 |
| 49 | .779 | .784 | .752 | .795 | .758 |
| 50 | .718 | .722 | .750 | .753 | .732 |
| 51 | .420 | .430 | .428 | .508 | .450 |
| 52 | .001 | .007 | .028 | .125 | .048 |
| 53 | -.484 | -.489 | -.464 | -.312 | -.426 |
| 54 | -1.198 | -1.188 | -1.215 | -.933 | -1.121 |
| 55 | -1.628 | -1.646 | -1.615 | -1.319 | -1.545 |
| 56 | -2.224 | -2.235 | -2.351 | -1.723 | -2.130 |
| 57 | -2.346 | -2.409 | -2.449 | -1.737 | -2.241 |
| 58 | -2.476 | -2.491 | -2.787 | -1.692 | -2.355 |
| 59 | -2.292 | -2.353 | -2.439 | -1.648 | -2.182 |
| 60 | -2.374 | -2.406 | -2.649 | -1.681 | -2.265 |
| 61 | -2.322 | -2.382 | -2.441 | -1.726 | -2.198 |
| 62 | -2.487 | -2.503 | -2.753 | -1.889 | -2.395 |
| 63 | -2.317 | -2.353 | -2.486 | -1.935 | -2.243 |
| 64 | -2.225 | -2.235 | -2.498 | -1.799 | -2.049 |
| 65 | -1.655 | -1.749 | -1.831 | -1.290 | -1.538 |
| 66 | -1.633 | -1.739 | -1.747 | -1.162 | -1.509 |
| 67 | -1.383 | -1.505 | -1.542 | -1.261 | -1.286 |
| 68 | -1.335 | -1.390 | -1.542 | -1.609 | -1.377 |
| 69 | -1.436 | -1.388 | -1.477 | -1.250 | -1.471 |
| 70 | -1.046 | -1.120 | -1.140 | -1.596 | -1.196 |
| 71 | -1.018 | -1.013 | -.960 | -1.346 | -1.073 |
| 72 | -.904 | -.893 | -.832 | -1.400 | -.944 |
| 73 | -.881 | -.879 | -.763 | -1.346 | -.919 |
| 74 | -.883 | -.853 | -.774 | -1.324 | -.906 |
| 75 | -.843 | -.826 | -.712 | -1.299 | -.860 |
| 76 | -.864 | -.815 | -.740 | -1.340 | -.873 |
| 77 | -.929 | -.890 | -.740 | -1.467 | -.934 |
| 78 | -.998 | -.960 | -.883 | -1.412 | -1.014 |
| 79 | -.800 | -.790 | -.703 | -1.187 | -.830 |
| 80 | -.540 | -.537 | -.508 | -.804 | -.587 |
| 81 | -.176 | -.188 | -.172 | -.380 | -.226 |
| 82 | .197 | .199 | .201 | .064 | .156 |
| 83 | .479 | .483 | .452 | .409 | .436 |
| 84 | .757 | .752 | .765 | .722 | .737 |

Figure 26 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 150^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .690 | .682 | .673 | .671 | .688 |
| 2 | .550 | .532 | .580 | .556 | .544 |
| 3 | .293 | .257 | .353 | .277 | .249 |
| 4 | -.168 | -.207 | -.094 | -.174 | -.214 |
| 5 | -.630 | -.670 | -.540 | -.624 | -.676 |
| 6 | -.789 | -.805 | -.758 | -.653 | -.648 |
| 7 | -.928 | -.926 | -.930 | -.897 | -.902 |
| 8 | -.754 | -.776 | -.781 | -.776 | -.773 |
| 9 | -1.135 | -1.141 | -1.101 | -1.100 | -1.126 |
| 10 | -1.071 | -1.104 | -1.048 | -1.108 | -1.099 |
| 11 | -1.376 | -1.291 | -1.322 | -1.272 | -1.186 |
| 12 | -.666 | -.670 | -.724 | -.722 | -.711 |
| 13 | -.755 | -.760 | -.665 | -.663 | -.678 |
| 14 | -.971 | -.985 | -.956 | -1.059 | -1.093 |
| 15 | -1.053 | -1.047 | -1.066 | -1.077 | -1.045 |
| 16 | -1.098 | -1.101 | -1.094 | -1.154 | -1.163 |
| 17 | -1.165 | -1.149 | -1.162 | -1.167 | -1.172 |
| 18 | -1.158 | -1.170 | -1.151 | -1.187 | -1.215 |
| 19 | -1.267 | -1.242 | -1.260 | -1.241 | -1.265 |
| 20 | -1.027 | -1.012 | -1.072 | -1.050 | -1.040 |
| 21 | -.780 | -.730 | -.823 | -.756 | -.736 |
| 22 | -.315 | -.275 | -.389 | -.316 | -.274 |
| 23 | .170 | .215 | .108 | .170 | .213 |
| 24 | .459 | .484 | .422 | .467 | .499 |
| 25 | .502 | .510 | .478 | .479 | .509 |
| 26 | .415 | .400 | .444 | .432 | .419 |
| 27 | .211 | .173 | .262 | .195 | .175 |
| 28 | -.143 | -.195 | -.072 | -.155 | -.190 |
| 29 | -.609 | -.656 | -.531 | -.606 | -.646 |
| 30 | -.844 | -.889 | -.816 | -.868 | -.885 |
| 31 | -1.041 | -1.034 | -1.012 | -1.006 | -1.004 |
| 32 | -.994 | -1.013 | -.990 | -1.010 | -.987 |
| 33 | -1.069 | -1.057 | -1.028 | -1.036 | -1.019 |
| 34 | -.962 | -1.152 | -.943 | -1.062 | -1.144 |
| 35 | -1.085 | -1.282 | -1.063 | -1.198 | -1.341 |
| 36 | -.710 | -.734 | -.778 | -.733 | -.759 |
| 37 | -.616 | -.692 | -.607 | -.678 | -.701 |
| 38 | -.925 | -.980 | -.801 | -.956 | -.992 |
| 39 | -1.107 | -1.142 | -1.054 | -.998 | -1.105 |
| 40 | -1.027 | -1.058 | -.969 | -1.087 | -1.099 |
| 41 | -1.142 | -1.128 | -1.102 | -1.080 | -1.138 |
| 42 | -1.012 | -1.016 | -1.030 | -1.076 | -1.062 |

Figure 27 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.208 | -1.183 | -1.183 | -1.197 | -1.215 |
| 44 | -1.024 | -1.006 | -1.075 | -1.063 | -1.046 |
| 45 | -.877 | -.833 | -.909 | -.864 | -.844 |
| 46 | -.489 | -.447 | -.547 | -.481 | -.440 |
| 47 | -.040 | .013 | -.101 | -.029 | .010 |
| 48 | .251 | .289 | .218 | .279 | .304 |
| 49 | .408 | .414 | .416 | .424 | .429 |
| 50 | .432 | .406 | .438 | .415 | .419 |
| 51 | .265 | .244 | .318 | .270 | .257 |
| 52 | .067 | .016 | .118 | .051 | .039 |
| 53 | -.172 | -.224 | -.110 | -.193 | -.212 |
| 54 | -.501 | -.548 | -.410 | -.488 | -.521 |
| 55 | -.704 | -.743 | -.637 | -.712 | -.732 |
| 56 | -.969 | -1.006 | -.892 | -.947 | -.950 |
| 57 | -.965 | -.989 | -.960 | -.985 | -.943 |
| 58 | -.938 | -.949 | -.896 | -.915 | -.891 |
| 59 | -.909 | -.941 | -.900 | -.934 | -.908 |
| 60 | -.955 | -.969 | -.912 | -.932 | -.913 |
| 61 | -.904 | -.945 | -.889 | -.929 | -.925 |
| 62 | -.977 | -.967 | -.945 | -.923 | -.915 |
| 63 | -.876 | -.915 | -.864 | -1.019 | -.951 |
| 64 | -1.032 | -1.007 | -.979 | -.965 | -1.027 |
| 65 | -.934 | -.874 | -.967 | -.936 | -.883 |
| 66 | -.721 | -.651 | -.752 | -.641 | -.643 |
| 67 | -.582 | -.575 | -.611 | -.574 | -.581 |
| 68 | -.735 | -.759 | -.670 | -.789 | -.783 |
| 69 | -.908 | -.891 | -.945 | -1.034 | -.946 |
| 70 | -1.032 | -1.039 | -.999 | -1.146 | -1.053 |
| 71 | -.924 | -.918 | -.975 | -1.003 | -.954 |
| 72 | -1.098 | -1.059 | -1.084 | -1.094 | -1.094 |
| 73 | -.943 | -.927 | -.988 | -.978 | -.970 |
| 74 | -1.040 | -1.000 | -1.034 | -1.016 | -1.037 |
| 75 | -.957 | -.940 | -.986 | -.987 | -.995 |
| 76 | -1.120 | -1.086 | -1.093 | -1.099 | -1.135 |
| 77 | -1.051 | -1.033 | -1.090 | -1.085 | -1.086 |
| 78 | -1.072 | -1.021 | -1.089 | -1.046 | -1.056 |
| 79 | -.845 | -.807 | -.912 | -.858 | -.844 |
| 80 | -.637 | -.575 | -.692 | -.617 | -.591 |
| 81 | -.343 | -.300 | -.411 | -.352 | -.305 |
| 82 | -.048 | -.014 | -.117 | -.043 | -.012 |
| 83 | .159 | .182 | .114 | .165 | .191 |
| 84 | .396 | .408 | .357 | .389 | .411 |

Figure 27 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .679 | .681 | .669 | .676 | .689 |
| 2 | .545 | .529 | .565 | .552 | .535 |
| 3 | .272 | .255 | .337 | .305 | .248 |
| 4 | -.148 | -.151 | -.075 | -.109 | -.176 |
| 5 | -.568 | -.557 | -.486 | -.523 | -.600 |
| 6 | -.588 | -.534 | -.520 | -.477 | -.513 |
| 7 | -.800 | -.811 | -.826 | -.737 | -.794 |
| 8 | -1.094 | -1.102 | -.795 | -.693 | -.805 |
| 9 | -1.214 | -1.092 | -1.330 | -1.136 | -1.262 |
| 10 | -1.009 | -.989 | -.909 | -.854 | -.903 |
| 11 | -.817 | -.771 | -.900 | -.833 | -.831 |
| 12 | -.631 | -.636 | -.644 | -.614 | -.640 |
| 13 | -.650 | -.708 | -.640 | -.577 | -.766 |
| 14 | -1.015 | -1.000 | -.969 | -1.157 | -1.058 |
| 15 | -1.050 | -1.067 | -1.084 | -1.109 | -1.117 |
| 16 | -1.113 | -1.104 | -1.096 | -1.219 | -1.165 |
| 17 | -1.126 | -1.135 | -1.156 | -1.227 | -1.187 |
| 18 | -1.146 | -1.163 | -1.130 | -1.230 | -1.217 |
| 19 | -1.221 | -1.220 | -1.253 | -1.314 | -1.274 |
| 20 | -1.005 | -.997 | -1.040 | -1.067 | -1.037 |
| 21 | -.719 | -.699 | -.797 | -.786 | -.728 |
| 22 | -.273 | -.257 | -.350 | -.308 | -.264 |
| 23 | .199 | .224 | .125 | .168 | .221 |
| 24 | .477 | .492 | .423 | .462 | .496 |
| 25 | .496 | .498 | .495 | .485 | .508 |
| 26 | .404 | .388 | .415 | .413 | .391 |
| 27 | .166 | .142 | .219 | .176 | .132 |
| 28 | -.195 | -.226 | -.142 | -.175 | -.255 |
| 29 | -.685 | -.692 | -.603 | -.637 | -.719 |
| 30 | -.960 | -.955 | -.886 | -.877 | -.978 |
| 31 | -1.161 | -1.130 | -1.132 | -1.060 | -1.126 |
| 32 | -1.045 | -1.014 | -1.013 | -.954 | -1.020 |
| 33 | -1.203 | -1.189 | -1.169 | -1.099 | -1.180 |
| 34 | -.668 | -.647 | -.689 | -.583 | -.643 |
| 35 | -.676 | -.646 | -.676 | -.602 | -.637 |
| 36 | -.453 | -.443 | -.451 | -.415 | -.428 |
| 37 | -.499 | -.558 | -.489 | -.502 | -.550 |
| 38 | -.983 | -1.064 | -.933 | -.949 | -1.101 |
| 39 | -1.070 | -1.070 | -1.060 | -1.002 | -1.111 |
| 40 | -1.070 | -1.091 | -1.046 | -1.132 | -1.151 |
| 41 | -1.085 | -1.080 | -1.099 | -1.138 | -1.135 |
| 42 | -1.000 | -1.010 | -.999 | -1.114 | -1.066 |

Figure 28 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.153 | -1.161 | -1.165 | -1.281 | -1.216 |
| 44 | -.994 | -.978 | -1.017 | -1.077 | -1.021 |
| 45 | -.814 | -.790 | -.867 | -.893 | -.816 |
| 46 | -.442 | -.407 | -.492 | -.471 | -.409 |
| 47 | .001 | .035 | -.050 | -.032 | .044 |
| 48 | .279 | .304 | .243 | .268 | .314 |
| 49 | .414 | .411 | .404 | .414 | .415 |
| 50 | .385 | .368 | .413 | .392 | .385 |
| 51 | .209 | .191 | .257 | .230 | .189 |
| 52 | -.028 | -.059 | .041 | -.008 | -.055 |
| 53 | -.283 | -.314 | -.207 | -.243 | -.323 |
| 54 | -.638 | -.660 | -.554 | -.589 | -.667 |
| 55 | -.862 | -.882 | -.777 | -.795 | -.894 |
| 56 | -1.171 | -1.160 | -1.092 | -1.076 | -1.172 |
| 57 | -1.236 | -1.219 | -1.172 | -1.116 | -1.214 |
| 58 | -1.231 | -1.196 | -1.208 | -1.106 | -1.180 |
| 59 | -1.163 | -1.128 | -1.112 | -1.041 | -1.120 |
| 60 | -1.218 | -1.191 | -1.190 | -1.107 | -1.182 |
| 61 | -1.157 | -1.138 | -1.110 | -1.048 | -1.141 |
| 62 | -1.355 | -1.225 | -1.286 | -1.285 | -1.223 |
| 63 | -1.119 | -1.102 | -1.027 | -1.069 | -1.118 |
| 64 | -.564 | -.519 | -.527 | -.523 | -.533 |
| 65 | -.531 | -.517 | -.494 | -.498 | -.542 |
| 66 | -.400 | -.413 | -.393 | -.365 | -.414 |
| 67 | -.500 | -.543 | -.464 | -.514 | -.557 |
| 68 | -.836 | -.951 | -.774 | -.989 | -.975 |
| 69 | -.913 | -.976 | -.903 | -.988 | -.961 |
| 70 | -1.009 | -1.109 | -1.013 | -1.188 | -1.139 |
| 71 | -.878 | -.875 | -.882 | -1.016 | -.924 |
| 72 | -.964 | -.977 | -.989 | -1.089 | -1.020 |
| 73 | -.851 | -.843 | -.859 | -.964 | -.890 |
| 74 | -.895 | -.901 | -.920 | -1.019 | -.951 |
| 75 | -.855 | -.852 | -.857 | -.961 | -.899 |
| 76 | -.973 | -.979 | -.975 | -1.124 | -1.044 |
| 77 | -.946 | -.938 | -.951 | -1.043 | -.992 |
| 78 | -.928 | -.916 | -.970 | -1.035 | -.960 |
| 79 | -.742 | -.712 | -.779 | -.800 | -.749 |
| 80 | -.518 | -.493 | -.588 | -.577 | -.512 |
| 81 | -.259 | -.239 | -.318 | -.287 | -.248 |
| 82 | .020 | .051 | -.044 | -.018 | .055 |
| 83 | .205 | .227 | .158 | .184 | .228 |
| 84 | .415 | .417 | .383 | .400 | .436 |

Figure 28 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .678 | .663 | .678 | .665 | .685 |
| 2 | .550 | .542 | .559 | .563 | .538 |
| 3 | .296 | .270 | .313 | .354 | .260 |
| 4 | -.088 | -.097 | -.041 | -.002 | -.139 |
| 5 | -.472 | -.463 | -.439 | -.358 | -.538 |
| 6 | -.402 | -.394 | -.362 | -.291 | -.432 |
| 7 | -1.239 | -1.106 | -.992 | -.616 | -.984 |
| 8 | -1.205 | -1.219 | -1.330 | -.903 | -1.269 |
| 9 | -1.440 | -1.398 | -1.590 | -1.039 | -1.542 |
| 10 | -.971 | -.944 | -1.074 | -.880 | -1.043 |
| 11 | -.850 | -.809 | -.960 | -.793 | -.900 |
| 12 | -.559 | -.579 | -.599 | -.599 | -.630 |
| 13 | -.599 | -.710 | -.596 | -.596 | -.757 |
| 14 | -1.041 | -1.055 | -.998 | -1.217 | -1.074 |
| 15 | -1.063 | -1.074 | -1.105 | -1.116 | -1.099 |
| 16 | -1.126 | -1.138 | -1.085 | -1.305 | -1.147 |
| 17 | -1.149 | -1.135 | -1.158 | -1.295 | -1.158 |
| 18 | -1.157 | -1.185 | -1.101 | -1.340 | -1.194 |
| 19 | -1.239 | -1.220 | -1.244 | -1.386 | -1.240 |
| 20 | -1.018 | -1.016 | -1.017 | -1.145 | -1.027 |
| 21 | -.736 | -.706 | -.800 | -.848 | -.705 |
| 22 | -.285 | -.269 | -.355 | -.368 | -.257 |
| 23 | .192 | .213 | .126 | .123 | .225 |
| 24 | .471 | .486 | .416 | .427 | .500 |
| 25 | .491 | .487 | .487 | .481 | .500 |
| 26 | .400 | .396 | .412 | .411 | .398 |
| 27 | .170 | .144 | .226 | .215 | .139 |
| 28 | -.188 | -.207 | -.124 | -.118 | -.232 |
| 29 | -.633 | -.647 | -.606 | -.538 | -.697 |
| 30 | -.894 | -.902 | -.868 | -.765 | -.951 |
| 31 | -1.050 | -1.005 | -1.074 | -.865 | -1.065 |
| 32 | -.914 | -.899 | -.920 | -.772 | -.960 |
| 33 | -.952 | -.876 | -1.041 | -.783 | -.924 |
| 34 | -.797 | -.735 | -.871 | -.556 | -.867 |
| 35 | -.615 | -.568 | -.672 | -.492 | -.620 |
| 36 | -.565 | -.524 | -.609 | -.432 | -.591 |
| 37 | -.517 | -.575 | -.551 | -.527 | -.614 |
| 38 | -.990 | -1.119 | -.897 | -1.218 | -1.121 |
| 39 | -1.089 | -1.078 | -1.047 | -1.033 | -1.098 |
| 40 | -1.069 | -1.130 | -1.008 | -1.213 | -1.139 |
| 41 | -1.100 | -1.084 | -1.077 | -1.201 | -1.102 |
| 42 | -1.008 | -1.027 | -.973 | -1.204 | -1.031 |

Figure 29 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.172 | -1.160 | -1.149 | -1.349 | -1.173 |
| 44 | -.996 | -.999 | -.997 | -1.149 | -1.001 |
| 45 | -.823 | -.791 | -.860 | -.945 | -.802 |
| 46 | -.431 | -.412 | -.484 | -.514 | -.415 |
| 47 | -.013 | .024 | -.042 | -.052 | .041 |
| 48 | .273 | .305 | .246 | .253 | .317 |
| 49 | .408 | .414 | .396 | .403 | .420 |
| 50 | .385 | .368 | .416 | .397 | .372 |
| 51 | .216 | .206 | .244 | .242 | .192 |
| 52 | -.016 | -.047 | .047 | .028 | -.060 |
| 53 | -.263 | -.292 | -.206 | -.211 | -.319 |
| 54 | -.608 | -.620 | -.550 | -.513 | -.668 |
| 55 | -.833 | -.850 | -.776 | -.725 | -.896 |
| 56 | -1.120 | -1.110 | -1.099 | -.983 | -1.164 |
| 57 | -1.190 | -1.184 | -1.164 | -1.029 | -1.225 |
| 58 | -1.189 | -1.131 | -1.221 | -.976 | -1.190 |
| 59 | -1.107 | -1.086 | -1.108 | -.925 | -1.134 |
| 60 | -1.180 | -1.133 | -1.203 | -.994 | -1.181 |
| 61 | -1.120 | -1.116 | -1.099 | -.973 | -1.161 |
| 62 | -.971 | -1.059 | -1.151 | -.857 | -1.104 |
| 63 | -1.125 | -1.129 | -1.048 | -.990 | -1.152 |
| 64 | -.500 | -.516 | -.612 | -.404 | -.600 |
| 65 | -.539 | -.535 | -.547 | -.450 | -.588 |
| 66 | -.411 | -.432 | -.448 | -.356 | -.461 |
| 67 | -.504 | -.576 | -.511 | -.521 | -.591 |
| 68 | -.843 | -.960 | -.817 | -1.080 | -.968 |
| 69 | -.907 | -.985 | -.909 | -1.051 | -.968 |
| 70 | -1.017 | -1.094 | -.992 | -1.229 | -1.110 |
| 71 | -.881 | -.918 | -.873 | -1.103 | -.912 |
| 72 | -.995 | -.980 | -.981 | -1.145 | -.998 |
| 73 | -.879 | -.884 | -.847 | -1.043 | -.876 |
| 74 | -.930 | -.917 | -.930 | -1.089 | -.923 |
| 75 | -.879 | -.886 | -.847 | -1.048 | -.884 |
| 76 | -1.010 | -1.007 | -.974 | -1.200 | -1.011 |
| 77 | -.962 | -.968 | -.940 | -1.117 | -.972 |
| 78 | -.963 | -.933 | -.973 | -1.106 | -.938 |
| 79 | -.756 | -.737 | -.778 | -.864 | -.746 |
| 80 | -.542 | -.500 | -.591 | -.626 | -.495 |
| 81 | -.269 | -.237 | -.323 | -.336 | -.236 |
| 82 | .003 | .024 | -.046 | -.047 | .054 |
| 83 | .197 | .212 | .148 | .153 | .234 |
| 84 | .398 | .403 | .389 | .384 | .425 |

Figure 29 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .660 | .658 | .661 | .663 | .670 |
| 2 | .596 | .596 | .543 | .585 | .576 |
| 3 | .403 | .423 | .299 | .383 | .323 |
| 4 | -.013 | .020 | -.135 | -.040 | -.112 |
| 5 | -.485 | -.454 | -.606 | -.506 | -.575 |
| 6 | -.780 | -.695 | -.893 | -.810 | -.835 |
| 7 | -1.075 | -.937 | -1.180 | -1.115 | -1.095 |
| 8 | -.753 | -.780 | -.960 | -.860 | -.830 |
| 9 | -.817 | -1.434 | -.823 | -.824 | -.816 |
| 10 | -.785 | -.782 | -.784 | -.766 | -.779 |
| 11 | -1.173 | -1.017 | -1.290 | -1.257 | -1.302 |
| 12 | -.827 | -.669 | -.644 | -.781 | -.651 |
| 13 | -.651 | -.750 | -.777 | -.654 | -.721 |
| 14 | -.966 | -.995 | -.965 | -1.019 | -1.037 |
| 15 | -1.050 | -1.091 | -1.038 | -1.055 | -1.027 |
| 16 | -1.093 | -1.101 | -1.088 | -1.096 | -1.129 |
| 17 | -1.180 | -1.137 | -1.139 | -1.156 | -1.139 |
| 18 | -1.169 | -1.085 | -1.146 | -1.121 | -1.171 |
| 19 | -1.319 | -1.262 | -1.263 | -1.280 | -1.267 |
| 20 | -1.101 | -1.045 | -1.023 | -1.063 | -1.051 |
| 21 | -.863 | -.840 | -.740 | -.825 | -.755 |
| 22 | -.431 | -.411 | -.310 | -.403 | -.329 |
| 23 | .064 | .067 | .170 | .087 | .158 |
| 24 | .397 | .397 | .448 | .399 | .452 |
| 25 | .479 | .469 | .489 | .482 | .500 |
| 26 | .462 | .445 | .416 | .440 | .434 |
| 27 | .297 | .276 | .211 | .289 | .232 |
| 28 | -.047 | -.083 | -.148 | -.073 | -.144 |
| 29 | -.485 | -.533 | -.600 | -.521 | -.581 |
| 30 | -.764 | -.823 | -.845 | -.804 | -.848 |
| 31 | -.963 | -1.135 | -.998 | -1.012 | -.990 |
| 32 | -.924 | -1.063 | -.976 | -.966 | -.976 |
| 33 | -1.039 | -1.090 | -1.081 | -1.080 | -1.068 |
| 34 | -.911 | -.707 | -1.009 | -.928 | -.934 |
| 35 | -1.178 | -.741 | -1.191 | -1.169 | -1.101 |
| 36 | -.847 | -.491 | -.723 | -.811 | -.744 |
| 37 | -.627 | -.541 | -.630 | -.630 | -.614 |
| 38 | -.776 | -.935 | -.929 | -.795 | -.921 |
| 39 | -1.211 | -1.116 | -1.123 | -1.052 | -1.102 |
| 40 | -.972 | -1.020 | -1.023 | -.923 | -1.032 |
| 41 | -1.189 | -1.097 | -1.124 | -1.107 | -1.125 |
| 42 | -1.046 | -.962 | -1.002 | -.985 | -1.028 |

Figure 30 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.215 | -1.117 | -1.177 | -1.165 | -1.180 |
| 44 | -1.109 | -1.029 | -1.023 | -1.062 | -1.063 |
| 45 | -.958 | -.891 | -.849 | -.915 | -.868 |
| 46 | -.597 | -.539 | -.467 | -.560 | -.511 |
| 47 | -.148 | -.117 | -.040 | -.124 | -.053 |
| 48 | .194 | .204 | .264 | .197 | .255 |
| 49 | .430 | .437 | .436 | .440 | .450 |
| 50 | .412 | .387 | .381 | .402 | .395 |
| 51 | .355 | .319 | .274 | .337 | .302 |
| 52 | .178 | .109 | .074 | .142 | .103 |
| 53 | -.080 | -.172 | -.211 | -.112 | -.176 |
| 54 | -.314 | -.434 | -.444 | -.353 | -.413 |
| 55 | -.639 | -.791 | -.758 | -.683 | -.738 |
| 56 | -.794 | -.963 | -.879 | -.829 | -.870 |
| 57 | -.949 | -1.236 | -1.027 | -1.025 | -1.007 |
| 58 | -.867 | -1.215 | -.920 | -.929 | -.910 |
| 59 | -.885 | -1.250 | -.951 | -.968 | -.938 |
| 60 | -.830 | -1.148 | -.883 | -.894 | -.875 |
| 61 | -.924 | -1.194 | -.986 | -.994 | -.977 |
| 62 | -.865 | -1.165 | -.907 | -.937 | -.912 |
| 63 | -.941 | -.844 | -1.009 | -.963 | -.993 |
| 64 | -.901 | -.565 | -.920 | -.944 | -.931 |
| 65 | -.977 | -.498 | -.984 | -1.022 | -1.004 |
| 66 | -.909 | -.397 | -.975 | -.918 | -1.003 |
| 67 | -.587 | -.477 | -.580 | -.618 | -.596 |
| 68 | -.601 | -.659 | -.742 | -.678 | -.756 |
| 69 | -1.030 | -1.049 | -1.035 | -.991 | -.997 |
| 70 | -1.016 | -.941 | -.971 | -.910 | -.952 |
| 71 | -1.250 | -1.037 | -1.015 | -1.036 | -1.001 |
| 72 | -1.067 | -.883 | -1.016 | -1.014 | -1.045 |
| 73 | -1.126 | -.934 | -1.013 | -1.028 | -1.025 |
| 74 | -1.019 | -.834 | -.954 | -.965 | -.991 |
| 75 | -1.061 | -.897 | -1.003 | -1.005 | -1.027 |
| 76 | -1.058 | -.865 | -1.010 | -.999 | -1.057 |
| 77 | -1.185 | -1.012 | -1.119 | -1.132 | -1.141 |
| 78 | -1.090 | -.917 | -.987 | -1.027 | -1.036 |
| 79 | -.994 | -.862 | -.882 | -.951 | -.902 |
| 80 | -.741 | -.628 | -.609 | -.694 | -.641 |
| 81 | -.468 | -.388 | -.337 | -.438 | -.370 |
| 82 | -.184 | -.131 | -.064 | -.154 | -.105 |
| 83 | .108 | .150 | .193 | .128 | .184 |
| 84 | .274 | .288 | .325 | .286 | .319 |

Figure 30 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .658 | .652 | .671 | .658 | .680 |
| 2 | .568 | .587 | .531 | .580 | .557 |
| 3 | .350 | .403 | .265 | .369 | .310 |
| 4 | -.075 | .003 | -.185 | -.050 | -.122 |
| 5 | -.561 | -.474 | -.654 | -.502 | -.595 |
| 6 | -.855 | -.751 | -.913 | -.778 | -.838 |
| 7 | -1.149 | -1.028 | -1.172 | -1.054 | -1.081 |
| 8 | -.797 | -.744 | -.820 | -.776 | -.766 |
| 9 | -.998 | -1.049 | -.857 | -.751 | -.829 |
| 10 | -.850 | -.711 | -.843 | -.851 | -.762 |
| 11 | -.907 | -.912 | -.871 | -.869 | -.816 |
| 12 | -.679 | -.652 | -.674 | -.653 | -.628 |
| 13 | -.721 | -.751 | -.781 | -.623 | -.774 |
| 14 | -.912 | -.924 | -.951 | -1.082 | -1.005 |
| 15 | -1.040 | -1.059 | -1.052 | -1.110 | -1.086 |
| 16 | -1.043 | -1.054 | -1.056 | -1.154 | -1.098 |
| 17 | -1.093 | -1.103 | -1.107 | -1.177 | -1.140 |
| 18 | -1.068 | -1.068 | -1.104 | -1.138 | -1.127 |
| 19 | -1.214 | -1.227 | -1.219 | -1.289 | -1.262 |
| 20 | -1.003 | -1.032 | -.966 | -1.056 | -.999 |
| 21 | -.765 | -.808 | -.689 | -.814 | -.735 |
| 22 | -.350 | -.407 | -.258 | -.377 | -.286 |
| 23 | .117 | .067 | .208 | .095 | .187 |
| 24 | .418 | .384 | .472 | .413 | .470 |
| 25 | .490 | .465 | .496 | .475 | .509 |
| 26 | .422 | .435 | .394 | .434 | .408 |
| 27 | .238 | .273 | .166 | .246 | .201 |
| 28 | -.143 | -.087 | -.224 | -.117 | -.185 |
| 29 | -.615 | -.554 | -.698 | -.569 | -.661 |
| 30 | -.918 | -.870 | -.965 | -.860 | -.919 |
| 31 | -1.225 | -1.187 | -1.226 | -1.124 | -1.181 |
| 32 | -1.149 | -1.122 | -1.138 | -1.062 | -1.089 |
| 33 | -1.218 | -1.149 | -1.209 | -1.116 | -1.154 |
| 34 | -1.001 | -.921 | -.860 | -.927 | -.763 |
| 35 | -.752 | -.647 | -.744 | -.691 | -.690 |
| 36 | -.622 | -.561 | -.560 | -.578 | -.488 |
| 37 | -.510 | -.476 | -.543 | -.461 | -.495 |
| 38 | -.916 | -.921 | -1.034 | -.845 | -1.044 |
| 39 | -1.031 | -1.068 | -1.000 | -1.022 | -1.112 |
| 40 | -.990 | -1.003 | -1.050 | -1.023 | -1.081 |
| 41 | -1.045 | -1.059 | -1.039 | -1.097 | -1.113 |
| 42 | -.942 | -.959 | -.954 | -1.015 | -.995 |

Figure 31 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.093 | -1.094 | -1.119 | -1.185 | -1.163 |
| 44 | -.994 | -1.024 | -.964 | -1.060 | -1.004 |
| 45 | -.833 | -.873 | -.784 | -.899 | -.824 |
| 46 | -.497 | -.541 | -.410 | -.523 | -.444 |
| 47 | -.062 | -.126 | -.006 | -.113 | -.006 |
| 48 | .233 | .196 | .270 | .211 | .280 |
| 49 | .444 | .429 | .441 | .435 | .454 |
| 50 | .373 | .390 | .360 | .389 | .368 |
| 51 | .280 | .303 | .225 | .300 | .249 |
| 52 | .054 | .102 | .001 | .095 | .021 |
| 53 | -.228 | -.184 | -.315 | -.197 | -.284 |
| 54 | -.491 | -.439 | -.572 | -.453 | -.539 |
| 55 | -.864 | -.809 | -.946 | -.801 | -.913 |
| 56 | -1.032 | -.985 | -1.084 | -.964 | -1.043 |
| 57 | -1.298 | -1.265 | -1.306 | -1.192 | -1.267 |
| 58 | -1.258 | -1.251 | -1.214 | -1.136 | -1.171 |
| 59 | -1.274 | -1.261 | -1.225 | -1.145 | -1.200 |
| 60 | -1.175 | -1.166 | -1.136 | -1.072 | -1.108 |
| 61 | -1.257 | -1.220 | -1.231 | -1.149 | -1.209 |
| 62 | -1.212 | -1.185 | -1.239 | -1.140 | -1.225 |
| 63 | -1.348 | -1.337 | -1.099 | -1.182 | -1.041 |
| 64 | -.608 | -.572 | -.621 | -.603 | -.559 |
| 65 | -.605 | -.583 | -.582 | -.576 | -.878 |
| 66 | -.439 | -.489 | -.433 | -.478 | -.400 |
| 67 | -.458 | -.467 | -.503 | -.466 | -.592 |
| 68 | -.634 | -.685 | -.742 | -.681 | -.716 |
| 69 | -1.052 | -1.018 | -1.066 | -.966 | -.939 |
| 70 | -.946 | -.944 | -.987 | -.925 | -.920 |
| 71 | -1.053 | -1.021 | -.925 | -.961 | -.903 |
| 72 | -.874 | -.880 | -.897 | -.959 | -.919 |
| 73 | -.900 | -.902 | -.872 | -.961 | -.915 |
| 74 | -.809 | -.824 | -.824 | -.911 | -.864 |
| 75 | -.848 | -.857 | -.866 | -.951 | -.911 |
| 76 | -.840 | -.841 | -.883 | -.947 | -.920 |
| 77 | -.957 | -.969 | -.976 | -1.084 | -1.028 |
| 78 | -.875 | -.898 | -.862 | -.963 | -.898 |
| 79 | -.798 | -.838 | -.760 | -.882 | -.798 |
| 80 | -.577 | -.616 | -.512 | -.622 | -.537 |
| 81 | -.319 | -.364 | -.246 | -.367 | -.266 |
| 82 | -.074 | -.105 | .000 | -.090 | -.011 |
| 83 | .199 | .155 | .243 | .164 | .240 |
| 84 | .316 | .298 | .351 | .308 | .343 |

Figure 31 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .657 | .659 | .670 | .655 | .684 |
| 2 | .581 | .605 | .539 | .583 | .555 |
| 3 | .357 | .412 | .275 | .405 | .311 |
| 4 | -.062 | .002 | -.175 | .012 | -.128 |
| 5 | -.546 | -.492 | -.663 | -.427 | -.582 |
| 6 | -.793 | -.820 | -.892 | -.656 | -.794 |
| 7 | -1.041 | -1.148 | -1.121 | -.885 | -1.007 |
| 8 | -.809 | -.843 | -.910 | -.645 | -.831 |
| 9 | -1.564 | -.847 | -1.414 | -.808 | -1.365 |
| 10 | -.872 | -.819 | -.918 | -.854 | -.882 |
| 11 | -1.083 | -1.203 | -1.067 | -.840 | -.990 |
| 12 | -.703 | -.896 | -.702 | -.676 | -.661 |
| 13 | -.685 | -.640 | -.804 | -.620 | -.808 |
| 14 | -.992 | -.961 | -1.005 | -1.153 | -1.019 |
| 15 | -1.056 | -.994 | -1.060 | -1.163 | -1.098 |
| 16 | -1.091 | -1.044 | -1.100 | -1.241 | -1.106 |
| 17 | -1.116 | -1.111 | -1.096 | -1.277 | -1.138 |
| 18 | -1.093 | -1.086 | -1.119 | -1.241 | -1.124 |
| 19 | -1.231 | -1.249 | -1.212 | -1.395 | -1.260 |
| 20 | -1.025 | -1.048 | -.980 | -1.139 | -.999 |
| 21 | -.775 | -.823 | -.692 | -.891 | -.736 |
| 22 | -.355 | -.404 | -.264 | -.431 | -.292 |
| 23 | .111 | .064 | .205 | .055 | .187 |
| 24 | .428 | .397 | .479 | .380 | .468 |
| 25 | .476 | .479 | .492 | .471 | .503 |
| 26 | .435 | .450 | .405 | .432 | .416 |
| 27 | .147 | .303 | .162 | .274 | .192 |
| 28 | -.310 | -.055 | -.223 | -.079 | -.192 |
| 29 | -.621 | -.509 | -.706 | -.514 | -.652 |
| 30 | -.938 | -.818 | -.984 | -.779 | -.909 |
| 31 | -1.195 | -1.053 | -1.200 | -1.005 | -1.161 |
| 32 | -1.132 | -1.006 | -1.145 | -.940 | -1.073 |
| 33 | -1.157 | -1.117 | -1.196 | -.993 | -1.108 |
| 34 | -.789 | -1.028 | -.864 | -.587 | -.819 |
| 35 | -.841 | -1.332 | -.840 | -.656 | -.776 |
| 36 | -.554 | -.883 | -.620 | -.428 | -.550 |
| 37 | -.564 | -.658 | -.612 | -.463 | -.570 |
| 38 | -.931 | -.813 | -1.089 | -.915 | -1.092 |
| 39 | -1.067 | -1.094 | -1.013 | -1.088 | -1.083 |
| 40 | -.992 | -.914 | -1.079 | -1.133 | -1.096 |
| 41 | -1.064 | -1.075 | -1.040 | -1.198 | -1.085 |
| 42 | -.958 | -.952 | -.971 | -1.115 | -.986 |

Figure 32 - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.108 | -1.085 | -1.109 | -1.293 | -1.147 |
| 44 | -1.025 | -1.033 | -.982 | -1.139 | -.999 |
| 45 | -.845 | -.880 | -.774 | -.978 | -.814 |
| 46 | -.504 | -.560 | -.414 | -.584 | -.434 |
| 47 | -.092 | -.116 | .004 | -.142 | -.011 |
| 48 | .224 | .206 | .289 | .192 | .279 |
| 49 | .439 | .435 | .445 | .439 | .454 |
| 50 | .381 | .413 | .360 | .386 | .371 |
| 51 | .284 | .339 | .224 | .313 | .247 |
| 52 | .071 | .153 | -.005 | .108 | .018 |
| 53 | -.220 | -.110 | -.317 | -.160 | -.288 |
| 54 | -.492 | -.360 | -.588 | -.408 | -.539 |
| 55 | -.846 | -.707 | -.928 | -.734 | -.898 |
| 56 | -1.024 | -.870 | -1.090 | -.881 | -1.036 |
| 57 | -1.270 | -1.084 | -1.298 | -1.117 | -1.267 |
| 58 | -1.248 | -1.007 | -1.231 | -1.041 | -1.181 |
| 59 | -1.260 | -1.029 | -1.236 | -1.056 | -1.200 |
| 60 | -1.181 | -.965 | -1.163 | -.978 | -1.110 |
| 61 | -1.208 | -1.058 | -1.223 | -1.060 | -1.194 |
| 62 | -1.253 | -.994 | -1.237 | -1.071 | -1.164 |
| 63 | -.862 | -1.104 | -.691 | -.775 | -.651 |
| 64 | -.623 | -1.061 | -.697 | -.524 | -.616 |
| 65 | -.527 | -1.163 | -.532 | -.440 | -.524 |
| 66 | -.426 | -1.162 | -.513 | -.392 | -.433 |
| 67 | -.491 | -.596 | -.535 | -.452 | -.533 |
| 68 | -.726 | -.687 | -.827 | -.728 | -.781 |
| 69 | -1.125 | -.937 | -1.201 | -1.047 | -1.083 |
| 70 | -.973 | -.888 | -1.033 | -1.031 | -1.008 |
| 71 | -1.000 | -1.080 | -.997 | -1.046 | -.966 |
| 72 | -.898 | -.891 | -.914 | -1.053 | -.931 |
| 73 | -.904 | -.979 | -.884 | -1.049 | -.913 |
| 74 | -.846 | -.869 | -.836 | -.990 | -.862 |
| 75 | -.875 | -.907 | -.870 | -1.042 | -.903 |
| 76 | -.883 | -.889 | -.898 | -1.043 | -.918 |
| 77 | -.991 | -1.023 | -.981 | -1.173 | -1.021 |
| 78 | -.918 | -.958 | -.877 | -1.036 | -.905 |
| 79 | -.817 | -.889 | -.755 | -.951 | -.794 |
| 80 | -.592 | -.668 | -.520 | -.677 | -.543 |
| 81 | -.337 | -.407 | -.250 | -.416 | -.276 |
| 82 | -.087 | -.140 | -.005 | -.141 | -.025 |
| 83 | .181 | .140 | .241 | .153 | .244 |
| 84 | .314 | .294 | .345 | .291 | .348 |

Figure 32 (continued) - Pressure Coefficients
 $\alpha = 40^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .806 | .815 | .797 | .792 | .815 |
| 2 | .665 | .629 | .693 | .659 | .629 |
| 3 | .333 | .250 | .359 | .307 | .269 |
| 4 | -.265 | -.356 | -.241 | -.276 | -.334 |
| 5 | -.864 | -.963 | -.842 | -.860 | -.937 |
| 6 | -.844 | -.899 | -.868 | -.839 | -.866 |
| 7 | -1.100 | -1.158 | -1.162 | -1.110 | -1.108 |
| 8 | -.965 | -1.039 | -1.020 | -.998 | -.984 |
| 9 | -1.433 | -1.493 | -1.521 | -1.419 | -1.441 |
| 10 | -1.425 | -1.555 | -1.499 | -1.536 | -1.484 |
| 11 | -1.799 | -1.904 | -1.750 | -1.888 | -1.841 |
| 12 | -1.064 | -.926 | -1.034 | -1.030 | -1.006 |
| 13 | -.875 | -1.043 | -.926 | -.907 | -.903 |
| 14 | -1.625 | -1.499 | -1.304 | -1.598 | -1.683 |
| 15 | -1.565 | -1.498 | -1.336 | -1.577 | -1.579 |
| 16 | -1.656 | -1.579 | -1.521 | -1.642 | -1.681 |
| 17 | -1.682 | -1.587 | -1.529 | -1.655 | -1.684 |
| 18 | -1.582 | -1.537 | -1.519 | -1.548 | -1.590 |
| 19 | -1.802 | -1.716 | -1.705 | -1.752 | -1.810 |
| 20 | -1.498 | -1.417 | -1.490 | -1.474 | -1.470 |
| 21 | -1.203 | -1.106 | -1.199 | -1.169 | -1.170 |
| 22 | -.568 | -.492 | -.626 | -.561 | -.530 |
| 23 | .057 | .151 | .030 | .071 | .115 |
| 24 | .484 | .544 | .471 | .495 | .515 |
| 25 | .639 | .670 | .646 | .636 | .654 |
| 26 | .585 | .550 | .601 | .571 | .558 |
| 27 | .319 | .244 | .319 | .300 | .280 |
| 28 | -.132 | -.233 | -.145 | -.164 | -.183 |
| 29 | -.724 | -.845 | -.766 | -.731 | -.768 |
| 30 | -1.006 | -1.123 | -1.140 | -1.036 | -1.015 |
| 31 | -1.187 | -1.300 | -1.343 | -1.212 | -1.200 |
| 32 | -1.185 | -1.289 | -1.335 | -1.220 | -1.186 |
| 33 | -1.227 | -1.338 | -1.372 | -1.252 | -1.236 |
| 34 | -1.168 | -1.311 | -1.300 | -1.158 | -1.250 |
| 35 | -1.314 | -1.419 | -1.441 | -1.255 | -1.369 |
| 36 | -1.074 | -.998 | -1.252 | -1.049 | -1.039 |
| 37 | -.817 | -.819 | -.758 | -.822 | -.856 |
| 38 | -1.333 | -1.381 | -1.521 | -1.312 | -1.373 |
| 39 | -1.351 | -1.407 | -1.435 | -1.305 | -1.368 |
| 40 | -1.475 | -1.403 | -1.449 | -1.478 | -1.498 |
| 41 | -1.537 | -1.437 | -1.469 | -1.485 | -1.551 |
| 42 | -1.451 | -1.313 | -1.342 | -1.450 | -1.459 |

Figure 33 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.689 | -1.520 | -1.503 | -1.640 | -1.717 |
| 44 | -1.498 | -1.358 | -1.425 | -1.488 | -1.468 |
| 45 | -1.303 | -1.129 | -1.217 | -1.265 | -1.270 |
| 46 | -.756 | -.627 | -.751 | -.756 | -.689 |
| 47 | -.154 | -.042 | -.138 | -.120 | -.104 |
| 48 | .298 | .354 | .310 | .309 | .324 |
| 49 | .609 | .618 | .617 | .603 | .606 |
| 50 | .654 | .623 | .665 | .638 | .649 |
| 51 | .474 | .400 | .499 | .464 | .432 |
| 52 | .230 | .120 | .262 | .213 | .188 |
| 53 | -.072 | -.202 | -.044 | -.092 | -.126 |
| 54 | -.460 | -.617 | -.447 | -.488 | -.513 |
| 55 | -.709 | -.876 | -.744 | -.751 | -.750 |
| 56 | -.992 | -1.179 | -1.068 | -1.020 | -1.007 |
| 57 | -.941 | -1.123 | -1.098 | -.997 | -.920 |
| 58 | -.938 | -1.139 | -1.135 | -.974 | -.948 |
| 59 | -.958 | -1.143 | -1.181 | -1.012 | -.965 |
| 60 | -.931 | -1.141 | -1.149 | -.972 | -.947 |
| 61 | -.912 | -1.134 | -1.147 | -.940 | -.921 |
| 62 | -.945 | -1.154 | -1.165 | -.995 | -.955 |
| 63 | -1.026 | -1.231 | -1.245 | -1.023 | -1.008 |
| 64 | -1.017 | -1.230 | -1.463 | -1.103 | -1.002 |
| 65 | -1.029 | -1.188 | -1.535 | -1.102 | -.978 |
| 66 | -.825 | -.884 | -1.059 | -.820 | -.765 |
| 67 | -.724 | -.791 | -.743 | -.703 | -.748 |
| 68 | -1.292 | -1.127 | -.981 | -1.273 | -1.619 |
| 69 | -1.415 | -1.150 | -1.219 | -1.425 | -1.669 |
| 70 | -1.715 | -1.388 | -1.468 | -1.697 | -1.812 |
| 71 | -1.476 | -1.246 | -1.437 | -1.461 | -1.469 |
| 72 | -1.666 | -1.389 | -1.484 | -1.599 | -1.676 |
| 73 | -1.441 | -1.249 | -1.368 | -1.403 | -1.433 |
| 74 | -1.563 | -1.326 | -1.400 | -1.482 | -1.555 |
| 75 | -1.444 | -1.245 | -1.339 | -1.398 | -1.429 |
| 76 | -1.651 | -1.403 | -1.426 | -1.566 | -1.670 |
| 77 | -1.586 | -1.375 | -1.484 | -1.540 | -1.568 |
| 78 | -1.615 | -1.374 | -1.480 | -1.532 | -1.585 |
| 79 | -1.301 | -1.112 | -1.268 | -1.254 | -1.247 |
| 80 | -1.010 | -.793 | -.969 | -.949 | -.939 |
| 81 | -.577 | -.406 | -.585 | -.551 | -.504 |
| 82 | -.164 | .001 | -.181 | -.127 | -.096 |
| 83 | .176 | .282 | .159 | .187 | .214 |
| 84 | .533 | .590 | .516 | .530 | .566 |

Figure 33 (continued) – Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .791 | .783 | .796 | .789 | .804 |
| 2 | .619 | .588 | .651 | .621 | .584 |
| 3 | .221 | .195 | .309 | .272 | .196 |
| 4 | -.349 | -.398 | -.273 | -.294 | -.381 |
| 5 | -.920 | -.991 | -.856 | -.861 | -.959 |
| 6 | -.852 | -.941 | -.860 | -.805 | -.830 |
| 7 | -1.141 | -1.119 | -1.183 | -1.039 | -1.085 |
| 8 | -.987 | -.944 | -.999 | -.912 | -.922 |
| 9 | -1.568 | -1.383 | -1.234 | -1.040 | -1.163 |
| 10 | -1.105 | -1.047 | -1.032 | -.871 | -.913 |
| 11 | -1.096 | -.997 | -.932 | -.833 | -.899 |
| 12 | -.805 | -.821 | -.769 | -.696 | -.710 |
| 13 | -1.016 | -1.120 | -.901 | -.991 | -1.203 |
| 14 | -1.747 | -1.443 | -1.541 | -1.533 | -1.509 |
| 15 | -1.475 | -1.472 | -1.624 | -1.601 | -1.572 |
| 16 | -1.507 | -1.498 | -1.577 | -1.652 | -1.589 |
| 17 | -1.515 | -1.470 | -1.589 | -1.634 | -1.600 |
| 18 | -1.456 | -1.450 | -1.450 | -1.599 | -1.540 |
| 19 | -1.640 | -1.599 | -1.672 | -1.764 | -1.740 |
| 20 | -1.360 | -1.328 | -1.409 | -1.464 | -1.386 |
| 21 | -1.033 | -.987 | -1.126 | -1.140 | -1.075 |
| 22 | -.436 | -.415 | -.535 | -.531 | -.439 |
| 23 | .179 | .202 | .093 | .126 | .186 |
| 24 | .554 | .559 | .499 | .519 | .547 |
| 25 | .657 | .648 | .640 | .646 | .668 |
| 26 | .516 | .496 | .551 | .515 | .487 |
| 27 | .166 | .124 | .229 | .033 | .131 |
| 28 | -.346 | -.386 | -.267 | -.557 | -.374 |
| 29 | -1.025 | -1.056 | -.925 | -.947 | -1.053 |
| 30 | -1.395 | -1.418 | -1.312 | -1.289 | -1.340 |
| 31 | -1.633 | -1.634 | -1.617 | -1.466 | -1.551 |
| 32 | -1.493 | -1.516 | -1.480 | -1.382 | -1.405 |
| 33 | -1.681 | -1.682 | -1.670 | -1.525 | -1.646 |
| 34 | -1.117 | -1.114 | -1.653 | -1.100 | -1.038 |
| 35 | -.803 | -.806 | -.816 | -.742 | -.799 |
| 36 | -.642 | -.655 | -.923 | -.664 | -.639 |
| 37 | -.716 | -.880 | -.804 | -.917 | -.917 |
| 38 | -1.306 | -1.397 | -1.168 | -1.187 | -1.105 |
| 39 | -1.335 | -1.373 | -1.272 | -1.392 | -1.422 |
| 40 | -1.350 | -1.348 | -1.304 | -1.418 | -1.342 |
| 41 | -1.302 | -1.271 | -1.293 | -1.395 | -1.388 |
| 42 | -1.187 | -1.169 | -1.212 | -1.343 | -1.268 |

Figure 34 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.355 | -1.362 | -1.360 | -1.528 | -1.500 |
| 44 | -1.217 | -1.189 | -1.259 | -1.345 | -1.252 |
| 45 | -.994 | -.942 | -1.072 | -1.103 | -1.058 |
| 46 | -.536 | -.473 | -.608 | -.592 | -.537 |
| 47 | .040 | .073 | -.048 | -.015 | .080 |
| 48 | .400 | .420 | .350 | .366 | .427 |
| 49 | .603 | .607 | .604 | .606 | .590 |
| 50 | .560 | .528 | .594 | .556 | .544 |
| 51 | .316 | .290 | .370 | .309 | .259 |
| 52 | -.018 | -.095 | .056 | -.012 | -.080 |
| 53 | -.379 | -.452 | -.298 | -.356 | -.436 |
| 54 | -.898 | -.956 | -.796 | -.854 | -.963 |
| 55 | -1.223 | -1.272 | -1.124 | -1.155 | -1.227 |
| 56 | -1.654 | -1.659 | -1.608 | -1.541 | -1.660 |
| 57 | -1.736 | -1.748 | -1.712 | -1.588 | -1.621 |
| 58 | -1.705 | -1.700 | -1.694 | -1.492 | -1.596 |
| 59 | -1.614 | -1.641 | -1.602 | -1.476 | -1.513 |
| 60 | -1.708 | -1.709 | -1.714 | -1.547 | -1.662 |
| 61 | -1.653 | -1.673 | -1.639 | -1.532 | -1.574 |
| 62 | -1.932 | -1.784 | -1.851 | -1.832 | -1.749 |
| 63 | -1.667 | -1.706 | -1.674 | -1.563 | -1.623 |
| 64 | -.893 | -.812 | -1.288 | -.890 | -.839 |
| 65 | -.814 | -.838 | -.786 | -.802 | -.838 |
| 66 | -.717 | -.740 | -.853 | -.792 | -.838 |
| 67 | -.837 | -.913 | -.885 | -1.013 | -.977 |
| 68 | -1.064 | -1.109 | -1.128 | -1.198 | -1.131 |
| 69 | -.952 | -.990 | -1.004 | -1.097 | -1.002 |
| 70 | -.985 | -.990 | -1.009 | -1.118 | -1.045 |
| 71 | -.849 | -.835 | -.870 | -.988 | -.916 |
| 72 | -.955 | -.929 | -.967 | -1.092 | -1.050 |
| 73 | -.885 | -.852 | -.886 | -1.006 | -.944 |
| 74 | -.938 | -.905 | -.943 | -1.085 | -1.039 |
| 75 | -.890 | -.864 | -.888 | -1.028 | -.953 |
| 76 | -.971 | -.949 | -.959 | -1.158 | -1.117 |
| 77 | -1.014 | -.987 | -1.012 | -1.159 | -1.080 |
| 78 | -1.006 | -.959 | -1.035 | -1.149 | -1.087 |
| 79 | -.816 | -.764 | -.860 | -.915 | -.826 |
| 80 | -.551 | -.498 | -.624 | -.613 | -.563 |
| 81 | -.234 | -.180 | -.310 | -.265 | -.218 |
| 82 | .123 | .157 | .047 | .104 | .150 |
| 83 | .356 | .387 | .298 | .350 | .362 |
| 84 | .611 | .615 | .591 | .604 | .635 |

Figure 34 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .793 | .809 | .798 | .797 | .799 |
| 2 | .626 | .586 | .648 | .643 | .596 |
| 3 | .243 | .216 | .315 | .333 | .190 |
| 4 | -.296 | -.320 | -.234 | -.176 | -.370 |
| 5 | -.834 | -.855 | -.783 | -.685 | -.930 |
| 6 | -.712 | -.685 | -.711 | -.575 | -.803 |
| 7 | -1.062 | -.994 | -1.006 | -.804 | -1.018 |
| 8 | -1.620 | -1.341 | -1.353 | -.666 | -1.280 |
| 9 | -1.740 | -1.563 | -1.955 | -1.087 | -1.805 |
| 10 | -1.432 | -1.319 | -1.433 | -.757 | -1.364 |
| 11 | -1.125 | -1.087 | -1.303 | -.895 | -1.200 |
| 12 | -.870 | -.864 | -.949 | -.702 | -.906 |
| 13 | -1.005 | -1.205 | -.985 | -1.009 | -1.237 |
| 14 | -1.511 | -1.463 | -1.467 | -1.439 | -1.524 |
| 15 | -1.502 | -1.543 | -1.509 | -1.538 | -1.549 |
| 16 | -1.533 | -1.499 | -1.491 | -1.709 | -1.547 |
| 17 | -1.498 | -1.533 | -1.500 | -1.750 | -1.528 |
| 18 | -1.431 | -1.438 | -1.378 | -1.753 | -1.478 |
| 19 | -1.621 | -1.665 | -1.614 | -1.936 | -1.650 |
| 20 | -1.345 | -1.317 | -1.352 | -1.577 | -1.345 |
| 21 | -1.046 | -1.040 | -1.108 | -1.256 | -1.017 |
| 22 | -.456 | -.419 | -.528 | -.583 | -.411 |
| 23 | .160 | .187 | .088 | .067 | .212 |
| 24 | .541 | .537 | .481 | .482 | .570 |
| 25 | .656 | .657 | .652 | .649 | .664 |
| 26 | .518 | .489 | .543 | .528 | .498 |
| 27 | .173 | .138 | .240 | .214 | .102 |
| 28 | -.332 | -.341 | -.251 | -.268 | -.411 |
| 29 | -.997 | -1.038 | -.928 | -.873 | -1.058 |
| 30 | -1.352 | -1.315 | -1.300 | -1.170 | -1.376 |
| 31 | -1.563 | -1.524 | -1.591 | -1.320 | -1.545 |
| 32 | -1.377 | -1.324 | -1.406 | -1.176 | -1.388 |
| 33 | -1.642 | -1.637 | -1.661 | -1.456 | -1.655 |
| 34 | -.866 | -.789 | -.932 | -.716 | -.848 |
| 35 | -.872 | -.864 | -.939 | -.764 | -.920 |
| 36 | -.582 | -.563 | -.630 | -.550 | -.575 |
| 37 | -.833 | -.970 | -.884 | -1.165 | -.971 |
| 38 | -1.388 | -1.358 | -1.235 | -1.309 | -1.246 |
| 39 | -1.343 | -1.425 | -1.286 | -1.540 | -1.395 |
| 40 | -1.357 | -1.351 | -1.298 | -1.472 | -1.326 |
| 41 | -1.299 | -1.339 | -1.274 | -1.524 | -1.347 |
| 42 | -1.183 | -1.181 | -1.173 | -1.399 | -1.213 |

Figure 35 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.346 | -1.409 | -1.321 | -1.678 | -1.416 |
| 44 | -1.212 | -1.192 | -1.233 | -1.418 | -1.221 |
| 45 | -1.000 | -.994 | -1.048 | -1.208 | -.876 |
| 46 | -.545 | -.489 | -.603 | -.643 | -.274 |
| 47 | .041 | .051 | -.042 | -.044 | .092 |
| 48 | .401 | .388 | .345 | .356 | .451 |
| 49 | .603 | .586 | .596 | .596 | .604 |
| 50 | .565 | .514 | .599 | .574 | .526 |
| 51 | .321 | .191 | .365 | .325 | .262 |
| 52 | -.009 | -.066 | .053 | .008 | -.102 |
| 53 | -.363 | -.419 | -.301 | -.342 | -.454 |
| 54 | -1.012 | -.929 | -.813 | -.844 | -.984 |
| 55 | -1.339 | -1.194 | -1.136 | -1.133 | -1.278 |
| 56 | -1.616 | -1.664 | -1.607 | -1.534 | -1.706 |
| 57 | -1.710 | -1.668 | -1.716 | -1.568 | -1.743 |
| 58 | -1.704 | -1.695 | -1.775 | -1.534 | -1.709 |
| 59 | -1.591 | -1.536 | -1.622 | -1.439 | -1.614 |
| 60 | -1.700 | -1.716 | -1.763 | -1.579 | -1.724 |
| 61 | -1.625 | -1.604 | -1.643 | -1.515 | -1.669 |
| 62 | -1.961 | -1.774 | -1.903 | -1.843 | -1.805 |
| 63 | -1.601 | -1.587 | -1.615 | -1.518 | -1.674 |
| 64 | -.796 | -.773 | -.814 | -.833 | -.806 |
| 65 | -.789 | -.811 | -.800 | -.869 | -.879 |
| 66 | -.706 | -.788 | -.729 | -.890 | -.781 |
| 67 | -.895 | -.942 | -.875 | -1.100 | -.964 |
| 68 | -1.204 | -1.188 | -1.205 | -1.249 | -1.154 |
| 69 | -1.107 | -1.018 | -1.091 | -1.125 | -1.021 |
| 70 | -1.108 | -1.088 | -1.090 | -1.128 | -1.070 |
| 71 | -.945 | -.872 | -.938 | -1.036 | -.912 |
| 72 | -.954 | -1.011 | -.933 | -1.166 | -1.004 |
| 73 | -.895 | -.886 | -.888 | -1.058 | -.929 |
| 74 | -.946 | -.987 | -.938 | -1.136 | -.987 |
| 75 | -.893 | -.902 | -.884 | -1.049 | -.932 |
| 76 | -.974 | -1.040 | -.950 | -1.218 | -1.047 |
| 77 | -1.012 | -1.017 | -.995 | -1.185 | -1.054 |
| 78 | -1.018 | -1.042 | -1.042 | -1.200 | -1.028 |
| 79 | -.837 | -.803 | -.862 | -.935 | -.806 |
| 80 | -.559 | -.544 | -.631 | -.663 | -.527 |
| 81 | -.235 | -.215 | -.311 | -.296 | -.200 |
| 82 | .111 | .140 | .057 | .077 | .165 |
| 83 | .346 | .354 | .301 | .327 | .387 |
| 84 | .613 | .620 | .591 | .603 | .628 |

Figure 35 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .759 | .784 | .790 | .780 | .810 |
| 2 | .701 | .665 | .668 | .716 | .678 |
| 3 | .457 | .414 | .328 | .423 | .343 |
| 4 | -.075 | -.141 | -.234 | -.124 | -.229 |
| 5 | -.714 | -.862 | -.886 | -.760 | -.844 |
| 6 | -1.039 | -1.217 | -1.148 | -1.080 | -1.123 |
| 7 | -1.363 | -1.573 | -1.409 | -1.401 | -1.402 |
| 8 | -.995 | -1.113 | -1.093 | -1.033 | -1.094 |
| 9 | -1.056 | -1.158 | -1.084 | -1.066 | -1.063 |
| 10 | -.947 | -1.036 | -1.207 | -.989 | -1.162 |
| 11 | -1.777 | -1.949 | -1.795 | -1.787 | -1.843 |
| 12 | -1.115 | -1.033 | -.974 | -1.079 | -.968 |
| 13 | -.854 | -.956 | -.908 | -.859 | -.908 |
| 14 | -1.459 | -1.415 | -1.596 | -1.456 | -1.601 |
| 15 | -1.636 | -1.537 | -1.609 | -1.569 | -1.619 |
| 16 | -1.541 | -1.441 | -1.614 | -1.519 | -1.582 |
| 17 | -1.697 | -1.558 | -1.623 | -1.626 | -1.615 |
| 18 | -1.516 | -1.363 | -1.495 | -1.487 | -1.463 |
| 19 | -1.804 | -1.651 | -1.730 | -1.729 | -1.714 |
| 20 | -1.534 | -1.376 | -1.425 | -1.499 | -1.414 |
| 21 | -1.288 | -1.166 | -1.104 | -1.227 | -1.118 |
| 22 | -.712 | -.610 | -.521 | -.674 | -.552 |
| 23 | -.085 | -.002 | .092 | -.031 | .342 |
| 24 | .388 | .417 | .506 | .436 | .713 |
| 25 | .604 | .635 | .658 | .621 | .643 |
| 26 | .604 | .571 | .582 | .609 | .574 |
| 27 | .421 | .339 | .309 | .394 | .328 |
| 28 | -.022 | -.159 | -.176 | -.074 | -.152 |
| 29 | -.598 | -.818 | -.787 | -.646 | -.773 |
| 30 | -.926 | -1.156 | -1.084 | -1.013 | -1.072 |
| 31 | -1.131 | -1.453 | -1.230 | -1.226 | -1.244 |
| 32 | -1.119 | -1.362 | -1.229 | -1.218 | -1.228 |
| 33 | -1.286 | -1.553 | -1.359 | -1.346 | -1.355 |
| 34 | -1.125 | -1.339 | -1.188 | -1.207 | -1.180 |
| 35 | -1.234 | -1.500 | -1.344 | -1.316 | -1.330 |
| 36 | -1.143 | -1.290 | -1.104 | -1.222 | -1.112 |
| 37 | -.810 | -.869 | -.819 | -.798 | -.814 |
| 38 | -1.248 | -1.209 | -1.361 | -1.330 | -1.313 |
| 39 | -1.338 | -1.241 | -1.291 | -1.281 | -1.246 |
| 40 | -1.396 | -1.147 | -1.443 | -1.403 | -1.394 |
| 41 | -1.563 | -1.332 | -1.463 | -1.474 | -1.430 |
| 42 | -1.450 | -1.145 | -1.394 | -1.406 | -1.357 |

Figure 36 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.673 | -1.328 | -1.604 | -1.560 | -1.563 |
| 44 | -1.564 | -1.271 | -1.457 | -1.518 | -1.426 |
| 45 | -1.402 | -1.122 | -1.219 | -1.299 | -1.209 |
| 46 | -.896 | -.674 | -.700 | -.845 | -.702 |
| 47 | -.292 | -.160 | -.113 | -.249 | -.148 |
| 48 | .201 | .263 | .330 | .218 | .293 |
| 49 | .614 | .654 | .652 | .628 | .648 |
| 50 | .602 | .568 | .602 | .616 | .586 |
| 51 | .569 | .484 | .481 | .547 | .483 |
| 52 | .326 | .190 | .215 | .304 | .217 |
| 53 | .029 | -.200 | -.153 | -.038 | -.146 |
| 54 | -.275 | -.545 | -.466 | -.364 | -.459 |
| 55 | -.674 | -1.035 | -.867 | -.767 | -.856 |
| 56 | -.842 | -1.208 | -1.006 | -.959 | -.996 |
| 57 | -1.001 | -1.532 | -1.116 | -1.131 | -1.144 |
| 58 | -.881 | -1.343 | -1.013 | -1.008 | -1.018 |
| 59 | -.928 | -1.441 | -1.057 | -1.048 | -1.077 |
| 60 | -.853 | -1.296 | -.981 | -.975 | -.993 |
| 61 | -.981 | -1.464 | -1.106 | -1.087 | -1.120 |
| 62 | -.918 | -1.358 | -1.054 | -1.053 | -1.068 |
| 63 | -1.032 | -1.589 | -1.312 | -1.255 | -1.339 |
| 64 | -.972 | -1.562 | -1.209 | -1.201 | -1.236 |
| 65 | -1.109 | -1.846 | -1.403 | -1.383 | -1.383 |
| 66 | -1.081 | -1.615 | -1.279 | -1.259 | -1.255 |
| 67 | -.643 | -.826 | -.657 | -.648 | -.638 |
| 68 | -1.032 | -.944 | -1.233 | -.976 | -1.174 |
| 69 | -1.702 | -1.122 | -1.527 | -1.581 | -1.509 |
| 70 | -1.583 | -1.006 | -1.564 | -1.465 | -1.515 |
| 71 | -1.734 | -1.181 | -1.515 | -1.718 | -1.504 |
| 72 | -1.601 | -1.028 | -1.461 | -1.461 | -1.411 |
| 73 | -1.619 | -1.101 | -1.411 | -1.490 | -1.391 |
| 74 | -1.482 | -.976 | -1.325 | -1.348 | -1.292 |
| 75 | -1.573 | -1.055 | -1.384 | -1.403 | -1.351 |
| 76 | -1.519 | -.977 | -1.377 | -1.366 | -1.326 |
| 77 | -1.752 | -1.176 | -1.562 | -1.564 | -1.531 |
| 78 | -1.579 | -1.087 | -1.397 | -1.455 | -1.367 |
| 79 | -1.502 | -1.053 | -1.257 | -1.347 | -1.242 |
| 80 | -1.107 | -.762 | -.883 | -1.007 | -.877 |
| 81 | -.740 | -.444 | -.513 | -.653 | -.515 |
| 82 | -.307 | -.099 | -.119 | -.252 | -.136 |
| 83 | .106 | .277 | .255 | .153 | .248 |
| 84 | .358 | .447 | .469 | .397 | .447 |

Figure 36 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .789 | .770 | .799 | .772 | .796 |
| 2 | .622 | .659 | .575 | .691 | .634 |
| 3 | .303 | .388 | .191 | .356 | .268 |
| 4 | -.292 | -.174 | -.413 | -.203 | -.313 |
| 5 | -.989 | -.839 | -1.108 | -.819 | -.963 |
| 6 | -1.373 | -1.215 | -1.435 | -1.149 | -1.260 |
| 7 | -1.756 | -1.591 | -1.761 | -1.480 | -1.558 |
| 8 | -1.286 | -1.144 | -1.297 | -1.128 | -1.252 |
| 9 | -1.190 | -1.019 | -1.221 | -1.052 | -.991 |
| 10 | -1.123 | -.881 | -1.004 | -.817 | -.788 |
| 11 | -1.287 | -1.056 | -1.295 | -1.024 | -.913 |
| 12 | -.925 | -.833 | -.832 | -.740 | -.719 |
| 13 | -1.079 | -1.034 | -1.165 | -.788 | -1.039 |
| 14 | -1.327 | -1.315 | -1.305 | -1.611 | -1.532 |
| 15 | -1.411 | -1.449 | -1.400 | -1.650 | -1.480 |
| 16 | -1.355 | -1.412 | -1.369 | -1.671 | -1.584 |
| 17 | -1.406 | -1.463 | -1.420 | -1.644 | -1.516 |
| 18 | -1.266 | -1.337 | -1.311 | -1.533 | -1.484 |
| 19 | -1.532 | -1.580 | -1.582 | -1.739 | -1.677 |
| 20 | -1.264 | -1.352 | -1.224 | -1.495 | -1.391 |
| 21 | -1.025 | -1.120 | -.951 | -1.175 | -1.050 |
| 22 | -.486 | -.590 | -.373 | -.622 | -.484 |
| 23 | .101 | .020 | .218 | .016 | .143 |
| 24 | .470 | .431 | .550 | .465 | .534 |
| 25 | .659 | .630 | .671 | .649 | .664 |
| 26 | .532 | .554 | .496 | .592 | .530 |
| 27 | .231 | .306 | .137 | .294 | .191 |
| 28 | -.302 | -.209 | -.403 | -.213 | -.356 |
| 29 | -1.009 | -.900 | -1.111 | -.861 | -1.009 |
| 30 | -1.375 | -1.310 | -1.418 | -1.264 | -1.371 |
| 31 | -1.843 | -1.720 | -1.781 | -1.495 | -1.588 |
| 32 | -1.657 | -1.591 | -1.630 | -1.463 | -1.538 |
| 33 | -1.828 | -1.708 | -1.807 | -1.533 | -1.623 |
| 34 | -1.761 | -1.710 | -1.646 | -1.545 | -1.625 |
| 35 | -1.011 | -.847 | -1.011 | -.862 | -.865 |
| 36 | -1.017 | -.943 | -.994 | -.850 | -.915 |
| 37 | -.705 | -.650 | -.769 | -.690 | -.738 |
| 38 | -1.175 | -1.254 | -1.327 | -1.208 | -1.133 |
| 39 | -1.263 | -1.323 | -1.247 | -1.253 | -1.226 |
| 40 | -1.117 | -1.215 | -1.198 | -1.433 | -1.333 |
| 41 | -1.209 | -1.284 | -1.234 | -1.387 | -1.310 |
| 42 | -1.013 | -1.088 | -1.048 | -1.355 | -1.257 |

Figure 37 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.174 | -1.241 | -1.260 | -1.486 | -1.415 |
| 44 | -1.112 | -1.211 | -1.104 | -1.415 | -1.294 |
| 45 | -.954 | -1.045 | -.927 | -1.170 | -1.037 |
| 46 | -.524 | -.628 | -.464 | -.722 | -.574 |
| 47 | -.033 | -.108 | .075 | .117 | -.003 |
| 48 | .337 | .297 | .411 | .534 | .378 |
| 49 | .659 | .644 | .667 | .644 | .654 |
| 50 | .533 | .561 | .495 | .583 | .549 |
| 51 | .379 | .425 | .310 | .429 | .347 |
| 52 | .055 | .125 | -.033 | .150 | .033 |
| 53 | -.381 | -.294 | -.503 | -.272 | -.409 |
| 54 | -.730 | -.659 | -.847 | -.641 | -.775 |
| 55 | -1.315 | -1.192 | -1.423 | -1.110 | -1.281 |
| 56 | -1.504 | -1.425 | -1.564 | -1.345 | -1.485 |
| 57 | -1.983 | -1.850 | -1.948 | -1.616 | -1.776 |
| 58 | -1.830 | -1.768 | -1.719 | -1.502 | -1.621 |
| 59 | -1.936 | -1.813 | -1.846 | -1.544 | -1.656 |
| 60 | -1.746 | -1.678 | -1.657 | -1.475 | -1.559 |
| 61 | -1.913 | -1.824 | -1.870 | -1.567 | -1.696 |
| 62 | -1.771 | -1.714 | -1.765 | -1.582 | -1.742 |
| 63 | -2.249 | -2.179 | -2.252 | -1.906 | -2.078 |
| 64 | -1.798 | -1.719 | -1.468 | -1.448 | -1.331 |
| 65 | -1.109 | -.989 | -1.051 | -.993 | -1.903 |
| 66 | -1.715 | -1.657 | -1.300 | -1.344 | -1.144 |
| 67 | -.875 | -.833 | -.928 | -.848 | -1.202 |
| 68 | -1.139 | -1.171 | -1.101 | -1.286 | -1.169 |
| 69 | -1.065 | -1.053 | -1.150 | -1.301 | -1.111 |
| 70 | -.884 | -.927 | -.956 | -1.250 | -1.051 |
| 71 | -.915 | -.935 | -.959 | -1.158 | -.994 |
| 72 | -.768 | -.799 | -.822 | -1.091 | -.960 |
| 73 | -.839 | -.897 | -.870 | -1.103 | -.989 |
| 74 | -.744 | -.818 | -.774 | -1.059 | -.941 |
| 75 | -.816 | -.879 | -.848 | -1.091 | -.989 |
| 76 | -.757 | -.823 | -.790 | -1.088 | -.987 |
| 77 | -.918 | -.976 | -.978 | -1.232 | -1.136 |
| 78 | -.855 | -.939 | -.854 | -1.159 | -1.035 |
| 79 | -.812 | -.892 | -.781 | -1.035 | -.895 |
| 80 | -.550 | -.637 | -.493 | -.753 | -.609 |
| 81 | -.270 | -.351 | -.182 | -.392 | -.265 |
| 82 | .034 | -.035 | .102 | -.048 | .055 |
| 83 | .372 | .306 | .441 | .297 | .386 |
| 84 | .500 | .470 | .524 | .487 | .536 |

Figure 37 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .787 | .772 | .778 | .752 | .789 |
| 2 | .637 | .662 | .595 | .676 | .620 |
| 3 | .311 | .408 | .193 | .402 | .269 |
| 4 | -.285 | -.138 | -.403 | -.115 | -.320 |
| 5 | -.939 | -.797 | -1.092 | -.694 | -.955 |
| 6 | -1.270 | -1.112 | -1.383 | -.962 | -1.217 |
| 7 | -1.602 | -1.427 | -1.674 | -1.231 | -1.480 |
| 8 | -1.275 | -1.041 | -1.375 | -.935 | -1.295 |
| 9 | -1.623 | -1.407 | -1.549 | -.777 | -1.329 |
| 10 | -1.293 | -1.009 | -1.387 | -.766 | -1.137 |
| 11 | -1.388 | -1.257 | -1.404 | -.813 | -1.204 |
| 12 | -.971 | -.856 | -1.056 | -.714 | -.878 |
| 13 | -1.074 | -1.067 | -1.231 | -.764 | -1.167 |
| 14 | -1.437 | -1.429 | -1.364 | -1.664 | -1.508 |
| 15 | -1.443 | -1.527 | -1.369 | -1.702 | -1.483 |
| 16 | -1.430 | -1.473 | -1.394 | -1.762 | -1.531 |
| 17 | -1.428 | -1.517 | -1.353 | -1.756 | -1.471 |
| 18 | -1.303 | -1.345 | -1.320 | -1.647 | -1.407 |
| 19 | -1.550 | -1.632 | -1.503 | -1.888 | -1.625 |
| 20 | -1.297 | -1.363 | -1.243 | -1.582 | -1.330 |
| 21 | -1.041 | -1.161 | -.914 | -1.282 | -1.014 |
| 22 | -.512 | -.605 | -.393 | -.687 | -.452 |
| 23 | .106 | -.003 | .217 | -.036 | .153 |
| 24 | .485 | .420 | .563 | .416 | .529 |
| 25 | .648 | .640 | .659 | .611 | .642 |
| 26 | .540 | .551 | .502 | .560 | .496 |
| 27 | .247 | .307 | .129 | .319 | .193 |
| 28 | -.285 | -.216 | -.430 | -.151 | -.342 |
| 29 | -.961 | -.854 | -1.098 | -.779 | -1.009 |
| 30 | -1.350 | -1.231 | -1.477 | -1.120 | -1.355 |
| 31 | -1.753 | -1.646 | -1.756 | -1.359 | -1.618 |
| 32 | -1.619 | -1.495 | -1.684 | -1.298 | -1.542 |
| 33 | -1.721 | -1.606 | -1.763 | -1.357 | -1.610 |
| 34 | -1.281 | -1.284 | -1.212 | -1.159 | -1.069 |
| 35 | -1.069 | -.880 | -1.224 | -.759 | -.991 |
| 36 | -.834 | -.782 | -.891 | -.722 | -.680 |
| 37 | -.794 | -.693 | -.894 | -.808 | -.784 |
| 38 | -1.256 | -1.256 | -1.390 | -1.275 | -1.522 |
| 39 | -1.247 | -1.325 | -1.194 | -1.397 | -1.305 |
| 40 | -1.195 | -1.244 | -1.247 | -1.495 | -1.315 |
| 41 | -1.229 | -1.324 | -1.161 | -1.482 | -1.294 |
| 42 | -1.067 | -1.123 | -1.068 | -1.409 | -1.169 |

Figure 38 - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.221 | -1.291 | -1.201 | -1.585 | -1.372 |
| 44 | -1.168 | -1.237 | -1.126 | -1.465 | -1.260 |
| 45 | -.987 | -1.081 | -.878 | -1.238 | -.996 |
| 46 | -.563 | -.646 | -.464 | -.752 | -.540 |
| 47 | -.038 | -.120 | .086 | -.155 | .243 |
| 48 | .346 | .279 | .429 | .272 | .580 |
| 49 | .650 | .647 | .641 | .631 | .653 |
| 50 | .546 | .555 | .516 | .562 | .534 |
| 51 | .396 | .440 | .283 | .431 | .330 |
| 52 | .080 | .139 | -.044 | .153 | .009 |
| 53 | -.356 | -.274 | -.510 | -.258 | -.430 |
| 54 | -.724 | -.625 | -.882 | -.603 | -.791 |
| 55 | -1.284 | -1.172 | -1.418 | -1.063 | -1.320 |
| 56 | -1.496 | -1.371 | -1.634 | -1.255 | -1.508 |
| 57 | -1.913 | -1.801 | -1.953 | -1.564 | -1.808 |
| 58 | -1.804 | -1.691 | -1.834 | -1.426 | -1.661 |
| 59 | -1.868 | -1.775 | -1.842 | -1.481 | -1.714 |
| 60 | -1.713 | -1.602 | -1.741 | -1.374 | -1.586 |
| 61 | -1.847 | -1.758 | -1.850 | -1.504 | -1.723 |
| 62 | -1.780 | -1.659 | -1.893 | -1.530 | -1.767 |
| 63 | -2.112 | -2.120 | -1.882 | -1.789 | -1.684 |
| 64 | -1.170 | -1.071 | -1.090 | -.981 | -.942 |
| 65 | -1.022 | -1.876 | -1.017 | -.848 | -1.532 |
| 66 | -1.248 | -.946 | -.888 | -.805 | -.757 |
| 67 | -.829 | -1.089 | -.937 | -.965 | -1.059 |
| 68 | -1.049 | -.998 | -1.036 | -1.220 | -1.044 |
| 69 | -1.150 | -1.098 | -1.177 | -1.254 | -1.115 |
| 70 | -.974 | -.941 | -.993 | -1.195 | -1.022 |
| 71 | -1.017 | -1.002 | -.944 | -1.114 | -1.000 |
| 72 | -.828 | -.870 | -.812 | -1.133 | -.943 |
| 73 | -.896 | -.948 | -.836 | -1.142 | -.956 |
| 74 | -.796 | -.862 | -.782 | -1.102 | -.912 |
| 75 | -.877 | -.942 | -.823 | -1.161 | -.963 |
| 76 | -.813 | -.870 | -.814 | -1.141 | -.955 |
| 77 | -.988 | -1.061 | -.951 | -1.304 | -1.097 |
| 78 | -.913 | -.985 | -.870 | -1.191 | -.991 |
| 79 | -.856 | -.949 | -.753 | -1.086 | -.880 |
| 80 | -.586 | -.672 | -.489 | -.778 | -.587 |
| 81 | -.298 | -.374 | -.174 | -.415 | -.252 |
| 82 | .016 | -.047 | .125 | -.069 | .063 |
| 83 | .350 | .294 | .425 | .282 | .394 |
| 84 | .499 | .456 | .552 | .461 | .529 |

Figure 38 (continued) - Pressure Coefficients
 $\alpha = 50^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .819 | .860 | .822 | .849 | .855 |
| 2 | .715 | .668 | .722 | .698 | .674 |
| 3 | .292 | .204 | .349 | .287 | .268 |
| 4 | -.386 | -.530 | -.357 | -.386 | -.425 |
| 5 | -1.063 | -1.264 | -1.064 | -1.059 | -1.117 |
| 6 | -.938 | -1.118 | -1.050 | -1.136 | -1.105 |
| 7 | -1.170 | -1.314 | -1.263 | -1.146 | -1.115 |
| 8 | -1.091 | -1.211 | -1.129 | -1.054 | -1.011 |
| 9 | -1.591 | -1.801 | -1.755 | -1.559 | -1.546 |
| 10 | -1.805 | -1.987 | -1.709 | -1.769 | -1.680 |
| 11 | -2.229 | -2.352 | -2.009 | -2.167 | -2.036 |
| 12 | -1.378 | -1.240 | -1.394 | -1.379 | -1.276 |
| 13 | -1.115 | -1.242 | -.987 | -1.105 | -1.266 |
| 14 | -2.293 | -2.096 | -1.928 | -2.268 | -2.230 |
| 15 | -2.129 | -2.101 | -2.105 | -2.148 | -2.090 |
| 16 | -2.280 | -2.066 | -1.969 | -2.290 | -2.318 |
| 17 | -2.257 | -2.069 | -2.143 | -2.279 | -2.314 |
| 18 | -2.100 | -1.861 | -1.869 | -2.098 | -2.183 |
| 19 | -2.415 | -2.166 | -2.194 | -2.440 | -2.521 |
| 20 | -2.030 | -1.821 | -1.893 | -2.021 | -2.048 |
| 21 | -1.709 | -1.490 | -1.650 | -1.721 | -1.734 |
| 22 | -.924 | -.741 | -.904 | -.901 | -.894 |
| 23 | -.127 | .024 | -.161 | -.108 | -.064 |
| 24 | .440 | .525 | .404 | .451 | .470 |
| 25 | .740 | .765 | .710 | .735 | .754 |
| 26 | .684 | .655 | .687 | .676 | .666 |
| 27 | .401 | .282 | .421 | .395 | .375 |
| 28 | -.094 | -.275 | -.098 | -.097 | -.119 |
| 29 | -.690 | -.967 | -.768 | -.698 | -.690 |
| 30 | -.883 | -1.250 | -1.090 | -.861 | -.791 |
| 31 | -1.122 | -1.450 | -1.310 | -1.112 | -1.069 |
| 32 | -1.149 | -1.450 | -1.321 | -1.147 | -1.071 |
| 33 | -1.029 | -1.415 | -1.292 | -1.024 | -.936 |
| 34 | -1.048 | -1.365 | -1.162 | -1.027 | -.994 |
| 35 | -1.123 | -1.415 | -1.269 | -1.116 | -1.107 |
| 36 | -1.002 | -1.167 | -1.143 | -.974 | -.911 |
| 37 | -1.011 | -.970 | -.865 | -1.116 | -1.432 |
| 38 | -1.915 | -1.711 | -1.767 | -1.943 | -1.865 |
| 39 | -1.952 | -1.706 | -1.774 | -1.995 | -2.149 |
| 40 | -2.145 | -1.815 | -1.868 | -2.085 | -2.122 |
| 41 | -2.079 | -1.834 | -1.980 | -2.119 | -2.183 |
| 42 | -1.930 | -1.719 | -1.786 | -1.952 | -1.977 |

Figure 39 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -2.241 | -1.993 | -2.024 | -2.285 | -2.407 |
| 44 | -1.983 | -1.781 | -1.891 | -1.984 | -2.002 |
| 45 | -1.717 | -1.522 | -1.669 | -1.725 | -1.762 |
| 46 | -1.051 | -.870 | -1.053 | -1.024 | -.969 |
| 47 | -.254 | -.142 | -.321 | -.264 | -.208 |
| 48 | .311 | .380 | .253 | .299 | .346 |
| 49 | .758 | .773 | .729 | .745 | .750 |
| 50 | .815 | .806 | .834 | .815 | .812 |
| 51 | .601 | .559 | .645 | .589 | .556 |
| 52 | .344 | .234 | .401 | .324 | .288 |
| 53 | .022 | -.127 | .065 | -.010 | -.047 |
| 54 | -.384 | -.550 | -.355 | -.382 | -.400 |
| 55 | -.592 | -.808 | -.617 | -.573 | -.573 |
| 56 | -.692 | -1.083 | -.839 | -.672 | -.630 |
| 57 | -.630 | -.982 | -.745 | -.604 | -.574 |
| 58 | -.714 | -.997 | -.860 | -.705 | -.682 |
| 59 | -.739 | -1.031 | -.911 | -.706 | -.683 |
| 60 | -.703 | -1.019 | -.817 | -.684 | -.685 |
| 61 | -.666 | -.976 | -.781 | -.636 | -.652 |
| 62 | -.730 | -1.037 | -.813 | -.731 | -.699 |
| 63 | -.723 | -1.039 | -.871 | -.833 | -.876 |
| 64 | -.791 | -1.020 | -.883 | -.859 | -.923 |
| 65 | -.765 | -.949 | -.826 | -.875 | -.992 |
| 66 | -.990 | -.663 | -.683 | -1.106 | -1.282 |
| 67 | -1.292 | -1.010 | -.847 | -1.247 | -1.359 |
| 68 | -1.576 | -1.818 | -1.748 | -1.423 | -1.533 |
| 69 | -1.443 | -1.770 | -1.655 | -1.375 | -1.449 |
| 70 | -1.615 | -2.153 | -1.902 | -1.600 | -1.620 |
| 71 | -1.562 | -1.749 | -1.672 | -1.536 | -1.520 |
| 72 | -1.722 | -1.947 | -1.720 | -1.646 | -1.646 |
| 73 | -1.655 | -1.678 | -1.578 | -1.561 | -1.521 |
| 74 | -1.775 | -1.634 | -1.698 | -1.786 | -1.817 |
| 75 | -1.677 | -1.531 | -1.569 | -1.675 | -1.672 |
| 76 | -1.957 | -1.730 | -1.757 | -1.989 | -1.913 |
| 77 | -1.863 | -1.736 | -1.732 | -1.881 | -1.787 |
| 78 | -1.839 | -1.709 | -1.834 | -1.897 | -1.899 |
| 79 | -1.490 | -1.382 | -1.539 | -1.495 | -1.460 |
| 80 | -1.130 | -1.005 | -1.198 | -1.111 | -1.069 |
| 81 | -.622 | -.515 | -.700 | -.604 | -.555 |
| 82 | -.116 | -.049 | -.216 | -.092 | -.048 |
| 83 | .269 | .319 | .189 | .282 | .313 |
| 84 | .688 | .723 | .629 | .686 | .724 |

Figure 39 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .829 | .849 | .842 | .852 | .869 |
| 2 | .644 | .581 | .706 | .653 | .624 |
| 3 | .157 | .054 | .262 | .227 | .140 |
| 4 | -.538 | -.681 | -.471 | -.479 | -.576 |
| 5 | -1.233 | -1.416 | -1.204 | -1.184 | -1.292 |
| 6 | -1.099 | -1.257 | -1.174 | -1.108 | -1.174 |
| 7 | -1.335 | -1.517 | -1.501 | -1.318 | -1.303 |
| 8 | -1.188 | -1.328 | -1.294 | -1.160 | -1.153 |
| 9 | -1.360 | -1.384 | -1.608 | -1.161 | -1.145 |
| 10 | -1.100 | -1.216 | -1.020 | -.887 | -.899 |
| 11 | -1.072 | -1.075 | -.988 | -.770 | -.793 |
| 12 | -.872 | -.974 | -.733 | -.718 | -.708 |
| 13 | -1.185 | -1.531 | -1.152 | -1.578 | -1.515 |
| 14 | -1.818 | -1.830 | -2.050 | -2.010 | -1.915 |
| 15 | -1.889 | -1.844 | -2.124 | -2.189 | -2.111 |
| 16 | -1.984 | -1.805 | -2.089 | -2.140 | -2.073 |
| 17 | -1.892 | -1.747 | -2.016 | -2.111 | -2.075 |
| 18 | -1.792 | -1.634 | -1.831 | -1.948 | -1.910 |
| 19 | -2.020 | -1.888 | -2.082 | -2.272 | -2.209 |
| 20 | -1.735 | -1.586 | -1.817 | -1.853 | -1.791 |
| 21 | -1.386 | -1.234 | -1.516 | -1.534 | -1.457 |
| 22 | -.692 | -.553 | -.812 | -.751 | -.676 |
| 23 | .030 | .172 | -.041 | .015 | .099 |
| 24 | .521 | .591 | .494 | .504 | .561 |
| 25 | .759 | .775 | .762 | .780 | .805 |
| 26 | .610 | .572 | .660 | .610 | .589 |
| 27 | .161 | .066 | .260 | .200 | .165 |
| 28 | -.463 | -.578 | -.351 | -.406 | -.444 |
| 29 | -1.264 | -1.441 | -1.198 | -1.235 | -1.273 |
| 30 | -1.716 | -1.881 | -1.688 | -1.577 | -1.589 |
| 31 | -1.992 | -2.197 | -2.035 | -1.802 | -1.812 |
| 32 | -1.855 | -2.043 | -1.897 | -1.684 | -1.713 |
| 33 | -2.078 | -2.290 | -2.125 | -1.943 | -1.966 |
| 34 | -2.136 | -2.458 | -2.283 | -2.026 | -2.010 |
| 35 | -1.016 | -1.121 | -1.167 | -1.008 | -1.048 |
| 36 | -1.260 | -1.432 | -1.345 | -1.277 | -1.256 |
| 37 | -1.113 | -1.176 | -1.157 | -1.247 | -1.262 |
| 38 | -1.384 | -1.463 | -1.250 | -1.246 | -1.188 |
| 39 | -1.380 | -1.369 | -1.325 | -1.437 | -1.360 |
| 40 | -1.417 | -1.305 | -1.283 | -1.393 | -1.354 |
| 41 | -1.415 | -1.275 | -1.378 | -1.495 | -1.478 |
| 42 | -1.351 | -1.163 | -1.297 | -1.391 | -1.400 |

Figure 40 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.517 | -1.308 | -1.433 | -1.633 | -1.668 |
| 44 | -1.448 | -1.258 | -1.440 | -1.487 | -1.472 |
| 45 | -1.182 | -1.011 | -1.215 | -1.271 | -1.245 |
| 46 | -.655 | -.529 | -.728 | -.686 | -.640 |
| 47 | -.004 | .132 | -.086 | -.026 | .057 |
| 48 | .448 | .527 | .395 | .425 | .491 |
| 49 | .775 | .760 | .784 | .767 | .766 |
| 50 | .715 | .677 | .769 | .746 | .731 |
| 51 | .434 | .357 | .518 | .447 | .408 |
| 52 | .033 | -.077 | .124 | .062 | .001 |
| 53 | -.389 | -.524 | -.300 | -.355 | -.421 |
| 54 | -1.023 | -1.183 | -.870 | -.951 | -.998 |
| 55 | -1.407 | -1.564 | -1.281 | -1.287 | -1.304 |
| 56 | -1.888 | -2.073 | -1.821 | -1.802 | -1.823 |
| 57 | -1.990 | -2.160 | -1.965 | -1.773 | -1.770 |
| 58 | -1.929 | -2.142 | -2.007 | -1.756 | -1.730 |
| 59 | -1.908 | -2.075 | -1.943 | -1.709 | -1.684 |
| 60 | -1.926 | -2.154 | -1.957 | -1.796 | -1.722 |
| 61 | -1.834 | -2.015 | -1.837 | -1.680 | -1.555 |
| 62 | -1.811 | -2.119 | -1.846 | -1.846 | -1.782 |
| 63 | -1.656 | -1.878 | -1.645 | -1.510 | -1.525 |
| 64 | -2.062 | -2.402 | -2.246 | -2.224 | -2.012 |
| 65 | -1.088 | -1.303 | -1.623 | -1.102 | -1.128 |
| 66 | -1.243 | -1.314 | -1.160 | -1.206 | -1.183 |
| 67 | -1.141 | -1.108 | -1.154 | -1.073 | -1.064 |
| 68 | -1.223 | -1.117 | -1.099 | -1.074 | -1.108 |
| 69 | -1.115 | -.987 | -1.005 | -.969 | -.986 |
| 70 | -1.058 | -.953 | -.976 | -1.010 | -1.055 |
| 71 | -.957 | -.826 | -.909 | -.888 | -.910 |
| 72 | -.963 | -.823 | -.912 | -.951 | -1.016 |
| 73 | -.923 | -.789 | -.878 | -.901 | -.921 |
| 74 | -.930 | -.787 | -.899 | -.948 | -.971 |
| 75 | -.879 | -.736 | -.849 | -.878 | -.883 |
| 76 | -.892 | -.764 | -.881 | -.936 | -.957 |
| 77 | -.958 | -.808 | -.891 | -.971 | -.990 |
| 78 | -1.011 | -.874 | -1.013 | -1.057 | -1.069 |
| 79 | -.850 | -.704 | -.858 | -.860 | -.842 |
| 80 | -.583 | -.473 | -.662 | -.617 | -.581 |
| 81 | -.233 | -.145 | -.311 | -.248 | -.213 |
| 82 | .170 | .260 | .086 | .172 | .226 |
| 83 | .458 | .510 | .401 | .448 | .489 |
| 84 | .763 | .798 | .735 | .779 | .809 |

Figure 40 (continued) – Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .847 | .844 | .858 | .840 | .870 |
| 2 | .639 | .629 | .681 | .686 | .624 |
| 3 | .157 | .107 | .287 | .278 | .154 |
| 4 | -.512 | -.569 | -.403 | -.349 | -.523 |
| 5 | -1.181 | -1.245 | -1.093 | -.976 | -1.201 |
| 6 | -.961 | -1.040 | -.957 | -.837 | -.988 |
| 7 | -1.267 | -1.263 | -1.260 | -1.033 | -1.203 |
| 8 | -1.447 | -1.066 | -1.039 | -.912 | -.964 |
| 9 | -1.938 | -1.703 | -1.904 | -.951 | -1.697 |
| 10 | -1.577 | -1.288 | -1.208 | -.849 | -1.182 |
| 11 | -1.366 | -1.349 | -1.422 | -.747 | -1.267 |
| 12 | -1.116 | -1.100 | -.991 | -.761 | -.912 |
| 13 | -1.330 | -1.669 | -1.318 | -1.720 | -1.554 |
| 14 | -1.791 | -1.945 | -1.952 | -2.221 | -1.878 |
| 15 | -1.857 | -1.936 | -1.988 | -2.277 | -2.150 |
| 16 | -1.864 | -1.890 | -1.945 | -2.271 | -1.899 |
| 17 | -1.908 | -1.817 | -1.949 | -2.208 | -1.992 |
| 18 | -1.777 | -1.714 | -1.745 | -2.087 | -1.812 |
| 19 | -2.011 | -1.971 | -2.042 | -2.375 | -2.130 |
| 20 | -1.684 | -1.657 | -1.752 | -1.991 | -1.730 |
| 21 | -1.414 | -1.308 | -1.498 | -1.624 | -1.400 |
| 22 | -.691 | -.602 | -.785 | -.836 | -.638 |
| 23 | .064 | .127 | -.027 | -.022 | .111 |
| 24 | .532 | .586 | .477 | .503 | .566 |
| 25 | .777 | .768 | .762 | .762 | .792 |
| 26 | .600 | .586 | .641 | .626 | .591 |
| 27 | .161 | .085 | .244 | .231 | .143 |
| 28 | -.458 | -.549 | -.363 | -.360 | -.469 |
| 29 | -1.299 | -1.400 | -1.219 | -1.173 | -1.327 |
| 30 | -1.709 | -1.817 | -1.662 | -1.572 | -1.673 |
| 31 | -1.985 | -1.999 | -2.035 | -1.763 | -1.880 |
| 32 | -1.743 | -1.835 | -1.787 | -1.613 | -1.696 |
| 33 | -2.101 | -2.153 | -2.123 | -1.900 | -2.051 |
| 34 | -1.182 | -1.524 | -1.848 | -1.801 | -1.262 |
| 35 | -1.045 | -1.086 | -1.043 | -1.033 | -.997 |
| 36 | -.770 | -.961 | -1.116 | -1.250 | -.822 |
| 37 | -1.156 | -1.277 | -1.145 | -1.386 | -1.247 |
| 38 | -1.445 | -1.479 | -1.409 | -1.319 | -1.254 |
| 39 | -1.460 | -1.472 | -1.442 | -1.442 | -1.428 |
| 40 | -1.447 | -1.405 | -1.382 | -1.433 | -1.380 |
| 41 | -1.465 | -1.378 | -1.435 | -1.523 | -1.484 |
| 42 | -1.351 | -1.262 | -1.289 | -1.447 | -1.376 |

Figure 41 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.573 | -1.434 | -1.461 | -1.645 | -1.617 |
| 44 | -1.456 | -1.349 | -1.424 | -1.537 | -1.439 |
| 45 | -1.192 | -1.092 | -1.248 | -1.302 | -1.223 |
| 46 | -.653 | -.569 | -.732 | -.738 | -.626 |
| 47 | .018 | .067 | -.092 | -.044 | .046 |
| 48 | .455 | .495 | .382 | .427 | .483 |
| 49 | .761 | .780 | .757 | .764 | .768 |
| 50 | .717 | .689 | .768 | .742 | .727 |
| 51 | .419 | .395 | .479 | .467 | .402 |
| 52 | -.012 | -.068 | .111 | .070 | -.018 |
| 53 | -.447 | -.508 | -.318 | -.347 | -.450 |
| 54 | -1.075 | -1.116 | -.949 | -.883 | -1.095 |
| 55 | -1.442 | -1.490 | -1.331 | -1.210 | -1.426 |
| 56 | -1.961 | -2.058 | -1.907 | -1.804 | -1.927 |
| 57 | -2.018 | -2.146 | -1.999 | -1.870 | -1.892 |
| 58 | -2.035 | -2.023 | -2.117 | -1.742 | -1.892 |
| 59 | -1.940 | -1.993 | -1.933 | -1.698 | -1.831 |
| 60 | -2.059 | -2.073 | -2.103 | -1.828 | -1.866 |
| 61 | -1.894 | -1.940 | -1.878 | -1.696 | -1.805 |
| 62 | -2.057 | -2.132 | -2.153 | -1.761 | -1.913 |
| 63 | -1.824 | -1.937 | -1.823 | -1.623 | -1.710 |
| 64 | -1.433 | -1.782 | -2.240 | -2.001 | -1.558 |
| 65 | -1.008 | -1.092 | -1.072 | -1.332 | -.967 |
| 66 | -1.171 | -1.213 | -1.295 | -1.353 | -1.164 |
| 67 | -1.129 | -1.115 | -1.081 | -1.242 | -1.076 |
| 68 | -1.362 | -1.209 | -1.219 | -1.199 | -1.183 |
| 69 | -1.180 | -1.058 | -1.047 | -1.075 | -1.025 |
| 70 | -1.141 | -1.010 | -1.034 | -1.075 | -1.078 |
| 71 | -.981 | -.885 | -.911 | -.954 | -.932 |
| 72 | -.990 | -.901 | -.963 | -1.027 | -1.039 |
| 73 | -.925 | -.857 | -.901 | -.967 | -.936 |
| 74 | -.969 | -.873 | -.935 | -.995 | -.997 |
| 75 | -.886 | -.830 | -.867 | -.934 | -.902 |
| 76 | -.906 | -.856 | -.911 | -.961 | -.986 |
| 77 | -.946 | -.922 | -.915 | -1.006 | -1.016 |
| 78 | -1.025 | -.967 | -1.048 | -1.074 | -1.090 |
| 79 | -.849 | -.793 | -.870 | -.895 | -.863 |
| 80 | -.614 | -.509 | -.665 | -.648 | -.596 |
| 81 | -.251 | -.162 | -.308 | -.286 | -.214 |
| 82 | .161 | .227 | .092 | .142 | .227 |
| 83 | .446 | .502 | .384 | .427 | .493 |
| 84 | .775 | .787 | .753 | .763 | .812 |

Figure 41 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 90^\circ$, $\phi_b = 270^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .799 | .842 | .841 | .820 | .829 |
| 2 | .755 | .709 | .694 | .758 | .725 |
| 3 | .372 | .341 | .300 | .408 | .329 |
| 4 | -.291 | -.361 | -.371 | -.258 | -.330 |
| 5 | -.999 | -1.208 | -1.165 | -.981 | -1.110 |
| 6 | -1.337 | -1.609 | -1.423 | -1.312 | -1.359 |
| 7 | -1.675 | -2.010 | -1.681 | -1.644 | -1.608 |
| 8 | -1.293 | -1.460 | -1.316 | -1.266 | -1.280 |
| 9 | -1.364 | -1.555 | -1.404 | -1.379 | -1.334 |
| 10 | -1.201 | -1.352 | -1.226 | -1.236 | -1.290 |
| 11 | -1.793 | -2.162 | -1.805 | -1.816 | -1.859 |
| 12 | -1.386 | -1.400 | -1.199 | -1.389 | -1.290 |
| 13 | -.970 | -1.076 | -1.067 | -.982 | -1.099 |
| 14 | -1.774 | -1.840 | -1.788 | -1.793 | -1.793 |
| 15 | -1.752 | -1.928 | -1.779 | -1.737 | -1.732 |
| 16 | -1.900 | -1.713 | -2.006 | -1.975 | -2.040 |
| 17 | -2.008 | -1.841 | -2.010 | -2.012 | -2.028 |
| 18 | -1.861 | -1.556 | -1.915 | -1.927 | -1.994 |
| 19 | -2.121 | -1.821 | -2.221 | -2.223 | -2.271 |
| 20 | -1.926 | -1.646 | -1.885 | -2.000 | -1.969 |
| 21 | -1.609 | -1.382 | -1.557 | -1.654 | -1.592 |
| 22 | -.960 | -.794 | -.843 | -.992 | -.898 |
| 23 | -.196 | -.081 | -.071 | -.208 | -.092 |
| 24 | .382 | .430 | .453 | .379 | .457 |
| 25 | .711 | .762 | .762 | .721 | .749 |
| 26 | .710 | .681 | .680 | .725 | .703 |
| 27 | .470 | .359 | .396 | .477 | .409 |
| 28 | -.061 | -.275 | -.170 | -.061 | -.137 |
| 29 | -.800 | -1.074 | -.853 | -.735 | -.798 |
| 30 | -1.188 | -1.519 | -1.135 | -1.092 | -1.082 |
| 31 | -1.358 | -1.872 | -1.267 | -1.221 | -1.186 |
| 32 | -1.323 | -1.804 | -1.270 | -1.274 | -1.225 |
| 33 | -1.498 | -1.963 | -1.438 | -1.398 | -1.357 |
| 34 | -1.385 | -1.823 | -1.287 | -1.240 | -1.261 |
| 35 | -1.453 | -2.280 | -1.302 | -1.291 | -1.241 |
| 36 | -1.414 | -2.179 | -1.104 | -1.217 | -1.089 |
| 37 | -.784 | -1.119 | -.841 | -.763 | -.827 |
| 38 | -1.872 | -1.505 | -1.965 | -1.967 | -2.077 |
| 39 | -1.713 | -1.105 | -1.962 | -1.876 | -2.042 |
| 40 | -1.849 | -1.125 | -2.051 | -2.053 | -2.187 |
| 41 | -1.900 | -1.203 | -2.020 | -2.040 | -2.080 |
| 42 | -1.756 | -1.121 | -1.819 | -1.871 | -1.898 |

Figure 42 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.841 | -1.211 | -2.062 | -2.033 | -2.101 |
| 44 | -1.848 | -1.277 | -1.885 | -1.982 | -1.970 |
| 45 | -1.649 | -1.147 | -1.597 | -1.692 | -1.646 |
| 46 | -1.102 | -.715 | -.961 | -1.127 | -1.015 |
| 47 | -.369 | -.134 | -.218 | -.368 | -.257 |
| 48 | .215 | .337 | .334 | .226 | .316 |
| 49 | .766 | .839 | .821 | .786 | .812 |
| 50 | .777 | .739 | .754 | .781 | .775 |
| 51 | .685 | .561 | .639 | .709 | .647 |
| 52 | .405 | .149 | .324 | .440 | .348 |
| 53 | .056 | -.201 | -.050 | .074 | -.017 |
| 54 | -.309 | -.634 | -.378 | -.264 | -.353 |
| 55 | -.790 | -1.231 | -.755 | -.679 | -.706 |
| 56 | -.938 | -1.435 | -.786 | -.803 | -.757 |
| 57 | -1.049 | -1.889 | -.820 | -.782 | -.758 |
| 58 | -.986 | -1.686 | -.781 | -.761 | -.746 |
| 59 | -1.042 | -1.682 | -.814 | -.759 | -.758 |
| 60 | -.988 | -1.545 | -.780 | -.747 | -.747 |
| 61 | -1.012 | -1.734 | -.923 | -.907 | -.841 |
| 62 | -.946 | -1.556 | -.872 | -.881 | -.879 |
| 63 | -1.021 | -1.588 | -.885 | -.849 | -.861 |
| 64 | -.990 | -1.485 | -.834 | -.840 | -.858 |
| 65 | -1.013 | -1.550 | -.898 | -.921 | -.899 |
| 66 | -1.129 | -1.535 | -1.075 | -.953 | -.908 |
| 67 | -.746 | -1.156 | -.877 | -.788 | -1.091 |
| 68 | -1.418 | -.872 | -1.435 | -1.443 | -1.513 |
| 69 | -1.686 | -.985 | -1.578 | -1.588 | -1.519 |
| 70 | -1.725 | -.805 | -1.592 | -1.642 | -1.540 |
| 71 | -1.989 | -.860 | -1.728 | -1.653 | -1.611 |
| 72 | -1.814 | -.747 | -1.668 | -1.615 | -1.551 |
| 73 | -1.861 | -.813 | -1.739 | -1.684 | -1.662 |
| 74 | -1.690 | -.742 | -1.644 | -1.659 | -1.602 |
| 75 | -1.722 | -.784 | -1.729 | -1.720 | -1.720 |
| 76 | -1.702 | -.725 | -1.687 | -1.688 | -1.752 |
| 77 | -1.942 | -.771 | -1.870 | -1.888 | -1.949 |
| 78 | -1.833 | -.836 | -1.671 | -1.759 | -1.771 |
| 79 | -1.623 | -.815 | -1.534 | -1.642 | -1.582 |
| 80 | -1.230 | -.610 | -1.090 | -1.235 | -1.142 |
| 81 | -.768 | -.314 | -.620 | -.779 | -.636 |
| 82 | -.285 | .025 | -.141 | -.285 | -.147 |
| 83 | .208 | .418 | .342 | .209 | .301 |
| 84 | .512 | .608 | .583 | .511 | .579 |

Figure 42 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .00$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .856 | .834 | .854 | .821 | .858 |
| 2 | .648 | .693 | .602 | .699 | .655 |
| 3 | .182 | .312 | .056 | .295 | .189 |
| 4 | -.569 | -.392 | -.715 | -.390 | -.527 |
| 5 | -1.520 | -1.302 | -1.616 | -1.191 | -1.336 |
| 6 | -2.013 | -1.782 | -2.026 | -1.542 | -1.646 |
| 7 | -2.507 | -2.262 | -2.436 | -1.893 | -1.956 |
| 8 | -1.808 | -1.609 | -1.784 | -1.395 | -1.482 |
| 9 | -1.806 | -1.643 | -1.825 | -1.481 | -1.566 |
| 10 | -1.141 | -.829 | -1.035 | -.709 | -1.564 |
| 11 | -1.863 | -1.373 | -1.748 | -1.138 | -1.195 |
| 12 | -1.204 | -.942 | -1.059 | -.693 | -1.010 |
| 13 | -1.424 | -1.258 | -1.381 | -1.020 | -1.306 |
| 14 | -1.546 | -1.567 | -1.875 | -1.926 | -1.655 |
| 15 | -1.573 | -1.647 | -1.623 | -2.115 | -2.006 |
| 16 | -1.516 | -1.618 | -1.642 | -2.008 | -1.919 |
| 17 | -1.531 | -1.651 | -1.614 | -2.062 | -1.978 |
| 18 | -1.385 | -1.500 | -1.504 | -1.792 | -1.806 |
| 19 | -1.630 | -1.733 | -1.756 | -2.107 | -2.124 |
| 20 | -1.440 | -1.576 | -1.469 | -1.797 | -1.747 |
| 21 | -1.169 | -1.313 | -1.148 | -1.517 | -1.417 |
| 22 | -.589 | -.731 | -.516 | -.834 | -.707 |
| 23 | .096 | -.030 | .197 | -.099 | .046 |
| 24 | .541 | .471 | .607 | .437 | .526 |
| 25 | .787 | .775 | .799 | .760 | .809 |
| 26 | .617 | .662 | .590 | .664 | .637 |
| 27 | .190 | .302 | .071 | .309 | .235 |
| 28 | -.512 | -.347 | -.637 | -.318 | -.412 |
| 29 | -1.429 | -1.289 | -1.544 | -1.077 | -1.239 |
| 30 | -1.956 | -1.821 | -2.004 | -1.504 | -1.562 |
| 31 | -2.676 | -2.437 | -2.474 | -1.809 | -1.775 |
| 32 | -2.452 | -2.250 | -2.290 | -1.767 | -1.734 |
| 33 | -2.664 | -2.438 | -2.500 | -1.951 | -1.971 |
| 34 | -2.460 | -2.322 | -2.480 | -1.987 | -2.002 |
| 35 | -2.458 | -2.211 | -1.921 | -1.631 | -2.464 |
| 36 | -1.533 | -1.393 | -1.456 | -1.212 | -1.271 |
| 37 | -1.672 | -1.456 | -1.468 | -1.139 | -1.535 |
| 38 | -1.302 | -1.315 | -1.378 | -1.260 | -1.239 |
| 39 | -1.273 | -1.338 | -1.338 | -1.267 | -1.318 |
| 40 | -1.058 | -1.148 | -1.163 | -1.341 | -1.311 |
| 41 | -1.075 | -1.172 | -1.158 | -1.414 | -1.383 |
| 42 | -.934 | -1.038 | -1.010 | -1.335 | -1.281 |

Figure 43 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|------------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.005 | -1.102 | -1.102 | -1.504 | -1.489 |
| 44 | -1.056 | -1.171 | -1.121 | -1.509 | -1.417 |
| 45 | -.908 | -1.038 | -.892 | -1.286 | -1.185 |
| 46 | -.493 | -.614 | -.451 | -.774 | -.653 |
| 47 | .055 | -.071 | .141 | -.115 | -.022 |
| 48 | .462 | .389 | .537 | .376 | .438 |
| 49 | .838 | .820 | .834 | .826 | .860 |
| 50 | .691 | .729 | .669 | .733 | .716 |
| 51 | .500 | .549 | .391 | .571 | .520 |
| 52 | .096 | .179 | -.016 | .222 | .143 |
| 53 | -.480 | -.372 | -.589 | -.262 | -.358 |
| 54 | -.956 | -.842 | -1.037 | -.680 | -.765 |
| 55 | -1.719 | -1.477 | -1.778 | -1.271 | -1.406 |
| 56 | -2.009 | -1.785 | -2.018 | -1.504 | -1.563 |
| 57 | -2.632 | -2.430 | -2.455 | -1.842 | -1.747 |
| 58 | -2.545 | -2.368 | -2.254 | -1.652 | -1.578 |
| 59 | -2.701 | -2.415 | -2.336 | -1.779 | -1.675 |
| 60 | -2.456 | -2.243 | -2.156 | -1.636 | -1.533 |
| 61 | -2.630 | -2.359 | -2.323 | -1.835 | -1.650 |
| 62 | -2.352 | -2.170 | -2.108 | -1.809 | -1.541 |
| 63 | -2.425 | -2.187 | -2.586 | -2.216 | -1.831 |
| 64 | -2.280 | -2.231 | -2.851 | -2.468 | -1.810 |
| 65 | -2.645 | -2.287 | -2.376 | -2.000 | -1.863 |
| 66 | -2.025 | -2.204 | -2.409 | -2.509 | -1.740 |
| 67 | -1.289 | -1.263 | -1.239 | -1.178 | -.981 |
| 68 | -1.144 | -1.289 | -1.306 | -1.453 | -1.136 |
| 69 | -.884 | -1.018 | -.963 | -1.114 | -.989 |
| 70 | -.765 | -.874 | -.849 | -1.022 | -.919 |
| 71 | -.742 | -.839 | -.804 | -1.061 | -.981 |
| 72 | -.657 | -.725 | -.725 | -.958 | -.878 |
| 73 | -.687 | -.725 | -.746 | -.984 | -.931 |
| 74 | -.624 | -.674 | -.681 | -.910 | -.845 |
| 75 | -.662 | -.701 | -.700 | -.948 | -.886 |
| 76 | -.601 | -.648 | -.653 | -.877 | -.817 |
| 77 | -.659 | -.683 | -.750 | -1.010 | -.945 |
| 78 | -.717 | -.760 | -.779 | -1.027 | -.927 |
| 79 | -.695 | -.753 | -.690 | -.988 | -.863 |
| 80 | -.475 | -.558 | -.446 | -.705 | -.576 |
| 81 | -.187 | -.271 | -.103 | -.365 | -.226 |
| 82 | .132 | .061 | .225 | .023 | .142 |
| 83 | .516 | .444 | .583 | .420 | .532 |
| 84 | .670 | .636 | .710 | .626 | .688 |

Figure 43 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .01$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 1 | .837 | .829 | .857 | .810 | .857 |
| 2 | .663 | .714 | .583 | .711 | .649 |
| 3 | .178 | .342 | .038 | .383 | .196 |
| 4 | -.568 | -.326 | -.741 | -.253 | -.514 |
| 5 | -1.437 | -1.182 | -1.670 | -1.031 | -1.329 |
| 6 | -1.844 | -1.573 | -2.020 | -1.303 | -1.635 |
| 7 | -2.251 | -1.965 | -2.369 | -1.575 | -1.942 |
| 8 | -1.740 | -1.468 | -1.822 | -1.222 | -1.459 |
| 9 | -1.511 | -1.189 | -1.548 | -1.101 | -1.440 |
| 10 | -1.708 | -1.165 | -1.749 | -.647 | -.694 |
| 11 | -1.813 | -1.330 | -1.861 | -.937 | -1.088 |
| 12 | -1.395 | -1.076 | -1.362 | -.689 | -.658 |
| 13 | -1.566 | -1.360 | -1.661 | -1.043 | -1.282 |
| 14 | -1.731 | -1.788 | -1.810 | -2.312 | -1.630 |
| 15 | -1.685 | -1.895 | -1.654 | -2.113 | -2.019 |
| 16 | -1.652 | -1.763 | -1.589 | -2.044 | -1.935 |
| 17 | -1.581 | -1.797 | -1.582 | -2.160 | -2.017 |
| 18 | -1.468 | -1.599 | -1.448 | -1.958 | -1.813 |
| 19 | -1.666 | -1.843 | -1.738 | -2.313 | -2.175 |
| 20 | -1.509 | -1.679 | -1.440 | -1.955 | -1.750 |
| 21 | -1.210 | -1.390 | -1.116 | -1.654 | -1.439 |
| 22 | -.644 | -.806 | -.483 | -.932 | -.706 |
| 23 | .057 | -.061 | .227 | -.190 | .037 |
| 24 | .523 | .456 | .626 | .371 | .512 |
| 25 | .784 | .763 | .806 | .752 | .791 |
| 26 | .630 | .668 | .578 | .667 | .627 |
| 27 | .197 | .336 | .068 | .364 | .245 |
| 28 | -.504 | -.289 | -.650 | -.221 | -.381 |
| 29 | -1.400 | -1.157 | -1.585 | -1.052 | -1.251 |
| 30 | -1.936 | -1.652 | -2.029 | -1.440 | -1.562 |
| 31 | -2.486 | -2.187 | -2.523 | -1.723 | -1.804 |
| 32 | -2.323 | -2.042 | -2.321 | -1.652 | -1.732 |
| 33 | -2.462 | -2.183 | -2.508 | -1.795 | -1.981 |
| 34 | -2.423 | -2.212 | -2.331 | -1.843 | -1.916 |
| 35 | -1.542 | -1.205 | -1.537 | -1.115 | -1.748 |
| 36 | -1.557 | -1.276 | -1.651 | -1.211 | -1.203 |
| 37 | -1.167 | -.979 | -1.343 | -1.281 | -1.263 |
| 38 | -1.292 | -1.385 | -1.444 | -1.288 | -1.241 |
| 39 | -1.229 | -1.350 | -1.279 | -1.435 | -1.319 |
| 40 | -1.102 | -1.289 | -1.131 | -1.454 | -1.360 |
| 41 | -1.120 | -1.301 | -1.138 | -1.555 | -1.477 |
| 42 | -1.029 | -1.178 | -1.000 | -1.467 | -1.360 |

Figure 44 - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .02$

| Pressure Port | Blowing Nozzle | | | | |
|---------------|----------------|--------|--------|--------|--------|
| | 1 | 2 | 3 | 4 | 5 |
| 43 | -1.082 | -1.258 | -1.108 | -1.653 | -1.599 |
| 44 | -1.172 | -1.339 | -1.127 | -1.607 | -1.488 |
| 45 | -.968 | -1.166 | -.894 | -1.392 | -1.262 |
| 46 | -.562 | -.732 | -.441 | -.853 | -.701 |
| 47 | .023 | -.126 | .163 | -.201 | -.046 |
| 48 | .445 | .356 | .534 | .313 | .413 |
| 49 | .822 | .818 | .840 | .828 | .859 |
| 50 | .698 | .732 | .640 | .730 | .702 |
| 51 | .471 | .571 | .363 | .587 | .507 |
| 52 | .088 | .202 | -.083 | .235 | .122 |
| 53 | -.473 | -.290 | -.658 | -.272 | -.392 |
| 54 | -.955 | -.744 | -1.133 | -.697 | -.806 |
| 55 | -1.675 | -1.402 | -1.881 | -1.277 | -1.430 |
| 56 | -1.992 | -1.704 | -2.125 | -1.504 | -1.585 |
| 57 | -2.537 | -2.218 | -2.662 | -1.872 | -1.835 |
| 58 | -2.469 | -2.140 | -2.455 | -1.697 | -1.597 |
| 59 | -2.542 | -2.173 | -2.546 | -1.784 | -1.755 |
| 60 | -2.400 | -2.039 | -2.346 | -1.607 | -1.555 |
| 61 | -2.459 | -2.203 | -2.495 | -1.833 | -1.725 |
| 62 | -2.223 | -1.964 | -2.243 | -1.790 | -1.726 |
| 63 | -2.754 | -2.226 | -2.923 | -2.288 | -2.493 |
| 64 | -2.824 | -2.399 | -2.736 | -2.343 | -2.662 |
| 65 | -2.235 | -2.219 | -1.904 | -1.734 | -2.452 |
| 66 | -2.870 | -2.551 | -2.742 | -2.380 | -2.708 |
| 67 | -1.230 | -1.285 | -1.226 | -1.240 | -1.307 |
| 68 | -1.458 | -1.428 | -1.431 | -1.446 | -1.510 |
| 69 | -.989 | -1.076 | -1.041 | -1.159 | -1.194 |
| 70 | -.881 | -.950 | -.904 | -1.051 | -1.069 |
| 71 | -.808 | -.916 | -.841 | -1.058 | -1.084 |
| 72 | -.747 | -.805 | -.741 | -.972 | -.945 |
| 73 | -.738 | -.868 | -.747 | -1.034 | -1.022 |
| 74 | -.695 | -.798 | -.684 | -.934 | -.909 |
| 75 | -.708 | -.833 | -.711 | -.975 | -.972 |
| 76 | -.680 | -.778 | -.662 | -.899 | -.890 |
| 77 | -.720 | -.823 | -.763 | -1.013 | -1.075 |
| 78 | -.803 | -.899 | -.774 | -1.036 | -1.006 |
| 79 | -.737 | -.880 | -.687 | -.984 | -.928 |
| 80 | -.531 | -.661 | -.433 | -.728 | -.605 |
| 81 | -.203 | -.355 | -.092 | -.363 | -.256 |
| 82 | .127 | -.004 | .224 | .014 | .120 |
| 83 | .493 | .410 | .605 | .419 | .518 |
| 84 | .663 | .612 | .709 | .615 | .671 |

Figure 44 (continued) - Pressure Coefficients
 $\alpha = 60^\circ$, $\phi_j = 120^\circ$, $\phi_b = 240^\circ$, $C_\mu = .02$

